

SERVICE BULLETIN
REVISION TRANSMITTAL SHEET

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ATA SYSTEM: 57

TITLE: WINGS - GENERAL - ONE TIME INSPECTION OF TAPERLOK BOLT AT WING TO FUSELAGE JUNCTION.

MODIFICATION No.: None

This page transmits Revision No.03 of Service Bulletin No.A320-57-1140

ADDITIONAL WORK****CONF ALL**

Additional work is required by this revision for aircraft inspected by the original issue of this Service Bulletin.

No additional work is required by this revision for aircraft inspected by this Service Bulletin at revision No. 01 or revision No. 02.

The ADDITIONAL WORK consists in inspecting new fasteners located in the center and outer wing box in the Rib1 area.

Unless otherwise stated by an Airworthiness Directive, AIRBUS recommends the additional work to be accomplished within one year after the issue of this revision.

REASON

Revision No. 03 issued to incorporate changes found necessary and carried out during the accomplishment of this Service Bulletin on aircraft with Manufacturer Serial Number (MSN) 2288.

NOTE: This revision satisfies the requirements of the subsequent Technical Adaptation(s) (TA) issued for the related Manufacturer Serial Number (MSN)s:

TA	MSN
80206461/012/2016	2288

CHANGES

The main changes of this revision are:

- Manpower updated to be in accordance with the Gantt Chart
- 'APPENDIX 04 - Inspection Report' included
- 'Procurement addresses' updated

5 DATE: Nov 21/06

SERVICE BULLETIN No.: A320-57-1140

REVISION No.: 03 - Apr 19/19

Page: 1

- Kit 571140A01 and Kit 571140A02 updated
- Supplementary Kit 571140A01S03 and Supplementary Kit 571140A02S03 added
- 'Standard Practices' SUBTASKS updated
- Figure A-FAAAA added
- Figures A-FBAAA, A-FCAAA, A-FCBAA, A-FCEAA, A-FCEAB, A-FCFAA, A-FCFAB, A-FCGAA, A-FCGAB, A-FFAAA, A-FFAAB, A-FFBAA and A-FFBAB updated
- Figure A-FRAAA removed
- Minor changes applied.

SUMMARY		
MATERIAL PRICE INFORMATION	MATERIAL PRICE INFORMATION TABLE UPDATED	R
	MATERIAL PRICE INFORMATION TABLE UPDATED	R
EFFECTIVITY	OPERATOR LIST UPDATED	R
REFERENCES / REPERCUSSIONS	REFERENCES/REPERCUSSIONS TABLE UPDATED	R
MANPOWER	MANPOWER UPDATED	R
	MANPOWER UPDATED	R
APPENDICES	APPENDICES UPDATED	R
PLANNING INFORMATION		
EFFECTIVITY	EFFECTIVITY RELATED INFORMATION UPDATED	R
Material Effectivity	MATERIAL EFFECTIVITY UPDATED	R
	MATERIAL SET UPDATED	
	MATERIAL SET QUANTITY UPDATED	
	MATERIAL EFFECTIVITY UPDATED	R
	MATERIAL SET UPDATED	
	MATERIAL SET QUANTITY UPDATED	
MANPOWER	MANPOWER UPDATED	R
	MANPOWER UPDATED	R
REFERENCES	REFERENCED DOCUMENTATION UPDATED	R
	REFERENCED DOCUMENTATION UPDATED	R
MATERIAL INFORMATION		
MATERIAL - PRICE AND AVAILABILITY		
Procurement Addresses	PROCUREMENT ADDRESSES UPDATED MATERIAL SET UPDATED	R
Price and Availability	PRICE AND AVAILABILITY UPDATED MATERIAL AVAILABILITY UPDATED MATERIAL SET UPDATED MATERIAL PRICE UPDATED	R
INDUSTRY SUPPORT INFORMATION	INDUSTRY SUPPORT INFORMATION UPDATED	R
LIST OF COMPONENTS	LIST OF COMPONENTS UPDATED	R

Kit 571140A01R04	PART MATERIAL SET UPDATED MATERIAL SET LABEL UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED	R
Kit 571140A02R04	PART MATERIAL SET UPDATED MATERIAL SET LABEL UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED	R
Supplementary Kit 571140A01S03	PART MATERIAL SET ADDED	N
Supplementary Kit 571140A02S03	PART MATERIAL SET ADDED	N
LIST OF MATERIALS - OPERATOR SUPPLIED		
Components	COMPONENTS UPDATED	R
Component COMPA01	PART MATERIAL SET UPDATED NOTE UPDATED	R
Component COMPA03	PART MATERIAL SET UPDATED NOTE UPDATED	R
Component COMPA04	PART MATERIAL SET UPDATED NOTE UPDATED	R
TOOLING		
Tools	TOOLS UPDATED	R
ACCOMPLISHMENT INSTRUCTIONS		
TASK 571140- 832-801-001-INSPECTION	TASK SOLUTION UPDATED NOTE UPDATED	R
Subtask 571140-910-001-001	SUBTASK SOLUTION UPDATED	R
Subtask 571140-910-001-001	SUBTASK SOLUTION UPDATED	R
Subtask 571140-941-001-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED REFERENCES UPDATED	R
Subtask 571140-941-001-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED REFERENCES UPDATED	R
Subtask 571140-832-001-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED CONFIGURATION UPDATED FIGURE REFERENCE UPDATED NOTE UPDATED REFERENCES UPDATED	R
Subtask 571140-832-002-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED CONFIGURATION UPDATED FIGURE REFERENCE UPDATED NOTE UPDATED REFERENCES UPDATED	R

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REVISION TRANSMITTAL SHEET

Subtask 571140-832-001-002	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED CONFIGURATION UPDATED FIGURE REFERENCE UPDATED NOTE UPDATED REFERENCES UPDATED	R
Subtask 571140-832-002-002	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED CONFIGURATION UPDATED FIGURE REFERENCE UPDATED NOTE UPDATED REFERENCES UPDATED	R
TASK 571140- 833-801-001-REPAIR	TASK SOLUTION UPDATED NOTE UPDATED	R
Subtask 571140-910-002-001	SUBTASK SOLUTION UPDATED	R
Subtask 571140-910-002-001	SUBTASK SOLUTION UPDATED	R
Subtask 571140-833-001-001	SUBTASK SOLUTION UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED ITEM NUMBER UPDATED PART NUMBER DESCRIPTION UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-002-001	SUBTASK SOLUTION UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED ITEM NUMBER UPDATED PART NUMBER DESCRIPTION UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-003-001	SUBTASK SOLUTION UPDATED NOTE UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-004-001	SUBTASK SOLUTION UPDATED NOTE UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-001-002	SUBTASK SOLUTION UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED ITEM NUMBER UPDATED PART NUMBER DESCRIPTION UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-002-002	SUBTASK SOLUTION UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED ITEM NUMBER UPDATED PART NUMBER DESCRIPTION UPDATED MATERIAL LIST UPDATED	R

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Subtask 571140-833-003-002	SUBTASK SOLUTION UPDATED NOTE UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-004-002	SUBTASK SOLUTION UPDATED NOTE UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-410-005-001	SUBTASK SOLUTION UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-410-006-001	SUBTASK SOLUTION UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-942-001-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED REFERENCES UPDATED	R
Subtask 571140-410-005-002	SUBTASK SOLUTION UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-410-006-002	SUBTASK SOLUTION UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-942-001-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED REFERENCES UPDATED	R
TASK 571140- 832-802-001-INSPECTION ADDITIONAL WORK	TASK SOLUTION UPDATED NOTE UPDATED	R
Task associated data - 001	MAN HOURS UPDATED ELAPSED TIME UPDATED	R
Task associated data - 002	MAN HOURS UPDATED ELAPSED TIME UPDATED	R
Subtask 571140-910-003-001	SUBTASK SOLUTION UPDATED	R
Subtask 571140-910-003-001	SUBTASK SOLUTION UPDATED	R
Subtask 571140-941-002-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED REFERENCES UPDATED	R
Subtask 571140-941-002-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED REFERENCES UPDATED	R
Subtask 571140-832-003-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED CONFIGURATION UPDATED FIGURE REFERENCE UPDATED NOTE UPDATED MANPOWER RESSOURCE UPDATED REFERENCES UPDATED	R
Subtask 571140-832-004-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED CONFIGURATION UPDATED FIGURE REFERENCE UPDATED NOTE UPDATED MANPOWER RESSOURCE UPDATED REFERENCES UPDATED	R

SERVICE BULLETIN REVISION TRANSMITTAL SHEET

Subtask 571140-832-003-002	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED CONFIGURATION UPDATED FIGURE REFERENCE UPDATED NOTE UPDATED MANPOWER RESSOURCE UPDATED REFERENCES UPDATED	R
Subtask 571140-832-004-002	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED CONFIGURATION UPDATED FIGURE REFERENCE UPDATED NOTE UPDATED MANPOWER RESSOURCE UPDATED REFERENCES UPDATED	R
TASK 571140- 833-802-001-REPAIR ADDITIONAL WORK	TASK SOLUTION UPDATED NOTE UPDATED	R
Task associated data - 001	MAN HOURS UPDATED ELAPSED TIME UPDATED	R
Task associated data - 002	MAN HOURS UPDATED ELAPSED TIME UPDATED	R
Subtask 571140-910-004-001	SUBTASK SOLUTION UPDATED	R
Subtask 571140-910-004-001	SUBTASK SOLUTION UPDATED	R
Subtask 571140-833-005-001	SUBTASK SOLUTION UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED PART NUMBER DESCRIPTION UPDATED MANPOWER RESSOURCE UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-006-001	SUBTASK SOLUTION UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED CONFIGURATION UPDATED NOTE UPDATED PART NUMBER DESCRIPTION UPDATED FIGURE REFERENCE UPDATED MANPOWER RESSOURCE UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-007-001	SUBTASK SOLUTION UPDATED NOTE UPDATED MANPOWER RESSOURCE UPDATED	R
Subtask 571140-833-008-001	SUBTASK SOLUTION UPDATED NOTE UPDATED MANPOWER RESSOURCE UPDATED	R

SERVICE BULLETIN REVISION TRANSMITTAL SHEET

Subtask 571140-833-005-002	SUBTASK SOLUTION UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED NOTE UPDATED PART NUMBER DESCRIPTION UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-006-002	SUBTASK SOLUTION UPDATED KEYWORD UPDATED PART NUMBER UPDATED QUANTITY UPDATED NOTE UPDATED PART NUMBER DESCRIPTION UPDATED MATERIAL LIST UPDATED	R
Subtask 571140-833-007-002	SUBTASK SOLUTION UPDATED NOTE UPDATED MANPOWER RESSOURCE UPDATED	R
Subtask 571140-833-008-002	SUBTASK SOLUTION UPDATED NOTE UPDATED MANPOWER RESSOURCE UPDATED	R
Subtask 571140-942-002-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED REFERENCES UPDATED	R
Subtask 571140-942-002-001	SUBTASK SOLUTION UPDATED REFERENCED DOCUMENTATION UPDATED REFERENCES UPDATED	R
Fig. A-FBAAA		
Fig. A-FBAAA Sheet 02	SHEET UPDATED	R
Fig. A-FBAAA Sheet 03	SHEET UPDATED	R
Fig. A-FFAAA		
Fig. A-FFAAA Sheet 01	SHEET UPDATED	R
Fig. A-FFAAB		
Fig. A-FFAAB Sheet 01	SHEET UPDATED	R
Fig. A-FCAAAA		
Fig. A-FCAAAA Sheet 01	SHEET UPDATED	R
Fig. A-FCAAAA Sheet 02	SHEET UPDATED	R
Fig. A-FCAAAA Sheet 03	SHEET UPDATED	R
Fig. A-FCAAAA Sheet 04	SHEET UPDATED	R
Fig. A-FCAAAA Sheet 05	SHEET UPDATED	R
Fig. A-FCAAAA Sheet 06	SHEET UPDATED	R
Fig. A-FCAAAA Sheet 07	SHEET UPDATED	R
Fig. A-FCBAA		
Fig. A-FCBAA Sheet 02	SHEET UPDATED	R
Fig. A-FCBAA Sheet 03	SHEET UPDATED	R
Fig. A-FCBAA Sheet 04	SHEET UPDATED	R
Fig. A-FCBAA Sheet 05	SHEET UPDATED	R

SERVICE BULLETIN REVISION TRANSMITTAL SHEET

Fig. A-FCBAA Sheet 06	SHEET UPDATED	R
Fig. A-FCBAA Sheet 07	SHEET UPDATED	R
Fig. A-FCCAA		
Fig. A-FCCAA Sheet 01	SHEET UPDATED	R
Fig. A-FCCAA Sheet 02	SHEET UPDATED	R
Fig. A-FCCAA Sheet 03	SHEET UPDATED	R
Fig. A-FCDA		
Fig. A-FCDA Sheet 01	SHEET UPDATED	R
Fig. A-FDDAA		
Fig. A-FDDAA Sheet 02	SHEET UPDATED	R
Fig. A-FCEAA		
Fig. A-FCEAA Sheet 02	SHEET UPDATED	R
Fig. A-FCEAA Sheet 03	SHEET UPDATED	R
Fig. A-FCEAB		
Fig. A-FCEAB Sheet 02	SHEET UPDATED	R
Fig. A-FCEAB Sheet 03	SHEET UPDATED	R
Fig. A-FFBAA		
Fig. A-FFBAA Sheet 01	SHEET UPDATED	R
Fig. A-FFBAB		
Fig. A-FFBAB Sheet 01	SHEET UPDATED	R
Fig. A-FCFAA		
Fig. A-FCFAA Sheet 01	SHEET UPDATED	R
Fig. A-FCFAA Sheet 02	SHEET UPDATED	R
Fig. A-FCFAA Sheet 03	SHEET UPDATED	R
Fig. A-FCFAA Sheet 04	SHEET UPDATED	R
Fig. A-FCFAA Sheet 05	SHEET UPDATED	R
Fig. A-FCFAA Sheet 06	SHEET UPDATED	R
Fig. A-FCFAA Sheet 07	SHEET UPDATED	R
Fig. A-FCFAB		
Fig. A-FCFAB Sheet 01	SHEET UPDATED	R
Fig. A-FCFAB Sheet 02	SHEET UPDATED	R
Fig. A-FCFAB Sheet 03	SHEET UPDATED	R
Fig. A-FCFAB Sheet 04	SHEET UPDATED	R
Fig. A-FCFAB Sheet 05	SHEET UPDATED	R
Fig. A-FCFAB Sheet 06	SHEET UPDATED	R
Fig. A-FCFAB Sheet 07	SHEET UPDATED	R
Fig. A-FCGAA		
Fig. A-FCGAA Sheet 01	SHEET UPDATED	R
Fig. A-FCGAA Sheet 02	SHEET UPDATED	R
Fig. A-FCGAA Sheet 03	SHEET UPDATED	R
Fig. A-FCGAA Sheet 04	SHEET UPDATED	R

SERVICE BULLETIN
REVISION TRANSMITTAL SHEET

Fig. A-FCGAA Sheet 05	SHEET UPDATED	R
Fig. A-FCGAA Sheet 06	SHEET UPDATED	R
Fig. A-FCGAA Sheet 07	SHEET UPDATED	R
Fig. A-FCGAB		
Fig. A-FCGAB Sheet 01	SHEET UPDATED	R
Fig. A-FCGAB Sheet 02	SHEET UPDATED	R
Fig. A-FCGAB Sheet 03	SHEET UPDATED	R
Fig. A-FCGAB Sheet 04	SHEET UPDATED	R
Fig. A-FCGAB Sheet 05	SHEET UPDATED	R
Fig. A-FCGAB Sheet 06	SHEET UPDATED	R
Fig. A-FCGAB Sheet 07	SHEET UPDATED	R
Fig. A-FRA	FIGURE DELETED FOR APPLICABLE MSN	D
APPENDIX 04 - Inspection Report		
APPENDIX CONTENT	APPENDIX CONTENT ADDED	N
Fig. A-FAAAA	FIGURE SOLUTION ADDED	N
Fig. A-FAAAA Sheet 01	SHEET ADDED	N

FILING INSTRUCTIONS

This Service Bulletin has been generated electronically and is reissued as a complete document. Replace the complete document.

Put this Revision Transmittal Sheet in front of the Service Bulletin.

HISTORY OF PREVIOUS REVISIONS

This Service Bulletin has been validated on A320 aircraft MSN 1387.

Revision No. 01 issued to update the detailed inspection area of the center and outer wing box in the Rib1 and modify the Accomplishment Timescale.

Revision No. 02 issued to correct typo mistakes for kits references.

REVISION SEQUENCE

ORIGINAL: Nov 21/06

REVISION No. : 01 - Nov 25/13

REVISION No. : 02 - Nov 07/14

REVISION No. : 03 - Apr 19/19

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SERVICE BULLETIN
SUMMARY

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This summary is for information only and is not approved for modification of the aircraft.

ATA SYSTEM: 57

**TITLE: WINGS - GENERAL - ONE TIME INSPECTION OF TAPERLOK BOLT AT
WING TO FUSELAGE JUNCTION.**

****CONF ALL**

MODIFICATIONS

None

REASON/DESCRIPTION/OPERATIONAL CONSEQUENCES

During installation of the wing to the center wing box in the AIRBUS A320 family Final Assembly Line (FAL), some taperloks used in the wing to fuselage junction at RIB 1, were found to be non-compliant with the specification.

These fasteners have been used on limited number of aircraft.

Fatigue tests on samples and calculations have demonstrated that a loss in pre-tension results in reduction of the fatigue life of the assembly.

AIRBUS decided to check also the condition of the taperloks compliant to the specification used on older aircraft.

At critical locations, the new calculated fatigue life falls below the aircraft Design Service Goal (DSG) of 48,000 Flight Cycles (FC).

Consequently, two Inspection Service Bulletins were launched to perform an internal (Service Bulletin No. A320-57-1129) and an external (Service Bulletin No. A320-57-1130) ultrasonic Inspection on the stiffeners and the lower panels at rib 1, on the center wing box side and the outer wing box side.

This Inspection Service Bulletin consists in carrying out a detailed inspection of fasteners located in center and outer wing box in rib 1 area.

In accordance with the result of the inspection :

- replace the nut by self-locking nut with appropriate washer.
- replace the damaged taperlok with a new taperlok at next nominal diameter with self-locking nut.

SERVICE BULLETIN
SUMMARY

Accomplishment of this Service Bulletin cancels the inspection requirements of inspection Service Bulletin No. A320-57-1129 & No. A320-57-1130 and restore the fatigue potential in the zone to the level initially certified.

GENERAL EVALUATION

EVALUATION TABLE			
COMPLIANCE	RECOMMENDED (1)	CANCELS INSPECTION SB	YES
POTENTIAL AD	NO	A/C OPERATION AFFECTED	NO
RELIABILITY AFFECTED	NO	PAX COMFORT AFFECTED	NO
COST SAVING	NO	ETOPS AFFECTED	NO
STRUCTURAL LIFE EXTN	NO	VENDOR SB INVOLVED	NO
DE-FUELLING/VENTILATING	YES	VALIDATION ON AIRCRAFT	YES
BOOKLET AVAILABLE	YES		

NOTE (1): Service Bulletin recommended to be accomplished to prevent significant operational disruptions.

NOTE: This Service Bulletin has been validated at original issue on A320 aircraft with Manufacturer Serial Number (MSN) 1387.

NOTE: Two booklets including photographs taken during the validation of this Service Bulletin are available on AirbusWorld - Maintenance and Engineering - AirN@v engineering (in the concerned SB file).

MATERIAL PRICE INFORMATION****CONF 001**

MATERIAL PRICE INFORMATION TABLE			
MATERIAL SET	QTY PER A/C	PRICE PER A/C (USD)	MAIN PARTS
Kit 571140A01R04	2	7,700.00	Bolt, nut, washer
Supplementary kit 571140A01S02	2	4,640.00	Bolt, nut, washer
Supplementary kit 571140A01S03	2	3,860.00	Washer
Kit tool 571140T01R01	1	1,000.00	Gage
Kit tool 571140T02R00	1	1,364.00	Test
TOTAL		18,564.00	
Component COMPA01	1	See SB	Screw, nut , washer
Component COMPA03	1	See SB	Screw, nut , washer

****CONF 002**

MATERIAL PRICE INFORMATION TABLE			
MATERIAL SET	QTY PER A/C	PRICE PER A/C (USD)	MAIN PARTS
Kit 571140A02R04	2	7,700.00	Bolt, nut, washer

SERVICE BULLETIN SUMMARY

MATERIAL PRICE INFORMATION TABLE			
MATERIAL SET	QTY PER A/C	PRICE PER A/C (USD)	MAIN PARTS
Supplementary kit 571140A02S02	2	4,260.00	Bolt, nut, washer
Supplementary kit 571140A02S03	2	3,860.00	Washer
Kit tool 571140T01R01	1	1,000.00	Gage
Kit tool 571140T02R00	1	1,364.00	Test
TOTAL		18,184.00	
Component COMPA01	1	See SB	Screw, nut , washer
Component COMPA04	1	See SB	Screw, nut , washer

EFFECTIVITY

****CONF ALL**

This Service Bulletin is applicable to this (these) operator(s) :

74T	AAF	AAR	AAV	ABL	AHY	AIJ	ALK	AMC	ANZ	ART	ARU	ASL
AUH	AWC	AXM	BAW	BEL	BKP	CES	CSN	CTV	D2F	DQA	EDW	EIN
ETD	EWG	FHY	GCR	HDA	HKE	IBE	IGO	JBU	JSA	JST	KCH	KNE
LDM	LGW	LLM	LZB	NWK	ORF	OTF	PIA	PIC	QTR	RWZ	RZO	SDR
SLK	SVR	TAI	TAM	TAP	THY	UAL	UEA	USA	VIV	VLG	VRD	

CONCURRENT REQUIREMENTS

None

REFERENCES / REPERCUSSIONS

TFU	57.11.00.006
OEB	None
AOT	None
SIL	None
ISI	None
LINE MAINTENANCE AFFECTED	No
LIFE LIMIT	None
OTHERS	OIT SE999.0078/04CL

NATURE OF THE WORK

AIRCRAFT	YES
EQUIPMENT	NO
HARD	NO
SOFT	NO
OBRM	NO

SERVICE BULLETIN
SUMMARY**MANPOWER******CONF 001**

Task 571140-832-801-001: INSPECTION	
TOTAL MANHOURS	69.00
ELAPSED TIME (HOURS)	18.00

Task 571140-833-801-001: REPAIR	
TOTAL MANHOURS	75.50
ELAPSED TIME (HOURS)	25.00

Task 571140-832-802-001: INSPECTION ADDITIONAL WORK	
TOTAL MANHOURS	59.00
ELAPSED TIME (HOURS)	15.50

Task 571140-833-802-001: REPAIR ADDITIONAL WORK	
TOTAL MANHOURS	65.50
ELAPSED TIME (HOURS)	20.50

****CONF 002**

Task 571140-832-801-001: INSPECTION	
TOTAL MANHOURS	69.00
ELAPSED TIME (HOURS)	18.00

Task 571140-833-801-001: REPAIR	
TOTAL MANHOURS	75.50
ELAPSED TIME (HOURS)	25.00

Task 571140-832-802-001: INSPECTION ADDITIONAL WORK	
TOTAL MANHOURS	37.00
ELAPSED TIME (HOURS)	10.00

Task 571140-833-802-001: REPAIR ADDITIONAL WORK	
TOTAL MANHOURS	41.50
ELAPSED TIME (HOURS)	14.50

APPENDICES****CONF ALL**

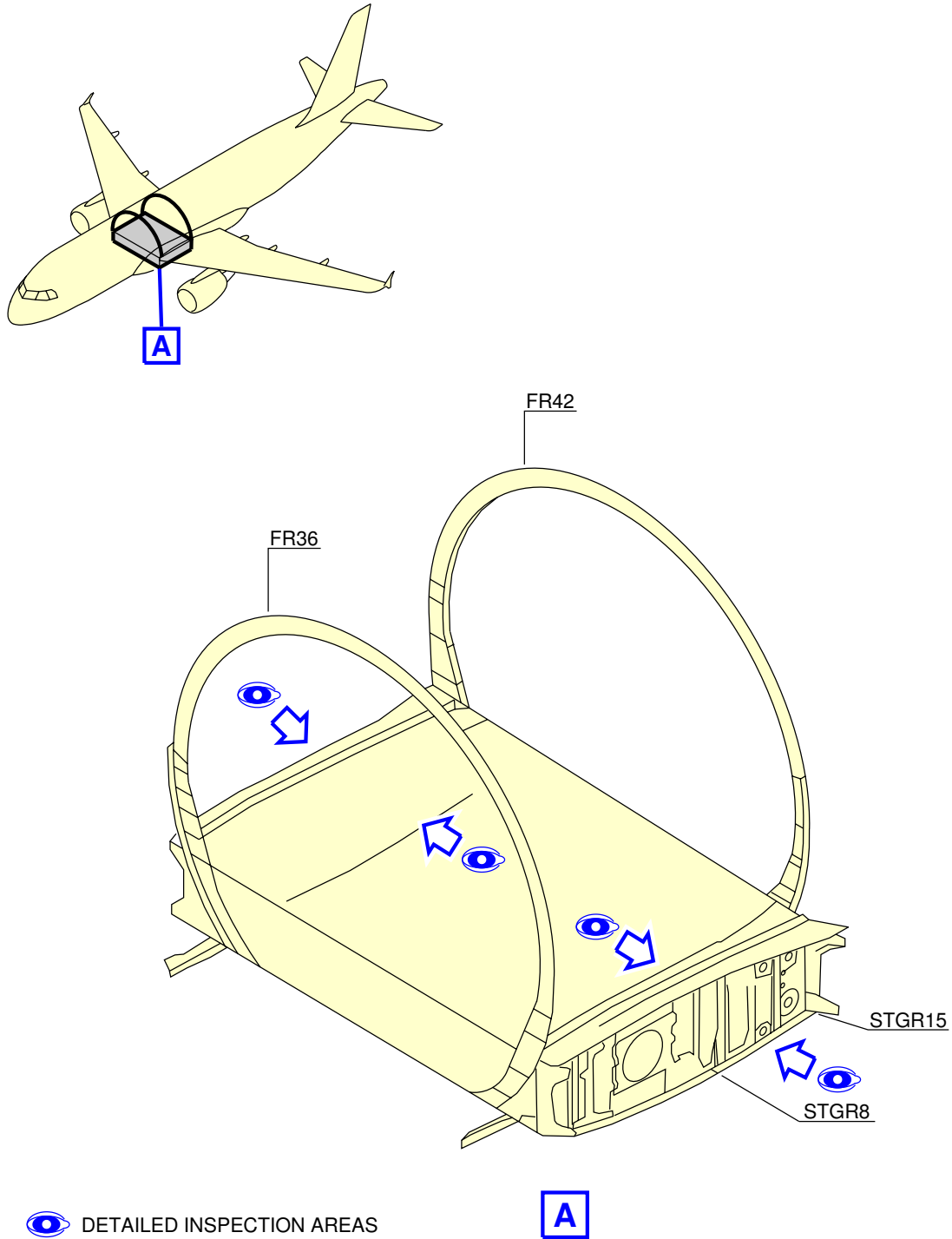
APPENDIX 01 - Weight Variant (WV) correspondence table

APPENDIX 02 - Elapsed Time Assumption

APPENDIX 03 - Elapsed Time Assumption (ADDITIONAL WORK)

APPENDIX 04 - Inspection Report

****CONF ALL**



 DETAILED INSPECTION AREAS

A

N_SB_571140_5_SAAA_01_00

Figure A-FSAAA - Sheet 01

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ATA SYSTEM: 57

TITLE: WINGS - GENERAL - ONE TIME INSPECTION OF TAPERLOK BOLT AT WING TO FUSELAGE JUNCTION.

MODIFICATION No.: None

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1. PLANNING INFORMATION

****CONF ALL**

A. EFFECTIVITY

(1) Models

320-214 320-216 320-232 320-233

(2) Effectivity by MSN

2164-2169 2171 2173 2175 2177-2178 2180 2182-2183 2185 2187 2189 2191 2193
2195 2197 2199 2201 2204 2206-2207 2210 2212 2215 2217 2219 2221 2223 2225
2227 2229 2231 2233 2235 2238-2239 2242 2244 2246 2248 2250 2252 2254
2256-2257 2259 2272 2274-2275 2278 2280 2282 2284 2286 2288 2291-2292 2294
2297 2299 2301 2307 2310 2312 2314 2316 2322 2325 2327 2329 2331 2334 2336
2338 2340 2343 2345 2347 2349 2352 2354 2356 2359 2361 2364 2366 2368 2372
2374 2376 2384 2386 2388 2390-2391 2393 2395 2397 2399 2401 2403 2405 2407
2409 2411 2413 2415 2417 2419 2422-2423 2425 2428 2430 2432 2434 2437 2439
2441 2443 2445 2447 2449 2451 2453 2455 2457 2459 2461 2475 2478-2479 2482
2484 2486 2489 2491 2493 2496 2498 2502 2504 2506 2509 2511 2513 2515 2517
2520 2522 2524 2526 2529 2531 2533 2535 2537 2539-2540 2542 2562 2564 2566
2569 2571 2573 2576-2577 2580 2583-2584 2587 2589 2591 2594 2596 2598 2600
2602 2604 2606 2608-2609 2612-2613 2616 2619-2620 2623 2626-2627 2630 2633
2635 2637 2640 2642 2645 2647 2649 2651 2654 2656 2658 2661 2663 2665 2668
2670-2671 2674 2676 2678 2680 2683 2685 2688

(3) Effectivity by Operator

The Operator/MSN relationship is provided for information only and is correct at the time of issue in accordance with the information available to AIRBUS. Any future changes resulting from transfer of an aircraft from one operator to another will not be reflected in this list unless the Service Bulletin is revised for another reason.

OPERATOR	MSN
74T	2165
AAF	2180
AAR	2397 2459
AAV	2331 2340
ABL	2422 2430
AHY	2685
AIJ	2227 2539
ALK	2345
AMC	2189 2291 2665
ANZ	2173 2257 2297 2445 2533 2594 2663
ART	2233
ARU	2676
ASL	2587 2645
AUH	2403
AWC	2564
AXM	2183 2425 2612 2633 2656 2683
BAW	2441
BEL	2207
BKP	2254 2310 2366 2417 2509 2531 2600
CES	2219 2221 2235 2239 2244 2437 2451 2478 2493 2498
CSN	2354 2361 2484 2506 2511 2637
CTV	2598
D2F	2496
DQA	2347
EDW	2606 2627
EIN	2191 2206 2217 2250 2256 2272 2294 2364 2374 2399 2409 2411 2432 2486 2542 2583 2635
ETD	2167
EWG	2591
FHY	2524
GCR	2338 2475
HDA	2229 2238 2428
HKE	2299 2322
IBE	2225 2242 2248 2391
IGO	2195 2204 2275 2334 2359 2388 2405 2443 2457 2482 2584 2609 2626 2649 2670
JBU	2177 2201 2215 2231 2246 2259 2280 2284 2286 2307 2314 2336 2352 2368 2384 2386 2415 2447 2449 2461 2489 2491 2504 2520 2535 2577 2580 2640 2647 2671
JSA	2316 2356 2453 2604

A318/A319/A320/A321

SERVICE BULLETIN

OPERATOR	MSN
JST	2169 2185 2197 2292 2329 2423 2526 2537 2573 2608 2642 2651
KCH	2529
KNE	2171 2182 2199 2569
LDM	2502
LGW	2562
LLM	2413 2419 2439 2571 2688
LZB	2540 2596
NWK	2455 2515
ORF	2566
OTF	2479
PIA	2212 2274
PIC	2517
QTR	2288
RWZ	2193 2372 2393
RZO	2325 2390
SDR	2619 2668
SLK	2252
SVR	2166 2175 2187 2278 2327 2343 2349 2376 2623
TAI	2282 2301 2434
TAM	2513
TAP	2178
THY	2164 2395 2401 2522 2602
UAL	2680
UEA	2654
USA	2312 2613 2630
VIV	2210 2576 2661
VLG	2168 2223 2407 2589 2620 2658 2678
VRD	2616 2674

(4) Configuration by MSN

MSN	CONFIGURATION
2164 2165 2166 2167 2168 2169 2171 2173 2175 2177 2178 2180 2182 2183 2185 2187 2189 2191 2193 2195 2197 2199 2201 2204 2206 2207 2210 2212 2215 2217 2219 2221 2223 2225 2227 2229 2231 2233 2235 2238 2239 2242 2244 2246 2248 2250 2252 2254 2256 2257 2259 2272 2274 2275 2278 2280 2282 2284 2286 2288 2291 2292 2294 2297 2299 2301 2307 2310 2312 2314 2316 2322 2325 2327 2329 2331 2334 2336 2338 2340 2343 2345 2347 2349 2352 2354 2356 2359 2361 2364 2366 2368 2372 2374 2376 2384 2386 2388 2390 2391 2393 2395 2397 2399 2401 2403 2405 2407 2409 2411 2413 2415 2417 2419 2422 2423 2425 2428 2430 2432 2434 2437 2439 2441 2443 2445 2447 2449 2451 2453 2455 2457 2459 2461 2475 2478 2479 2482 2484 2486 2489 2491 2493 2496 2498 2502 2504 2506 2509 2511 2513 2515 2517 2520 2522 2524 2526 2529 2531 2533 2535 2537 2539 2540 2542 2562 2564 2566 2569 2571 2573 2576 2577 2580 2583 2584 2587 2589 2591 2594 2596 2598 2600 2602 2604 2606 2608 2609 2612 2613 2616 2619 2620 2623 2626 2627 2630 2633 2635 2637 2640 2642 2645 2647 2649 2651 2654 2656 2658 2661 2663 2665 2668 2670 2671 2674 2676 2678 2680 2683 2685 2688	001 002

(5) Configuration definition

****CONF 001**

Config. 001 concerns A320-200 aircraft model with a Weight Variant (WV) 000 thru 010 and 013 (Refer to APPENDIX 01 - Weight Variant (WV) correspondence table) and is valid for aircraft on which Modification No. 26503P4809 (REPLACE TAPER-LOCK FASTENERS BY PULL TYPE FASTENERS ON SECTION 21) is embodied.

****CONF 002**

Config. 002 concerns A320-200 aircraft model with a Weight Variant (WV) 011, 012, 014, 015 and 016 (Refer to APPENDIX 01 - Weight Variant (WV) correspondence table) and is valid for aircraft on which Modification No. 26503P4809 (REPLACE TAPER-LOCK FASTENERS BY PULL TYPE FASTENERS ON SECTION 21) is embodied.

(6) Material Effectivity

****CONF 001**

MATERIAL	QTY PER A/C	SEE NOTES
Kit 571140A01R04	2	
Supplementary kit 571140A01S02	2	
Supplementary kit 571140A01S03	2	
Kit tool 571140T01R01	1	
Kit tool 571140T02R00	1	
Component COMPA01	1	
Component COMPA03	1	

MATERIAL	QTY PER A/C	SEE NOTES
Consumable CMLA01	1	

****CONF 002**

MATERIAL	QTY PER A/C	SEE NOTES
Kit 571140A02R04	2	
Supplementary kit 571140A02S02	2	
Supplementary kit 571140A02S03	2	
Kit tool 571140T01R01	1	
Kit tool 571140T02R00	1	
Component COMPA01	1	
Component COMPA04	1	
Consumable CMLA01	1	

****CONF ALL****B. CONCURRENT REQUIREMENTS**

None

C. REASON**(1) History**

During installation of the wing to the center wing box in the AIRBUS A320 family Final Assembly Line (FAL), some taperlocks used in the wing to fuselage junction at RIB 1, were found to be non-compliant with the specification.

These fasteners have been used on limited number of aircraft.

Fatigue tests on samples and calculations have demonstrated that a loss in pre-tension results in reduction of the fatigue life of the assembly.

AIRBUS decided to check also the condition of the taperlocks compliant to the specification used on older aircraft.

At critical locations, the new calculated fatigue life falls below the aircraft Design Service Goal (DSG) of 48,000 Flight Cycles (FC).

Consequently, two Inspection Service Bulletins were launched to perform an internal (Service Bulletin No. A320-57-1129) and an external (Service Bulletin No. A320-57-1130) ultrasonic Inspection on the stiffeners and the lower panels at rib 1, on the center wing box side and the outer wing box side.

(2) Objective/Action

This Inspection Service Bulletin consists in carrying out a detailed inspection of fasteners located in center and outer wing box in rib 1 area.

In accordance with the result of the inspection :

SERVICE BULLETIN

- replace the nut by self-locking nut with appropriate washer.
- replace the damaged taperlok with a new taperlok at next nominal diameter with self-locking nut.

(3) Advantages

Accomplishment of this Service Bulletin cancels the inspection requirements of inspection Service Bulletin No. A320-57-1129 & No. A320-57-1130 and restore the fatigue potential in the zone to the level initially certified.

(4) Operational/Maintenance Consequences

None

D. DESCRIPTION

To accomplish this Service Bulletin it is necessary to :

****CONF 001**

Task 571140-832-801-001: INSPECTION

- (1) LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction
- (2) RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction

Task 571140-833-801-001: REPAIR

- (1) Replace the Nuts on the LH Side
- (2) Replace the Nuts on the RH Side
- (3) Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side
- (4) Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side

Task 571140-832-802-001: INSPECTION ADDITIONAL WORK

- (1) LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK
- (2) RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK

Task 571140-833-802-001: REPAIR ADDITIONAL WORK

- (1) Replace the Nuts on the LH Side, ADDITIONAL WORK
- (2) Replace the Nuts on the RH Side, ADDITIONAL WORK
- (3) Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side,

ADDITIONAL WORK

- (4) Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side,
ADDITIONAL WORK

****CONF 002**

Task 571140-832-801-001: INSPECTION

- (1) LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1
junction
- (2) RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1
junction

Task 571140-833-801-001: REPAIR

- (1) Replace the Nuts on the LH Side
- (2) Replace the Nuts on the RH Side
- (3) Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side
- (4) Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side

Task 571140-832-802-001: INSPECTION ADDITIONAL WORK

- (1) LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1
junction, ADDITIONAL WORK
- (2) RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1
junction, ADDITIONAL WORK

Task 571140-833-802-001: REPAIR ADDITIONAL WORK

- (1) Replace the Nuts on the LH Side, ADDITIONAL WORK
- (2) Replace the Nuts on the RH Side, ADDITIONAL WORK
- (3) Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side,
ADDITIONAL WORK
- (4) Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side,
ADDITIONAL WORK

****CONF ALL****E. COMPLIANCE**

- (1) Classification

RECOMMENDED: Service Bulletin recommended to be accomplished to prevent
significant operational disruptions.

(2) Accomplishment Timescale

Accomplishment of this Service Bulletin is recommended at the earliest opportunity where manpower and facilities are available.

F. APPROVAL

The technical content of this document is approved under authority of Design Organization Approval No. EASA.21J.031.

If an aircraft listed in the effectivity has a modification or repair embodied that is not of AIRBUS origin, and which affects the content of this Service Bulletin, the operator is responsible for obtaining approval by its airworthiness authority for any adaptation necessary before incorporation of the Service Bulletin.

G. MANPOWER

The manpower estimates given in this Service Bulletin are based on the direct labor cost to do the work. These estimates assume that the work will be done by experienced personnel, and may need to be revised upwards to suit operator's circumstances. The estimates do not include the time to prepare, plan or inspect the work. Manufacture and procurement of parts and tools, drying times for paints, sealants, etc., and general administration work are also not included.

****CONF 001**

Task 571140-832-801-001: INSPECTION	
Get Access	17.00
On Aircraft	
LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction	26.00
RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction	26.00
TOTAL MANHOURS	69.00
ELAPSED TIME (HOURS)	18.00

NOTE: For an explanation of the man-hours and elapsed time, refer to the Gantt Chart, [Fig. A-FABAA](#) given in APPENDIX 02 - Elapsed Time Assumption .

Task 571140-833-801-001: REPAIR	
On Aircraft	
Replace the Nuts on the LH Side	12.00
Replace the Nuts on the RH Side	12.00
Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side	16.00
Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side	16.00
Tests	5.00
Close-Up	14.50
TOTAL MANHOURS	75.50

SERVICE BULLETIN

Task 571140-833-801-001: REPAIR	
ELAPSED TIME (HOURS)	25.00

NOTE: For an explanation of the man-hours and elapsed time, refer to the Gantt Chart, [Fig. A-FACAA](#) given in APPENDIX 02 - Elapsed Time Assumption .

NOTE: The minimum man-hours necessary to accomplish this Service Bulletin are those for the Inspection. Depending on the inspection result, additional man-hours can apply.

Task 571140-832-802-001: INSPECTION ADDITIONAL WORK	
Get Access	17.00
On Aircraft	
LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK	21.00
RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK	21.00
TOTAL MANHOURS	59.00
ELAPSED TIME (HOURS)	15.50

NOTE: For an explanation of the man-hours and elapsed time, refer to the Gantt Chart, [Fig. A-FADAA](#) given in APPENDIX 03 - Elapsed Time Assumption (ADDITIONAL WORK) .

Task 571140-833-802-001: REPAIR ADDITIONAL WORK	
On Aircraft	
Replace the Nuts on the LH Side, ADDITIONAL WORK	10.00
Replace the Nuts on the RH Side, ADDITIONAL WORK	10.00
Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side, ADDITIONAL WORK	13.00
Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side, ADDITIONAL WORK	13.00
Tests	5.00
Close-Up	14.50
TOTAL MANHOURS	65.50
ELAPSED TIME (HOURS)	20.50

NOTE: For an explanation of the man-hours and elapsed time, refer to the Gantt Chart, [Fig. A-FAEAA](#) given in APPENDIX 03 - Elapsed Time Assumption (ADDITIONAL WORK) .

NOTE: The minimum man-hours necessary to accomplish this Service Bulletin are those for the Inspection. Depending on the inspection result, additional man-hours can apply.

****CONF 002**

Task 571140-832-801-001: INSPECTION	
Get Access	17.00
On Aircraft	
LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction	26.00
RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction	26.00
TOTAL MANHOURS	69.00
ELAPSED TIME (HOURS)	18.00

NOTE: For an explanation of the man-hours and elapsed time, refer to the Gantt Chart, [Fig. A-FABAA](#) given in APPENDIX 02 - Elapsed Time Assumption .

Task 571140-833-801-001: REPAIR	
On Aircraft	
Replace the Nuts on the LH Side	12.00
Replace the Nuts on the RH Side	12.00
Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side	16.00
Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side	16.00
Tests	5.00
Close-Up	14.50
TOTAL MANHOURS	75.50
ELAPSED TIME (HOURS)	25.00

NOTE: For an explanation of the man-hours and elapsed time, refer to the Gantt Chart, [Fig. A-FACAA](#) given in APPENDIX 02 - Elapsed Time Assumption .

NOTE: The minimum man-hours necessary to accomplish this Service Bulletin are those for the Inspection. Depending on the inspection result, additional man-hours can apply.

Task 571140-832-802-001: INSPECTION ADDITIONAL WORK	
Get Access	17.00
On Aircraft	
LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK	10.00
RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK	10.00
TOTAL MANHOURS	37.00
ELAPSED TIME (HOURS)	10.00

NOTE: For an explanation of the man-hours and elapsed time, refer to the Gantt Chart, [Fig. A-FADAB](#) given in APPENDIX 03 - Elapsed Time Assumption

(ADDITIONAL WORK) .

Task 571140-833-802-001: REPAIR ADDITIONAL WORK	
On Aircraft	
Replace the Nuts on the LH Side, ADDITIONAL WORK	5.00
Replace the Nuts on the RH Side, ADDITIONAL WORK	5.00
Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side, ADDITIONAL WORK	6.00
Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side, ADDITIONAL WORK	6.00
Tests	5.00
Close-Up	14.50
TOTAL MANHOURS	41.50
ELAPSED TIME (HOURS)	14.50

NOTE: For an explanation of the man-hours and elapsed time, refer to the Gantt Chart, [Fig. A-FAEAB](#) given in APPENDIX 03 - Elapsed Time Assumption (ADDITIONAL WORK) .

NOTE: The minimum man-hours necessary to accomplish this Service Bulletin are those for the Inspection. Depending on the inspection result, additional man-hours can apply.

****CONF ALL****H. WEIGHT AND BALANCE**

Not Changed

I. ELECTRICAL LOAD DATA

Not Changed

J. REFERENCES****CONF 001**

Aircraft Maintenance Manual (AMM)	12-34-24	20-21-11	20-28-00
	28-00-00	28-10-00	28-11-00
	28-21-00	28-21-42	28-21-54
	28-25-00	32-12-00	53-35-11
	57-17-11	57-27-11	57-27-12
Consumable Material List (CML)			
Drawing	R57102001		
Elec. Std. Practices Manual (ESPM)	20-55-00		
In Service Information (ISI)	00.00.00179		
Non Destructive Test Manual (NTM)	51-10-01		
Standards Manual (SM)			

Structural Repair Manual (SRM)	51-24-00 51-75-12	51-42-11	51-43-00
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****CONF 002**

Aircraft Maintenance Manual (AMM)	12-34-24 28-00-00 28-21-00 28-25-00 57-17-11	20-21-11 28-10-00 28-21-42 32-12-00 57-27-11	20-28-00 28-11-00 28-21-54 53-35-11 57-27-12
Consumable Material List (CML)			
Drawing	R57102001		
Elec. Std. Practices Manual (ESPM)	20-55-00		
In Service Information (ISI)	00.00.00179		
Non Destructive Test Manual (NTM)	51-10-01		
Standards Manual (SM)			
Structural Repair Manual (SRM)	51-24-00 51-75-12	51-42-11	51-43-00

****CONF ALL**

K. PUBLICATION AFFECTED

None

L. INTERCHANGEABILITY/MIXABILITY

Not Applicable

M. SPARES

None

2. MATERIAL INFORMATION****CONF ALL****A. MATERIAL - PRICE AND AVAILABILITY****(1) Procurement Addresses**

Customers with aircraft shown in the effectivity of this Service Bulletin should send a purchase order to AIRBUS. Quote the number of this Service Bulletin. The address is :

Kit 571140A01R04 Kit 571140A02R04 Supplementary kit 571140A01S02 Supplementary kit 571140A02S02 Supplementary kit 571140A01S03 Supplementary kit 571140A02S03	AIRBUS Material, Logistics and Suppliers ATP II B, Hein-Sass-Weg 24 21129 HAMBURG GERMANY For ordering by internet: http://spares.airbus.com For ordering by E-mail: airbus.kit@airbus.com Telephone Hotline: +49 40 743 51660
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For standard hardware the purchase order shall be addressed to AIRBUS. Quote the number of this Service Bulletin, the MSN and, if applicable the Retrofit Information Letter (RIL) reference. Please send your request to:

Component COMPA01 Component COMPA03 Component COMPA04	AIRBUS Material, Logistics and Suppliers Standard Parts For ordering by internet: http://spares.airbus.com or via SPEC2000 For ordering by E-mail: ifd.spares@airbus.com Phone: +49 (0) 40 50 76 40 03
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(2) Price and Availability

Kit 571140A01R04	
Cost	3,850.00 USD
Availability	180 days calendar days from receipt of order

The availability given above is the standard lead time from the date of your purchase order. If items are required before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions. The latest valid price and lead time for ordering purposes shall be obtained through the AIRBUS Spares portal or through a request for quotation to airbus.kit@airbus.com.

Kit 571140A02R04	
Cost	3,850.00 USD
Availability	180 days calendar days from receipt of order

The availability given above is the standard lead time from the date of your purchase order. If items are required before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions. The latest valid price and lead time for ordering purposes shall be obtained through the AIRBUS Spares portal or through a request for quotation to airbus.kit@airbus.com.

Supplementary kit 571140A01S02	
Cost	2,320.00 USD
Availability	180 days calendar days from receipt of order

The availability given above is the standard lead time from the date of your purchase order. If items are required before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions. The latest valid price and lead time for ordering purposes shall be obtained through the AIRBUS Spares portal or through a request for quotation to airbus.kit@airbus.com.

Supplementary kit 571140A02S02	
Cost	2,130.00 USD
Availability	180 days calendar days from receipt of order

The availability given above is the standard lead time from the date of your purchase order. If items are required before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions. The latest valid price and lead time for ordering purposes shall be obtained through the AIRBUS Spares portal or through a request for quotation to airbus.kit@airbus.com.

Supplementary kit 571140A01S03	
Cost	1,930.00 USD
Availability	99 days calendar days from receipt of order

The availability given above is the standard lead time from the date of your purchase order. If items are required before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions. The latest valid price and lead time for ordering purposes shall be obtained through the AIRBUS Spares portal or through a request for quotation to airbus.kit@airbus.com.

Supplementary kit 571140A02S03	
Cost	1,930.00 USD
Availability	99 days calendar days from receipt of order

The availability given above is the standard lead time from the date of your purchase order. If items are required before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions. The latest valid price and lead time for ordering purposes shall be obtained through the AIRBUS Spares portal or through a request for quotation to airbus.kit@airbus.com.

Component COMPA01

Operator supplied standard hardware parts required to accomplish this Service Bulletin are to be supplied from the operator's stock.

Component COMPA03

Operator supplied standard hardware parts required to accomplish this Service Bulletin are to be supplied from the operator's stock.

Component COMPA04

Operator supplied standard hardware parts required to accomplish this Service Bulletin are to be supplied from the operator's stock.

B. INDUSTRY SUPPORT INFORMATION

AIRBUS will share the costs of accomplishment of this Service Bulletin by providing the kits at no charge for orders placed before Aug 30/19.

For the accomplishment of this Service Bulletin before Aug 30/19, AIRBUS will share the costs by crediting the operator supplied standard hardware parts. Claims for this allowance should be received no later than 90 days after completion of the last affected aircraft of your fleet.

AIRBUS will credit the man hours indicated in this Service Bulletin at the agreed in-house warranty labor rate for claims received no later than 90 days after completion of the last affected aircraft of your fleet, for the accomplishment of this Service Bulletin before Aug 30/19.

C. LIST OF COMPONENTS

This table shows the supplemental kits (S kits) needed to update the kits already held in your stock to the latest status.

Kit held by operator	Supplementary Kit required	Latest status of Kit
571140A01R01	571140A01S02 AND 571140A01S03	571140A01R04
571140A01R04	None	571140A01R04

This table shows the supplemental kits (S kits) needed to update the kits already held in your stock to the latest status.

Kit held by operator	Supplementary Kit required	Latest status of Kit
571140A02R01	571140A02S02 AND 571140A02S03	571140A02R04
571140A02R03	571140A02S03	571140A02R04

SERVICE BULLETIN

Kit held by operator	Supplementary Kit required	Latest status of Kit
571140A02R04	None	571140A02R04

Kit 571140A01R04

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
10	ASNA2532-7	34	NUT				
11	ASNA2531-7	6	NUT				
13	ASNA2532-8	16	NUT				
15	ASNA2531-6	2	NUT				
60	EN6115T2-3	14	BOLT				
61	ASNA2536-2	17	NUT				
62	EN6115T2-2	3	BOLT				
65	NAS1149D0332K	8	WASHER				
69	NAS1149F0332P	2	WASHER				
	NSA5372-616AX	2	WASHER				
	NSA5372-616BX	2	WASHER				
	NSA5372-616CX	2	WASHER				
	NSA5372-616DX	2	WASHER				
	NSA5372-616EX	2	WASHER				
	NSA5372-716AX	40	WASHER				
	NSA5372-716BX	40	WASHER				
	NSA5372-716CX	40	WASHER				
	NSA5372-716DX	40	WASHER				
	NSA5372-716EX	40	WASHER				
	NSA5372-816AX	16	WASHER				
	NSA5372-816BX	16	WASHER				
	NSA5372-816CX	16	WASHER				
	NSA5372-816DX	16	WASHER				
	NSA5372-816EX	16	WASHER				

NOTE:

If you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin, refer to Standards Manual (SM). The SM will give you the correct part number to part number relationship.

Kit 571140A02R04

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
10	ASNA2532-7	34	NUT				
11	ASNA2531-7	6	NUT				
13	ASNA2532-8	16	NUT				
15	ASNA2531-6	2	NUT				
60	EN6115T2-3	14	BOLT				
61	ASNA2536-2	17	NUT				
62	EN6115T2-2	3	BOLT				
65	NAS1149D0332K	8	WASHER				

SERVICE BULLETIN

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
69	NAS1149F0332P	2	WASHER				
	NSA5372-616AX	2	WASHER				
	NSA5372-616BX	2	WASHER				
	NSA5372-616CX	2	WASHER				
	NSA5372-616DX	2	WASHER				
	NSA5372-616EX	2	WASHER				
	NSA5372-716AX	40	WASHER				
	NSA5372-716BX	40	WASHER				
	NSA5372-716CX	40	WASHER				
	NSA5372-716DX	40	WASHER				
	NSA5372-716EX	40	WASHER				
	NSA5372-816AX	16	WASHER				
	NSA5372-816BX	16	WASHER				
	NSA5372-816CX	16	WASHER				
	NSA5372-816DX	16	WASHER				
	NSA5372-816EX	16	WASHER				

NOTE: If you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin, refer to Standards Manual (SM). The SM will give you the correct part number to part number relationship.

Supplementary kit 571140A01S02

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
10	ASNA2532-7	11	NUT				
11	ASNA2531-7	2	NUT				
13	ASNA2532-8	16	NUT				
15	ASNA2531-6	2	NUT				
60	EN6115T2-3	14	BOLT				
61	ASNA2536-2	17	NUT				
62	EN6115T2-2	3	BOLT				
65	NAS1149D0332K	8	WASHER				
69	NAS1149F0332P	2	WASHER				
	NSA5372-716AX	31	WASHER				
	NSA5372-716BX	31	WASHER				
	NSA5372-716CX	31	WASHER				
	NSA5372-716DX	31	WASHER				
	NSA5372-716EX	31	WASHER				

NOTE: If you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin, refer to Standards Manual (SM). The SM will give you the correct part number to part number relationship.

Supplementary kit 571140A01S03

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
	NSA5372-616AX	2	WASHER				
	NSA5372-616BX	2	WASHER				
	NSA5372-616CX	2	WASHER				
	NSA5372-616DX	2	WASHER				
	NSA5372-616EX	2	WASHER				
	NSA5372-816AX	16	WASHER				
	NSA5372-816BX	16	WASHER				
	NSA5372-816CX	16	WASHER				
	NSA5372-816DX	16	WASHER				
	NSA5372-816EX	16	WASHER				

NOTE: If you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin, refer to Standards Manual (SM). The SM will give you the correct part number to part number relationship.

Supplementary kit 571140A02S02

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
10	ASNA2532-7	7	NUT				
11	ASNA2531-7	2	NUT				
13	ASNA2532-8	16	NUT				
15	ASNA2531-6	2	NUT				
60	EN6115T2-3	14	BOLT				
61	ASNA2536-2	17	NUT				
62	EN6115T2-2	3	BOLT				
65	NAS1149D0332K	8	WASHER				
69	NAS1149F0332P	2	WASHER				
	NSA5372-716AX	27	WASHER				
	NSA5372-716BX	27	WASHER				
	NSA5372-716CX	27	WASHER				
	NSA5372-716DX	27	WASHER				
	NSA5372-716EX	27	WASHER				

NOTE: If you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin, refer to Standards Manual (SM). The SM will give you the correct part number to part number relationship.

Supplementary kit 571140A02S03

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
	NSA5372-616AX	2	WASHER				
	NSA5372-616BX	2	WASHER				

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
	NSA5372-616CX	2	WASHER				
	NSA5372-616DX	2	WASHER				
	NSA5372-616EX	2	WASHER				
	NSA5372-816AX	16	WASHER				
	NSA5372-816BX	16	WASHER				
	NSA5372-816CX	16	WASHER				
	NSA5372-816DX	16	WASHER				
	NSA5372-816EX	16	WASHER				

NOTE: If you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin, refer to Standards Manual (SM). The SM will give you the correct part number to part number relationship.

D. LIST OF MATERIALS - OPERATOR SUPPLIED

(1) Consumable Materials

Consumable CMLA01

ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Wash Primer - Structure	04CMA2	As required	
	Primer Polyurethane Paint - Corrosion Inhibiting	04EAC2	As required	
	Top Coat Polyurethane - External Structure	04JAA3	As required	
	Polysulfide Sealant-General Purpose Brushable	06AAA1	As required	
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	
	Non Aqueous Cleaner	08BAA9	As required	

(2) Components

NOTE: To accomplish this Service Bulletin it is necessary to use the expendable parts listed in the subsequent referenced tasks:

- AMM Task 28-21-42-400-001
- AMM Task 28-21-54-400-001
- AMM Task 53-35-11-400-001
- AMM Task 57-17-11-400-001
- AMM Task 57-27-11-400-001

- AMM Task 57-27-12-400-001.

Component COMPA01

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
11	ASNA2531-7	8	NUT				
31	ABS1418K8-13	4	SCREW				
32	ABS1418K8-14	28	SCREW				
33	ABS1418K8-18	32	SCREW				
34	ABS1418K8-19	36	SCREW				
35	ASNA2532-8	136	NUT				
36	ABS1418K8-21	8	SCREW				
37	ABS1418K8-26	8	SCREW				
38	ABS1418K8-17	24	SCREW				
43	ABS1418K8-25	8	SCREW				
44	ASNA2531-8	24	NUT				
73	ABS1418K9-15	8	SCREW				
74	ABS1418K9-16	32	SCREW				
75	ABS1418K9-17	24	SCREW				
76	ASNA2532-9	64	NUT				
78	ABS1418K8-20	8	SCREW				
80	ABS1418K7-23	8	SCREW				
82	ABS1418K8-22	4	SCREW				
	NSA5372-716AX	8	WASHER				
	NSA5372-716BX	8	WASHER				
	NSA5372-716EX	8	WASHER				
	NSA5372-816AX	160	WASHER				
	NSA5372-816BX	160	WASHER				
	NSA5372-816EX	160	WASHER				
	NSA5372-916AX	64	WASHER				
	NSA5372-916BX	64	WASHER				
	NSA5372-916EX	64	WASHER				

NOTE: If you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin, refer to Standards Manual (SM). The SM will give you the correct part number to part number relationship.

NOTE: The quantities given in the component table are for all the taperlocks replacement.

NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

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Component COMPA03

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
11	ASNA2531-7	4	NUT				
32	ABS1418K8-14	8	SCREW				
34	ABS1418K8-19	8	SCREW				
35	ASNA2532-8	22	NUT				
37	ABS1418K8-26	2	SCREW				
43	ABS1418K8-25	2	SCREW				
44	ASNA2531-8	4	NUT				
73	ABS1418K9-15	4	SCREW				
74	ABS1418K9-16	16	SCREW				
75	ABS1418K9-17	12	SCREW				
76	ASNA2532-9	32	NUT				
78	ABS1418K8-20	4	SCREW				
80	ABS1418K7-23	4	SCREW				
82	ABS1418K8-22	2	SCREW				
	NSA5372-716AX	4	WASHER				
	NSA5372-716BX	4	WASHER				
	NSA5372-716EX	4	WASHER				
	NSA5372-816AX	26	WASHER				
	NSA5372-816BX	26	WASHER				
	NSA5372-816EX	26	WASHER				
	NSA5372-916AX	32	WASHER				
	NSA5372-916BX	32	WASHER				
	NSA5372-916EX	32	WASHER				

NOTE: If you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin, refer to Standards Manual (SM). The SM will give you the correct part number to part number relationship.

NOTE: The quantities given in the component table are for all the taperlocks replacement.

NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

Component COMPA04

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
11	ASNA2531-7	4	NUT				
34	ABS1418K8-19	8	SCREW				
35	ASNA2532-8	14	NUT				
37	ABS1418K8-26	2	SCREW				

ITEM	NEW PART N°	QTY (UM)	KEYWORD	ITEM	OLD PART N°	INT	SEE NOTES
43	ABS1418K8-25	2	SCREW				
44	ASNA2531-8	4	NUT				
73	ABS1418K9-15	4	SCREW				
74	ABS1418K9-16	16	SCREW				
75	ABS1418K9-17	12	SCREW				
76	ASNA2532-9	32	NUT				
78	ABS1418K8-20	4	SCREW				
80	ABS1418K7-23	4	SCREW				
82	ABS1418K8-22	2	SCREW				
	NSA5372-716AX	4	WASHER				
	NSA5372-716BX	4	WASHER				
	NSA5372-716EX	4	WASHER				
	NSA5372-816AX	18	WASHER				
	NSA5372-816BX	18	WASHER				
	NSA5372-816EX	18	WASHER				
	NSA5372-916AX	32	WASHER				
	NSA5372-916BX	32	WASHER				
	NSA5372-916EX	32	WASHER				

NOTE: If you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin, refer to Standards Manual (SM). The SM will give you the correct part number to part number relationship.

NOTE: The quantities given in the component table are for all the taperlocks replacement.

NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

(3) Equipment

None

E. PARTS TO BE RE-IDENTIFIED BY OPERATOR

None

F. TOOLING

(1) Tooling - Price and Availability

(a) Procurement Addresses

Customers with aircraft shown in the effectivity of this Service Bulletin should send a purchase order to AIRBUS. Quote the number of this Service Bulletin. The address is :

Kit tool 571140T01R01 Kit tool 571140T02R00	AIRBUS Material, Logistics and Suppliers IFD - In Flight Desk Weg beim Jaeger 150 D-22335 HAMBURG GERMANY For ordering by internet: http://spares.airbus.com For ordering by fax: +49 40 50 76 4012 For ordering by E-mail: ifd.spares@airbus.com
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(b) Price and Availability

Kit tool 571140T01R01	
Cost	1,000.00 USD
Availability	42 calendar days from receipt of order

AIRBUS will provide at no charge the loan of this specific tool for a period equivalent to the Service Bulletin elapsed time (8 hours a day) per aircraft application plus ten days for transportation. After relevant period, standard loan charges apply.

Kit tool 571140T02R00	
Cost	1,364.00 USD
Availability	60 calendar days from receipt of order

AIRBUS will provide at no charge the loan of this specific tool for a period equivalent to the Service Bulletin elapsed time (8 hours a day) per aircraft application plus ten days for transportation. After relevant period, standard loan charges apply.

(2) Tools

(1) To accomplish this Service Bulletin it is necessary to use the following tools and equipment.

(a) For the taperlok installation test it is necessary to use the tool No. 98D57103001000 defined in Kit 571140T01R01 and the tool No. 98D57103002000 defined in Kit 571140T02R00.

- (b) In case of finding, for the installation of the new taperlok it is necessary to use the drilling tool SOA-TL-R57102001B.

Procurement to be negotiated directly with :

S.O.A
International Managing Support
66 rue des Graves, BP30013
33326 EYSINES Cedex
Tel : +33.5.56.57.94.50
Fax : +33.5.56.28.95.17
Email: direction@soaweb.fr
FRANCE

- (c) Equipment and materials for the rotating probe inspection of the holes, refer to NTM 51-10-01 Part 6.

- (d) To accomplish this Service Bulletin it is necessary to use the tool 98D57004014000 listed in:

- AMM Task 28-21-42-000-001
- AMM Task 28-21-54-000-001

- (e) To accomplish this Service Bulletin it is necessary to use the tools listed in the subsequent referenced tasks:

- AMM Task 57-27-11-000-001
- AMM Task 57-27-11-400-001
- AMM Task 57-27-12-000-001
- AMM Task 57-27-12-400-001.

None

(3) Special Tools

Kit tool 571140T01R01

ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103001000	Gage	8	

Kit tool 571140T02R00

ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103002000	Test	1	

3. ACCOMPLISHMENT INSTRUCTIONS

Task 571140-832-801-001 - INSPECTION

WARNING: MAKE SURE THAT YOU OBEY ALL THE WARNINGS AND ALL THE CAUTIONS INCLUDED IN THE REFERENCED PROCEDURES.

CAUTION: ALWAYS OBEY THE PRECAUTIONS THAT FOLLOW TO KEEP ELECTRICAL WIRING IN A SATISFACTORY CONDITION (ELECTRICALLY AND MECHANICALLY SERVICEABLE). WHEN YOU DO MAINTENANCE WORK, REPAIRS OR MODIFICATIONS, ALWAYS KEEP ELECTRICAL WIRING, COMPONENTS AND THE WORK AREA AS CLEAN AS POSSIBLE. TO DO THIS :

- PUT PROTECTION, SUCH AS PLASTIC SHEETING, CLOTHS, ETC; AS NECESSARY ON WIRING AND COMPONENTS.
- REGULARLY REMOVE ALL SHAVINGS, UNWANTED MATERIAL AND OTHER CONTAMINATION.

THESE PRECAUTIONS WILL DECREASE THE RISK OF CONTAMINATION AND DAMAGE TO THE ELECTRICAL WIRING INSTALLATION.

IF THERE IS CONTAMINATION REFER TO ESPM 20-55-00.

NOTE: The accomplishment instructions of this Service Bulletin include procedures given in other documents or in other sections of the Service Bulletin. When the words 'refer to' are used and the operator has a procedure accepted by the local authority he belongs to, the accepted alternative procedure can be used. When the words 'in accordance with' are used then the given procedure must be followed.

NOTE: The access and close-up instructions, not comprising return to service tests, in this Service Bulletin do not constitute or affect the technical intent of the Service Bulletin. Operators can therefore, as deemed necessary, omit or add access and/or close-up steps to add flexibility to their maintenance operations as long as the technical intent of the Service Bulletin is met within the set parameters.

NOTE: Manual titles given in the accomplishment instructions are referred to by acronyms. Refer to paragraph 1.J., References, for the definition of acronyms.

NOTE: Discarded parts must be managed at the operator's discretion.

NOTE: This Service Bulletin contains the instructions for the on-aircraft maintenance necessary to ensure the continued airworthiness of the aircraft.

For any deviations to the instructions, including RC paragraph contained herein, contact AIRBUS for further instructions and approval.

NOTE: The purpose of flowcharts is to supplement the information given in the Procedure and Compliance paragraphs and not to serve as the primary source for tasks or compliance times given in this Service Bulletin.

Task Associated Data****CONF 001**

Zone	
From 141 to 540	
From 142 to 640	
Access	
Panel	147AZ 148AZ 540AB 540BZ 640AB 640BZ
Manpower	
TOTAL MANHOURS	69.00
ELAPSED TIME (HOURS)	18.00

****CONF 002**

Zone	
From 142 to 640	
Access	
Panel	148AZ 640AB 640BZ
Manpower	
TOTAL MANHOURS	69.00
ELAPSED TIME (HOURS)	18.00

****CONF ALL****A. GENERAL******CONF 001****(1) Subtask 571140-910-001-001 - Standard Practices**

Manpower Resources	
Skills	NON SPECIFIC

References	
Consumable Material List (CML)	
Structural Repair Manual (SRM)	51-42-11

(a) For the specification of Material Numbers (Mat. No.) given in this Service Bulletin, refer to the Consumable Material List (CML).

(b) Remove and install fasteners in accordance with SRM 51-42-11.

(2) Subtask 571140-839-001-001 - Administrative

Manpower Resources	
Skills	NON SPECIFIC

Write in the applicable aircraft records that you have done all the work given in the Service Bulletin.

****CONF 002****(1) Subtask 571140-910-001-001 - Standard Practices**

Manpower Resources	
Skills	NON SPECIFIC

References	
Consumable Material List (CML)	
Structural Repair Manual (SRM)	51-42-11

- (a) For the specification of Material Numbers (Mat. No.) given in this Service Bulletin, refer to the Consumable Material List (CML).
- (b) Remove and install fasteners in accordance with SRM 51-42-11.

(2) Subtask 571140-839-001-001 - Administrative

Manpower Resources	
Skills	NON SPECIFIC

Write in the applicable aircraft records that you have done all the work given in the Service Bulletin.

****CONF ALL****B. PREPARATION******CONF 001****(1) Subtask 571140-941-001-001 - Job Set-up**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 12-34-24-869-002 Task 28-00-00-910-001 Task 28-10-00-910-004 Task 28-25-00-650-001 Task 32-12-00-010-001

NOTE: The items given in this note shall be considered as the basic Aircraft configuration before you start a maintenance task:

- Aircraft on the ground resting on landing gear (the ground safety locks and the wheel chocks are in position on the landing gear),
- Engine shut down, thrust reversers closed and locked,
- Aircraft in clean configuration,
- Parking brake applied,

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- Aircraft electrical network de-energized,
 - Hydraulic systems depressurized,
 - Access to the cockpit and cabin is available,
 - All circuit breakers are in closed position,
 - All controls in NORM, AUTO or OFF position.
- (a) Make sure that the aircraft is electrically grounded, refer to AMM Task 12-34-24-869-002.
- (b) Put an access platform in position at zones Z540 and Z640.
- (c) Defuel the center tank, refer to AMM Task 28-25-00-650-001.
- (d) Defuel the LH and RH wing tanks, refer to AMM Task 28-25-00-650-001.
- (e) Degas and ventilate the center tank and the wing tanks, refer to AMM Task 28-00-00-910-001 and AMM Task 28-10-00-910-004.
- (f) Open the main landing gear doors, refer to AMM Task 32-12-00-010-001.

(2) Subtask 571140-010-001-001 - Remove/Open for Access on the LH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-000-001 Task 28-21-54-000-001 Task 53-35-11-000-001 Task 57-17-11-000-001 Task 57-27-11-000-001 Task 57-27-12-000-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

- (a) Remove the access cover 147AZ, refer to AMM Task 57-17-11-000-001.
- (b) Remove the wing access panel 540AB, refer to AMM Task 57-27-11-000-001.
- (c) Open the fuel surge access door 540BZ, refer to AMM Task 57-27-12-000-001.
- (d) Remove the fuel pump suction valve, refer to AMM Task 28-21-42-000-001.
- (e) Remove the covers Item (80) and (81) of the fuel pump strainer FIN 6QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-000-001.
- (f) If necessary, remove the panels 191LB and 147CB on the wing to fuselage, refer to AMM Task 53-35-11-000-001.

(3) Subtask 571140-010-002-001 - Remove/Open for Access on the RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-000-001 Task 28-21-54-000-801 Task 53-35-11-000-001 Task 57-17-11-000-001 Task 57-27-11-000-001 Task 57-27-12-000-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

- (a) Remove the access cover 148AZ, refer to AMM Task 57-17-11-000-001.
- (b) Remove the wing access panel 640AB, refer to AMM Task 57-27-11-000-001.
- (c) Open the fuel surge access door 640BZ, refer to AMM Task 57-27-12-000-001.
- (d) Remove the fuel pump suction valve, refer to AMM Task 28-21-42-000-001.
- (e) Remove the covers Item (80) and (81) of the fuel pump strainer FIN 8QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-000-801.
- (f) If necessary, remove the panels 192LB and 148DB on the wing to fuselage, refer to AMM Task 53-35-11-000-001.

(4) Subtask 571140-010-005-001 - Remove the Fuel Strainer, LH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- Remove for access the fuel strainer for LH side:
Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)
In accordance with SRM 51-42-11.

1	Strainer Assembly	Item (55)	Retain
1	Strainer Assembly	Item (57)	Retain

attached with:

8	Bolt	Item (66)	Retain
2	Bolt	Item (67)	Retain
2	Bolt	Item (68)	Retain
2	Washer	Item (69)	Discard

(5) Subtask 571140-010-006-001 - Remove the Fuel Strainer, RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- Remove for access the fuel strainer for RH side:
Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)
In accordance with SRM 51-42-11.

1	Strainer Assembly	Item (56)	Retain
1	Strainer Assembly	Item (58)	Retain
attached with:			
8	Bolt	Item (66)	Retain
2	Bolt	Item (67)	Retain
2	Bolt	Item (68)	Retain
2	Washer	Item (69)	Discard

****CONF 002****(1) Subtask 571140-941-001-001 - Job Set-up**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 12-34-24-869-002
	Task 28-00-00-910-001
	Task 28-10-00-910-004
	Task 28-25-00-650-001
	Task 32-12-00-010-001

NOTE: The items given in this note shall be considered as the basic Aircraft configuration before you start a maintenance task:

- Aircraft on the ground resting on landing gear (the ground safety locks and the wheel chocks are in position on the landing gear),
- Engine shut down, thrust reversers closed and locked,
- Aircraft in clean configuration,
- Parking brake applied,
- Aircraft electrical network de-energized,
- Hydraulic systems depressurized,
- Access to the cockpit and cabin is available,
- All circuit breakers are in closed position,
- All controls in NORM, AUTO or OFF position.

- (a) Make sure that the aircraft is electrically grounded, refer to AMM Task 12-34-24-869-002.
- (b) Put an access platform in position at zones Z540 and Z640.
- (c) Defuel the center tank, refer to AMM Task 28-25-00-650-001.
- (d) Defuel the LH and RH wing tanks, refer to AMM Task 28-25-00-650-001.
- (e) Degas and ventilate the center tank and the wing tanks, refer to AMM Task 28-00-00-910-001 and AMM Task 28-10-00-910-004.
- (f) Open the main landing gear doors, refer to AMM Task 32-12-00-010-001.

(2) Subtask 571140-010-001-001 - Remove/Open for Access on the LH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-000-001 Task 28-21-54-000-001 Task 53-35-11-000-001 Task 57-17-11-000-001 Task 57-27-11-000-001 Task 57-27-12-000-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

- (a) Remove the access cover 147AZ, refer to AMM Task 57-17-11-000-001.
- (b) Remove the wing access panel 540AB, refer to AMM Task 57-27-11-000-001.
- (c) Open the fuel surge access door 540BZ, refer to AMM Task 57-27-12-000-001.
- (d) Remove the fuel pump suction valve, refer to AMM Task 28-21-42-000-001.
- (e) Remove the covers Item (80) and (81) of the fuel pump strainer FIN 6QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-000-001.
- (f) If necessary, remove the panels 191LB and 147CB on the wing to fuselage, refer to AMM Task 53-35-11-000-001.

(3) Subtask 571140-010-002-001 - Remove/Open for Access on the RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-000-001 Task 28-21-54-000-801 Task 53-35-11-000-001 Task 57-17-11-000-001 Task 57-27-11-000-001 Task 57-27-12-000-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

- (a) Remove the access cover 148AZ, refer to AMM Task 57-17-11-000-001.
- (b) Remove the wing access panel 640AB, refer to AMM Task 57-27-11-000-001.
- (c) Open the fuel surge access door 640BZ, refer to AMM Task 57-27-12-000-001.
- (d) Remove the fuel pump suction valve, refer to AMM Task 28-21-42-000-001.
- (e) Remove the covers Item (80) and (81) of the fuel pump strainer FIN 8QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-000-801.

- (f) If necessary, remove the panels 192LB and 148DB on the wing to fuselage, refer to AMM Task 53-35-11-000-001.

(4) Subtask 571140-010-005-001 - Remove the Fuel Strainer, LH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- Remove for access the fuel strainer for LH side:
Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)
In accordance with SRM 51-42-11.

1	Strainer Assembly	Item (55)	Retain
1	Strainer Assembly	Item (57)	Retain
attached with:			
8	Bolt	Item (66)	Retain
2	Bolt	Item (67)	Retain
2	Bolt	Item (68)	Retain
2	Washer	Item (69)	Discard

(5) Subtask 571140-010-006-001 - Remove the Fuel Strainer, RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- Remove for access the fuel strainer for RH side:
Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)
In accordance with SRM 51-42-11.

1	Strainer Assembly	Item (56)	Retain
1	Strainer Assembly	Item (58)	Retain
attached with:			
8	Bolt	Item (66)	Retain
2	Bolt	Item (67)	Retain
2	Bolt	Item (68)	Retain
2	Washer	Item (69)	Discard

****CONF ALL**

C. PROCEDURE

****CONF 001**

- (1) **Subtask 571140-832-001-001 - LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction**

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	26.00
Minimum number of person	2
Subtask elapsed time	13.00
Skills	AIRFRAME

Material necessary to do the job

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Non Aqueous Cleaner	08BAA9	As required	

Kit tool 571140T01R01				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103001000	Gage	1	

References	
In Service Information (ISI)	00.00.00179
Structural Repair Manual (SRM)	51-42-11
Fig. A-FBAAA Inspection of the Fasteners	Sheet 01 Sheet 02 Sheet 03
Fig. A-FBCAA Thread Damage	Sheet 01
Fig. A-FCAAA LH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FFAAA Flow Chart	Sheet 01

(a) At threshold given in the PLANNING INFORMATION Paragraph E.(2):

Do a detailed inspection of lower panels in center and outer wing box at level of rib1 junction:

In accordance with SRM 51-42-11.

Refer to [Fig. A-FFAAA Sheet 01](#)

Refer to [Fig. A-FBAAA Sheet 01](#) to [Fig. A-FBAAA Sheet 03](#)

Refer to [Fig. A-FBCAA Sheet 01](#)

Refer to [Fig. A-FCAAA Sheet 01](#) to [Fig. A-FCAAA Sheet 04](#)

1 Remove the sealant on the nuts with:

Non Aqueous Cleaner	08BAA9	As required
------------------------	--------	-------------

2 Remove the nuts :

Refer to [Fig. A-FCAAA Sheet 01](#) to [Fig. A-FCAAA Sheet 04](#)

34	Nut	NSA5457-7E	Item (10)	Discard
	or			
34	Nut	NAS1727-7E	Item (10)	Discard
	or			
34	Nut	NSA5151-7	Item (10)	Discard
	and			

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6	Nut	NAS1726-7E	Item (11)	Discard
	or			
6	Nut	NSA5050-7	Item (11)	Discard
	and			
16	Nut	NSA5457-8E	Item (13)	Discard
	or			
16	Nut	NAS1727-8E	Item (13)	Discard
	or			
16	Nut	NSA5151-8	Item (13)	Discard
	and			
2	Nut	NAS1726-6E	Item (15)	Discard
	or			
2	Nut	NSA5050-6	Item (15)	Discard

NOTE: Identify the location of each removed nut.

- (b) Check if the taperlok gage fit and move as described in the gage set procedure:

98D57103001000 Gage set 1

- 1 If taperlok gage fit and move:

- a Do a visual inspection of the thread to determine the cause of thread marks found:

<1> If thread marks are found caused by an incorrect installation:

<a> Apply SUBTASK 571140-833-003 001 Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

<2> If no thread marks are found caused by an incorrect installation but thread was damaged during the nut removal:

<a> Apply SUBTASK 571140-833-003 001 Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

<3> If no thread marks are found caused by an incorrect installation and thread was not damaged during the nut removal:

<a> Apply SUBTASK 571140-833-001 001 Replace the Nuts on the LH Side before next flight.

2 If the taperlok gage do not fit and move but thread was damaged during the nut removal:

a Apply SUBTASK 571140-833-003 001 Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

3 If the taperlok gage do not fit and move and thread was not damaged during the nut removal:

a Apply SUBTASK 571140-833-001 001 Replace the Nuts on the LH Side before next flight.

(c) Upload the completed Inspection Report sheet given in APPENDIX 04 - Inspection Report in accordance with ISI 00.00.00179.

NOTE: Uploading the Inspection Report sheet is not Required for Compliance.

(2) Subtask 571140-832-002-001 - RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	26.00
Minimum number of person	2
Subtask elapsed time	13.00
Skills	AIRFRAME

Material necessary to do the job

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Non Aqueous Cleaner	08BAA9	As required	

Kit tool 571140T01R01				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103001000	Gage	1	

References	
In Service Information (ISI)	00.00.00179
Structural Repair Manual (SRM)	51-42-11

References	
Fig. A-FBAAA Inspection of the Fasteners	Sheet 01 Sheet 02 Sheet 03
Fig. A-FBCAA Thread Damage	Sheet 01
Fig. A-FCBAA RH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FFAAA Flow Chart	Sheet 01

(a) At threshold given in the PLANNING INFORMATION Paragraph E.(2):

Do a detailed inspection of lower panels in center and outer wing box at level of rib1 junction:

In accordance with SRM 51-42-11.

Refer to [Fig. A-FFAAA Sheet 01](#)

Refer to [Fig. A-FBAAA Sheet 01](#) to [Fig. A-FBAAA Sheet 03](#)

Refer to [Fig. A-FBCAA Sheet 01](#)

Refer to [Fig. A-FCBAA Sheet 01](#) to [Fig. A-FCBAA Sheet 04](#)

1 Remove the sealant on the nuts with:

Non Aqueous	08BAA9	As required
Cleaner		

2 Remove the nuts :

Refer to [Fig. A-FCBAA Sheet 01](#) to [Fig. A-FCBAA Sheet 04](#)

34	Nut	NSA5457-7E	Item (10)	Discard
	or			
34	Nut	NAS1727-7E	Item (10)	Discard
	or			
34	Nut	NSA5151-7	Item (10)	Discard
	and			
6	Nut	NAS1726-7E	Item (11)	Discard

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	or			
6	Nut	NSA5050-7	Item (11)	Discard
	and			
16	Nut	NSA5457-8E	Item (13)	Discard
	or			
16	Nut	NAS1727-8E	Item (13)	Discard
	or			
16	Nut	NSA5151-8	Item (13)	Discard
	and			
2	Nut	NAS1726-6E	Item (15)	Discard
	or			
2	Nut	NSA5050-6	Item (15)	Discard

NOTE: Identify the location of each removed nut.

(b) Check if the taperlok gage fit and move as described in the gage set procedure:

98D57103001000 Gage set 1

1 If taperlok gage fit and move:

a Do a visual inspection of the thread to determine the cause of thread marks found:

<1> If thread marks are found caused by an incorrect installation:

<a> Apply SUBTASK 57-1140-833-004 001 Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

<2> If no thread marks are found caused by an incorrect installation but thread was damaged during the nut removal:

<a> Apply SUBTASK 57-1140-833-004 001 Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

<3> If no thread marks are found caused by an incorrect installation and thread was not damaged during the nut removal:

<a> Apply SUBTASK 57-1140-833-002 001 Replace the Replace the Nuts on the RH Side before next flight.

2 If the taperlok gage do not fit and move but thread was damaged during the nut removal:

a Apply SUBTASK 57-1140-833-004 001 Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

3 If the taperlok gage do not fit and move and thread was not damaged during the nut removal:

a Apply SUBTASK 57-1140-833-002 001 Replace the Replace the Nuts on the RH Side before next flight.

(c) Upload the completed Inspection Report sheet given in APPENDIX 04 - Inspection Report in accordance with ISI 00.00.00179.

NOTE: Uploading the Inspection Report sheet is not Required for Compliance.

****CONF 002**

(1) Subtask 571140-832-001-002 - LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	26.00
Minimum number of person	2
Subtask elapsed time	13.00
Skills	AIRFRAME

Material necessary to do the job

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Non Aqueous Cleaner	08BAA9	As required	

Kit tool 571140T01R01				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103001000	Gage	1	

References	
In Service Information (ISI)	00.00.00179

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FFAAB Flow Chart	Sheet 01
Fig. A-FBAAA Inspection of the Fasteners	Sheet 01 Sheet 02 Sheet 03
Fig. A-FBCAA Thread Damage	Sheet 01
Fig. A-FCAAA LH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

(a) At threshold given in the PLANNING INFORMATION Paragraph E.(2):

Do a detailed inspection of lower panels in center and outer wing box at level of rib1 junction:

In accordance with SRM 51-42-11.

Refer to [Fig. A-FFAAB Sheet 01](#)

Refer to [Fig. A-FBAAA Sheet 01](#) to [Fig. A-FBAAA Sheet 03](#)

Refer to [Fig. A-FBCAA Sheet 01](#)

Refer to [Fig. A-FCAAA Sheet 01](#) to [Fig. A-FCAAA Sheet 04](#)

1 Remove the sealant on the nuts with:

Non Aqueous Cleaner	08BAA9	As required
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2 Remove the nuts :

Refer to [Fig. A-FCAAA Sheet 01](#) to [Fig. A-FCAAA Sheet 04](#)

34	Nut	NSA5457-7E	Item (10)	Discard
	or			
34	Nut	NAS1727-7E	Item (10)	Discard
	or			
34	Nut	NSA5151-7	Item (10)	Discard
	and			

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6	Nut	NAS1726-7E	Item (11)	Discard
	or			
6	Nut	NSA5050-7	Item (11)	Discard
	and			
16	Nut	NSA5457-8E	Item (13)	Discard
	or			
16	Nut	NAS1727-8E	Item (13)	Discard
	or			
16	Nut	NSA5151-8	Item (13)	Discard
	and			
2	Nut	NAS1726-6E	Item (15)	Discard
	or			
2	Nut	NSA5050-6	Item (15)	Discard

NOTE: Identify the location of each removed nut.

- (b) Check if the taperlok gage fit and move as described in the gage set procedure:

98D57103001000 Gage set 1

- 1 If taperlok gage fit and move:

- a Do a visual inspection of the thread to determine the cause of thread marks found:

<1> If thread marks are found caused by an incorrect installation:

<a> Apply SUBTASK 571140-833-003 002 Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

<2> If no thread marks are found caused by an incorrect installation but thread was damaged during the nut removal:

<a> Apply SUBTASK 571140-833-003 002 Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

<3> If no thread marks are found caused by an incorrect installation and thread was not damaged during the nut removal:

<a> Apply SUBTASK 571140-833-001 002 Replace the Nuts on the LH Side before next flight.

- 2 If the taperlok gage do not fit and move but thread was damaged during the nut removal:
 - a Apply SUBTASK 571140-833-003 002 Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.
- 3 If the taperlok gage do not fit and move and thread was not damaged during the nut removal:
 - a Apply SUBTASK 571140-833-001 002 Replace the Nuts on the LH Side before next flight.

- (c) Upload the completed Inspection Report sheet given in APPENDIX 04 - Inspection Report in accordance with ISI 00.00.00179.

NOTE: Uploading the Inspection Report sheet is not Required for Compliance.

(2) Subtask 571140-832-002-002 - RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction

Manpower Resources	
Manhours	26.00
Minimum number of person	2
Subtask elapsed time	13.00
Skills	AIRFRAME

Material necessary to do the job

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Non Aqueous Cleaner	08BAA9	As required	

Kit tool 571140T01R01				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103001000	Gage	1	

References	
In Service Information (ISI)	00.00.00179
Structural Repair Manual (SRM)	51-42-11
Fig. A-FBAAA Inspection of the Fasteners	Sheet 01 Sheet 02 Sheet 03
Fig. A-FBCAA Thread Damage	Sheet 01

References	
Fig. A-FCBAA RH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FCAAA LH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FFAAB Flow Chart	Sheet 01

(a) At threshold given in the PLANNING INFORMATION Paragraph E.(2):

Do a detailed inspection of lower panels in center and outer wing box at level of rib1 junction:

In accordance with SRM 51-42-11.

Refer to [Fig. A-FFAAB Sheet 01](#)

Refer to [Fig. A-FBAAA Sheet 01](#) to [Fig. A-FBAAA Sheet 03](#)

Refer to [Fig. A-FBCAA Sheet 01](#)

Refer to [Fig. A-FCBAA Sheet 01](#) to [Fig. A-FCBAA Sheet 04](#)

1 Remove the sealant on the nuts with:

Non Aqueous	08BAA9	As required
Cleaner		

2 Remove the nuts :

Refer to [Fig. A-FCBAA Sheet 01](#) to [Fig. A-FCBAA Sheet 04](#)

34	Nut	NSA5457-7E	Item (10)	Discard
	or			
34	Nut	NAS1727-7E	Item (10)	Discard
	or			
34	Nut	NSA5151-7	Item (10)	Discard
	and			

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6	Nut	NAS1726-7E	Item (11)	Discard
	or			
6	Nut	NSA5050-7	Item (11)	Discard
	and			
16	Nut	NSA5457-8E	Item (13)	Discard
	or			
16	Nut	NAS1727-8E	Item (13)	Discard
	or			
16	Nut	NSA5151-8	Item (13)	Discard
	and			
2	Nut	NAS1726-6E	Item (15)	Discard
	or			
2	Nut	NSA5050-6	Item (15)	Discard

NOTE: Identify the location of each removed nut.

- (b) Check if the taperlok gage fit and move as described in the gage set procedure:

98D57103001000 Gage set 1

- 1 If taperlok gage fit and move:

- a Do a visual inspection of the thread to determine the cause of thread marks found:

<1> If thread marks are found caused by an incorrect installation:

<a> Apply SUBTASK 57-1140-833-004 002 Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

<2> If no thread marks are found caused by an incorrect installation but thread was damaged during the nut removal:

<a> Apply SUBTASK 57-1140-833-004 002 Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

<3> If no thread marks are found caused by an incorrect installation and thread was not damaged during the nut removal:

<a> Apply SUBTASK 57-1140-833-002 002 Replace the Replace the Nuts on the RH Side before next flight.

- 2 If the taperlok gage do not fit and move but thread was damaged during the nut removal:
- a Apply SUBTASK 57-1140-833-004 002 Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.
- 3 If the taperlok gage do not fit and move and thread was not damaged during the nut removal:
- a Apply SUBTASK 57-1140-833-002 002 Replace the Replace the Nuts on the RH Side before next flight.

- (c) Upload the completed Inspection Report sheet given in APPENDIX 04 - Inspection Report in accordance with ISI 00.00.00179.

NOTE: Uploading the Inspection Report sheet is not Required for Compliance.

****CONF ALL**

D. TEST

****CONF 001**

None

****CONF 002**

None

****CONF ALL**

E. CLOSE-UP

****CONF 001**

None

****CONF 002**

None

Task 571140-833-801-001 - REPAIR

****CONF ALL**

WARNING: MAKE SURE THAT YOU OBEY ALL THE WARNINGS AND ALL THE CAUTIONS INCLUDED IN THE REFERENCED PROCEDURES.

CAUTION: ALWAYS OBEY THE PRECAUTIONS THAT FOLLOW TO KEEP ELECTRICAL WIRING IN A SATISFACTORY CONDITION (ELECTRICALLY AND MECHANICALLY SERVICEABLE). WHEN YOU DO MAINTENANCE WORK, REPAIRS OR MODIFICATIONS, ALWAYS KEEP ELECTRICAL WIRING, COMPONENTS AND THE WORK AREA AS CLEAN AS POSSIBLE. TO DO THIS :

- PUT PROTECTION, SUCH AS PLASTIC SHEETING, CLOTHS, ETC; AS NECESSARY ON WIRING AND COMPONENTS.
- REGULARLY REMOVE ALL SHAVINGS, UNWANTED MATERIAL AND OTHER CONTAMINATION.

THESE PRECAUTIONS WILL DECREASE THE RISK OF CONTAMINATION AND DAMAGE TO THE ELECTRICAL WIRING INSTALLATION.

IN CASE OF CONTAMINATION REFER TO ESPM 20-55-00.

NOTE: The accomplishment instructions of this Service Bulletin include procedures given in other documents or in other sections of the Service Bulletin. When the words 'refer to' are used and the operator has a procedure accepted by the local authority he belongs to, the accepted alternative procedure can be used. When the words 'in accordance with' are used then the given procedure must be followed.

NOTE: The access and close-up instructions, not comprising return to service tests, in this Service Bulletin do not constitute or affect the technical intent of the Service Bulletin. Operators can therefore, as deemed necessary, omit or add access and/or close-up steps to add flexibility to their maintenance operations as long as the technical intent of the Service Bulletin is met within the set parameters.

NOTE: Manual titles given in the accomplishment instructions are referred to by acronyms. Refer to paragraph 1.J., References, for the definition of acronyms.

NOTE: This Service Bulletin contains the instructions for the on-aircraft maintenance necessary to ensure the continued airworthiness of the aircraft.

For any deviations to the instructions contained herein, contact AIRBUS for further instructions and approval.

Task Associated Data

****CONF 001**

Zone
From 141 to 540
From 142 to 640

Access	
Panel	147AZ 148AZ 540AB 540BZ 640AB 640BZ
Manpower	
TOTAL MANHOURS	75.50
ELAPSED TIME (HOURS)	25.00

****CONF 002**

Zone	
From 141 to 540 From 142 to 640	
Access	
Panel	147AZ 148AZ 540AB 540BZ 640AB 640BZ
Manpower	
TOTAL MANHOURS	75.50
ELAPSED TIME (HOURS)	25.00

****CONF ALL****A. GENERAL******CONF 001****(1) Subtask 571140-910-002-001 - Standard Practices**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Consumable Material List (CML)	
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00 51-75-12

- (a) Apply the sealant in accordance with SRM 51-24-00.
- (b) For the specification of Material Numbers (Mat. No.) given in this Service Bulletin, refer to the Consumable Material List (CML).
- (c) Remove and install fasteners in accordance with SRM 51-42-11.
- (d) Get alternative and substitute fastener data in accordance with SRM 51-43-00.
- (e) Renew the protective finish in accordance with SRM 51-75-12.
- (f) To torque tighten the standard threaded fasteners, refer to AMM Task 20-21-11-911-001.

(2) Subtask 571140-839-002-001 - Administrative

Manpower Resources	
Skills	NON SPECIFIC

Write in the applicable aircraft records that you have done all the work given in the Service Bulletin.

****CONF 002****(1) Subtask 571140-910-002-001 - Standard Practices**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Consumable Material List (CML)	
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00 51-75-12

- (a) Apply the sealant in accordance with SRM 51-24-00.
- (b) For the specification of Material Numbers (Mat. No.) given in this Service Bulletin, refer to the Consumable Material List (CML).
- (c) Remove and install fasteners in accordance with SRM 51-42-11.
- (d) Get alternative and substitute fastener data in accordance with SRM 51-43-00.
- (e) Renew the protective finish in accordance with SRM 51-75-12.
- (f) To torque tighten the standard threaded fasteners, refer to AMM Task 20-21-11-911-001.

(2) Subtask 571140-839-002-001 - Administrative

Manpower Resources	
Skills	NON SPECIFIC

Write in the applicable aircraft records that you have done all the work given in the Service Bulletin.

****CONF ALL****B. PREPARATION******CONF 001**

None

****CONF 002**

None

****CONF ALL**

C. PROCEDURE

****CONF 001**

(1) Subtask 571140-833-001-001 - Replace the Nuts on the LH Side

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	12.00
Minimum number of person	2
Subtask elapsed time	6.00
Skills	AIRFRAME

Material necessary to do the job

Kit 571140A01R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
10	ASNA2532-7	34	NUT	
11	ASNA2531-7	6	NUT	
13	ASNA2532-8	16	NUT	
15	ASNA2531-6	2	NUT	
	NSA5372-616AX	2	WASHER	
	NSA5372-616BX	2	WASHER	
	NSA5372-616CX	2	WASHER	
	NSA5372-616DX	2	WASHER	
	NSA5372-616EX	2	WASHER	
	NSA5372-716AX	40	WASHER	
	NSA5372-716BX	40	WASHER	
	NSA5372-716CX	40	WASHER	
	NSA5372-716DX	40	WASHER	
	NSA5372-716EX	40	WASHER	
	NSA5372-816AX	16	WASHER	
	NSA5372-816BX	16	WASHER	
	NSA5372-816CX	16	WASHER	
	NSA5372-816DX	16	WASHER	
	NSA5372-816EX	16	WASHER	

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11 51-43-00
Fig. A-FCAAA LH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FCCAA Definition of the Washer Thickness for the Original Taperlok	Sheet 01 Sheet 02 Sheet 03

NOTE: The nuts are removed during the inspection.

(a) Install the nuts.

In accordance with SRM 51-42-11 and SRM 51-43-00.

Refer to AMM Task 20-21-11-911-001.

Refer to [Fig. A-FCAAA Sheet 01](#), [Fig. A-FCAAA Sheet 02](#), [Fig. A-FCAAA Sheet 05](#), [Fig. A-FCAAA Sheet 06](#), [Fig. A-FCAAA Sheet 07](#)

34 Nut ASNA2532-7 Item 10

6 Nut ASNA2531-7 Item 11

with

40 Washer NSA5372-716AX

and/or

40 Washer NSA5372-716BX

and/or

40 Washer NSA5372-716CX

and/or

40 Washer NSA5372-716DX

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		and/or		
	40	Washer	NSA5372-716EX	
		and		
	2	Nut	ASNA2531-6	Item 15
		with		
	2	Washer	NSA5372-616AX	
		and/or		
	2	Washer	NSA5372-616BX	
		and/or		
	2	Washer	NSA5372-616CX	
		and/or		
	2	Washer	NSA5372-616DX	
		and/or		
	2	Washer	NSA5372-616EX	
		and		
	16	Nut	ASNA2532-8	Item 13
		with		
	16	Washer	NSA5372-816AX	
		and/or		
	16	Washer	NSA5372-816BX	
		and/or		
	16	Washer	NSA5372-816CX	
		and/or		
	16	Washer	NSA5372-816DX	
		and/or		
	16	Washer	NSA5372-816EX	

NOTE: Install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCCAA Sheet 01](#).

NOTE: If necessary, the installation of the three washers is approved in this condition.

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NOTE: Torque the nuts between 3.80daN and 4.60daN (342lbf.in. and 411lbf.in.).

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(2) Subtask 571140-833-002-001 - Replace the Nuts on the RH Side

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	12.00
Minimum number of person	2
Subtask elapsed time	6.00
Skills	AIRFRAME

Material necessary to do the job

Kit 571140A01R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
10	ASNA2532-7	34	NUT	
11	ASNA2531-7	6	NUT	
13	ASNA2532-8	16	NUT	
15	ASNA2531-6	2	NUT	
	NSA5372-616AX	2	WASHER	
	NSA5372-616BX	2	WASHER	
	NSA5372-616CX	2	WASHER	
	NSA5372-616DX	2	WASHER	
	NSA5372-616EX	2	WASHER	
	NSA5372-716AX	40	WASHER	
	NSA5372-716BX	40	WASHER	
	NSA5372-716CX	40	WASHER	
	NSA5372-716DX	40	WASHER	
	NSA5372-716EX	40	WASHER	
	NSA5372-816AX	16	WASHER	
	NSA5372-816BX	16	WASHER	
	NSA5372-816CX	16	WASHER	

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ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
	NSA5372-816DX	16	WASHER	
	NSA5372-816EX	16	WASHER	

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11 51-43-00
Fig. A-FCCAA Definition of the Washer Thickness for the Original Taperlok	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCBAA RH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

NOTE: The nuts are removed during the inspection.

(a) Install the nuts.

In accordance with SRM 51-42-11 and SRM 51-43-00.

Refer to AMM Task 20-21-11-911-001.

Refer to [Fig. A-FCBAA Sheet 01](#), [Fig. A-FCBAA Sheet 02](#), [Fig. A-FCBAA Sheet 05](#), [Fig. A-FCBAA Sheet 06](#), [Fig. A-FCBAA Sheet 07](#)

34	Nut	ASNA2532-7	Item 10
6	Nut	ASNA2531-7	Item 11
	with		
40	Washer	NSA5372-716AX	
	and/or		
40	Washer	NSA5372-716BX	
	and/or		

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40	Washer	NSA5372-716CX	
	and/or		
40	Washer	NSA5372-716DX	
	and/or		
40	Washer	NSA5372-716EX	
	and		
2	Nut	ASNA2531-6	Item 15
	with		
2	Washer	NSA5372-616AX	
	and/or		
2	Washer	NSA5372-616BX	
	and/or		
2	Washer	NSA5372-616CX	
	and/or		
2	Washer	NSA5372-616DX	
	and/or		
2	Washer	NSA5372-616EX	
	and		
16	Nut	ASNA2532-8	Item 13
	with		
16	Washer	NSA5372-816AX	
	and/or		
16	Washer	NSA5372-816BX	
	and/or		
16	Washer	NSA5372-816CX	
	and/or		
16	Washer	NSA5372-816DX	
	and/or		
16	Washer	NSA5372-816EX	

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NOTE: Install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCCAA Sheet 01](#).

NOTE: If necessary, the installation of the three washers is approved in this condition.

NOTE: Torque the nuts between 3.80daN and 4.60daN (342lbf.in. and 411lbf.in.).

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(3) Subtask 571140-833-003-001 - Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	16.00
Minimum number of person	2
Subtask elapsed time	10.00
Skills	AIRFRAME NON DESTRUCTIVE TESTING

Material necessary to do the job

Kit 571140A01R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
60	EN6115T2-3	14	BOLT	
61	ASNA2536-2	17	NUT	
62	EN6115T2-2	3	BOLT	
65	NAS1149D0332K	8	WASHER	

Component COMPA01				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
11	ASNA2531-7	2	NUT	

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ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
31	ABS1418K8-13	1	SCREW	
32	ABS1418K8-14	7	SCREW	
33	ABS1418K8-18	8	SCREW	
34	ABS1418K8-19	9	SCREW	
35	ASNA2532-8	34	NUT	
36	ABS1418K8-21	2	SCREW	
37	ABS1418K8-26	2	SCREW	
38	ABS1418K8-17	6	SCREW	
43	ABS1418K8-25	2	SCREW	
44	ASNA2531-8	6	NUT	
73	ABS1418K9-15	2	SCREW	
74	ABS1418K9-16	8	SCREW	
75	ABS1418K9-17	6	SCREW	
76	ASNA2532-9	16	NUT	
78	ABS1418K8-20	2	SCREW	
80	ABS1418K7-23	2	SCREW	
82	ABS1418K8-22	1	SCREW	
	NSA5372-716AX	2	WASHER	
	NSA5372-716BX	2	WASHER	
	NSA5372-716EX	2	WASHER	
	NSA5372-816AX	40	WASHER	
	NSA5372-816BX	40	WASHER	
	NSA5372-816EX	40	WASHER	
	NSA5372-916AX	16	WASHER	
	NSA5372-916BX	16	WASHER	
	NSA5372-916EX	16	WASHER	

NOTE: The quantities given in the component table are for all the taperlocks replacement.

NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Wash Primer - Structure	04CMA2	As required	
	Primer Polyurethane Paint - Corrosion Inhibiting	04EAC2	As required	
	Top Coat Polyurethane - External Structure	04JAA3	As required	
	Polysulfide Sealant-General Purpose Brushable	06AAA1	As required	

Kit tool 571140T02R00				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103002000	Test	1	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Drawing	R57102001
Non Destructive Test Manual (NTM)	51-10-01 PART 6
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00
Fig. A-FCAAA LH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FCDA Definition of the Washer Thickness for the New Taperlok	Sheet 01
Fig. A-FDAAA Removal/Installation of the Cooling Duct	Sheet 01
Fig. A-FDBAA Removal/Installation of the Belly Fairing Structure	Sheet 01 Sheet 02 Sheet 03

Refer to Fig. A-FDAAA Sheet 01

1

Retain

1	Duct	Item (52)	Retain
	attached with		
1	Clamp	Item (63)	Retain
8	Screw	Item (64)	Retain
8	Washer	Item (65)	Discard

2 Remove the structure of the belly fairing.

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (51)	Retain
	attached with		
3	Rivet	Item (62)	Discard
7	Rivet	Item (60)	Discard
10	Bush	Item (61)	Discard
	and/or		
1	Web 4 assy	Item (50)	Retain
	attached with		
7	Rivet	Item (60)	Discard
7	Bush	Item (61)	Discard

(b) In accordance with the result of the inspection, remove the taperlok(s) :

Refer to [Fig. A-FCAAA Sheet 01](#) to [Fig. A-FCAAA Sheet 04](#)

6	Screw	Item (16)	Discard
10	Screw	Item (5)	Discard
9	Screw	Item (7)	Discard
7	Screw	Item (3)	Discard
2	Screw	Item (14)	Discard
1	Screw	Item (1)	Discard
2	Screw	Item (12)	Discard

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2	Screw	Item (70)	Discard
8	Screw	Item (71)	Discard
6	Screw	Item (72)	Discard
2	Screw	Item (77)	Discard
2	Screw	Item (79)	Discard
1	Screw	Item (81)	Discard

NOTE: The nuts are removed during the inspection.

- (c) Do a rototest inspection of the holes for cracks, refer to NTM 51-10-01 PART 6.

1 If cracks are found, contact AIRBUS before any further work.

2 If no crack is found, do the following steps.

- (d) Oversize the hole to the next nominal diameter in accordance with the drilling procedure given in the Repair Instruction Drawing R57102001 and use the drilling tool SOA-TL-R57102001B.

NOTE: To prevent any accidental damage, we recommend to train to oversize the taperlok holes on the test specimen.

98D57103002000 Test specimen 1

- (e) Depending on the removal, install :

Refer to [Fig. A-FCAAA Sheet 01](#), [Fig. A-FCAAA Sheet 02](#), [Fig. A-FCAAA Sheet 05](#),
[Fig. A-FCAAA Sheet 06](#), [Fig. A-FCAAA Sheet 07](#)

6	Screw	ABS1418K8-17	Item 38
8	Screw	ABS1418K8-18	Item 33
9	Screw	ABS1418K8-19	Item 34
1	Screw	ABS1418K8-13	Item 31
7	Screw	ABS1418K8-14	Item 32
2	Screw	ABS1418K8-20	Item 78
1	Screw	ABS1418K8-22	Item 82
	with		
34	Nut	ASNA2532-8	Item 35

	and if necessary		
34	Washer	NSA5372-816AX	
	and/or		
34	Washer	NSA5372-816EX	
	and/or		
34	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K8-25	Item 43
2	Screw	ABS1418K8-26	Item 37
2	Screw	ABS1418K8-21	Item 36
	with		
6	Nut	ASNA2531-8	Item 44
	and if necessary		
6	Washer	NSA5372-816AX	
	and/or		
6	Washer	NSA5372-816EX	
	and/or		
6	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K9-15	Item 73
8	Screw	ABS1418K9-16	Item 74
6	Screw	ABS1418K9-17	Item 75
	with		
16	Nut	ASNA2532-9	Item 76
	and if necessary		
16	Washer	NSA5372-916AX	
	and/or		
16	Washer	NSA5372-916EX	

	and/or		
16	Washer	NSA5372-916BX	
	and		
2	Screw	ABS1418K7-23	Item 80
	with		
2	Nut	ASNA2531-7	Item 11
	and if necessary		
2	Washer	NSA5372-716AX	
	and/or		
2	Washer	NSA5372-716EX	
	and/or		
2	Washer	NSA5372-716BX	

NOTE: Install the screw in accordance with the installation principle given in the repair instruction R57102001.

NOTE: If necessary, install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCDA Sheet 01](#).

NOTE: Wet assembly, apply :

Polysulfide Sealant- 06ABB1	As required
Fuel Tank Fillet	

NOTE: Torque the nuts between 5.60daN and 6.70daN (493lbf.in. and 592lbf.in.).

NOTE: Outside the wing center box, apply on the screws :

Primer	04EAC2	As required
Polyurethane Paint		
- Corrosion		
Inhibiting		

Wash Primer -	04CMA2	As required
Structure		

Top Coat	04JAA3	As required
Polyurethane -		
External Structure		

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(f) If removed, install :

1 The structure of the belly fairing

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (51)	Retained at removal
---	----------------	-----------	------------------------

with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

3	Bolt	EN6115T2-2	Item 62
---	------	------------	---------

10	Nut	ASNA2536-2	Item 61
----	-----	------------	---------

and/or

1	Web 4 assy	Item (50)	Retained at removal
---	------------	-----------	------------------------

attached with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

7	Nut	ASNA2536-2	Item 61
---	-----	------------	---------

NOTE: Apply to the interface of the structure and all the parts :

Polysulfide Sealant- 06AAA1 As required
General Purpose
Brushable

2 The cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (53)	Retained at removal
---	------	-----------	------------------------

1	Duct	Item (52)	Retained at removal
---	------	-----------	------------------------

with

1	Clamp	Item (63)	Retained at removal
8	Screw	Item (64)	Retained at removal
8	Washer	NAS1149D0332K	Item 65

(4) Subtask 571140-833-004-001 - Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	16.00
Minimum number of person	2
Subtask elapsed time	10.00
Skills	AIRFRAME NON DESTRUCTIVE TESTING

Material necessary to do the job

Kit 571140A01R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
60	EN6115T2-3	14	BOLT	
61	ASNA2536-2	17	NUT	
62	EN6115T2-2	3	BOLT	
65	NAS1149D0332K	8	WASHER	

Component COMPA01				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
11	ASNA2531-7	2	NUT	
31	ABS1418K8-13	1	SCREW	
32	ABS1418K8-14	7	SCREW	
33	ABS1418K8-18	8	SCREW	
34	ABS1418K8-19	9	SCREW	
35	ASNA2532-8	34	NUT	
36	ABS1418K8-21	2	SCREW	
37	ABS1418K8-26	2	SCREW	
38	ABS1418K8-17	6	SCREW	
43	ABS1418K8-25	2	SCREW	

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ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
44	ASNA2531-8	6	NUIT	
73	ABS1418K9-15	2	SCREW	
74	ABS1418K9-16	8	SCREW	
75	ABS1418K9-17	6	SCREW	
76	ASNA2532-9	16	NUT	
78	ABS1418K8-20	2	SCREW	
80	ABS1418K7-23	2	SCREW	
82	ABS1418K8-22	1	SCREW	
	NSA5372-716AX	2	WASHER	
	NSA5372-716BX	2	WASHER	
	NSA5372-716EX	2	WASHER	
	NSA5372-816AX	40	WASHER	
	NSA5372-816BX	40	WASHER	
	NSA5372-816EX	40	WASHER	
	NSA5372-916AX	16	WASHER	
	NSA5372-916BX	16	WASHER	
	NSA5372-916EX	16	WASHER	

NOTE: The quantities given in the component table are for all the taperlocks replacement.

NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Wash Primer - Structure	04CMA2	As required	
	Primer Polyurethane Paint - Corrosion Inhibiting	04EAC2	As required	
	Top Coat Polyurethane - External Structure	04JAA3	As required	
	Polysulfide Sealant-General Purpose Brushable	06AAA1	As required	
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

Kit tool 571140T02R00				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103002000	Test	1	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Drawing	R57102001
Non Destructive Test Manual (NTM)	51-10-01 PART 6
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00
Fig. A-FCBAA RH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FCDAA Definition of the Washer Thickness for the New Taperlok	Sheet 01
Fig. A-FDAAA Removal/Installation of the Cooling Duct	Sheet 01
Fig. A-FDBAA Removal/Installation of the Belly Fairing Structure	Sheet 01 Sheet 02 Sheet 03

In accordance with SRM 51-42-11, SRM 51-43-00 and SRM 51-24-00.

Refer to AMM Task 20-21-11-911-001.

Refer to NTM 51-10-01 PART 6

Refer to [Fig. A-FCBAA Sheet 01](#) to [Fig. A-FCBAA Sheet 07](#)

(a) Depending on the result of the inspection, remove for access :

1 Remove the cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (54)	Retain
1	Duct	Item (52)	Retain
	attached with		
1	Clamp	Item (63)	Retain
8	Screw	Item (64)	Retain
8	Washer	Item (65)	Discard

2 Remove the structure of the belly fairing.

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed attached with	Item (171)	Retain
3	Rivet	Item (62)	Discard
7	Rivet	Item (60)	Discard
10	Bush and/or	Item (61)	Discard
1	Web 4 assy attached with	Item (170)	Retain
7	Rivet	Item (60)	Discard
7	Bush	Item (61)	Discard

(b) In accordance with the result of the inspection, remove the taperlok(s) :

Refer to [Fig. A-FCBAA Sheet 01](#) to [Fig. A-FCBAA Sheet 04](#)

6	Screw	Item (16)	Discard
10	Screw	Item (5)	Discard
9	Screw	Item (7)	Discard
7	Screw	Item (3)	Discard
2	Screw	Item (14)	Discard
1	Screw	Item (1)	Discard
2	Screw	Item (12)	Discard
2	Screw	Item (70)	Discard
8	Screw	Item (71)	Discard
6	Screw	Item (72)	Discard
2	Screw	Item (77)	Discard

2	Screw	Item (79)	Discard
1	Screw	Item (81)	Discard

NOTE: The nuts are removed during the inspection.

- (c) Do a rototest inspection of the holes for cracks refer to NTM 51-10-01 PART 6.

1 If cracks are found, contact AIRBUS before any further work.

2 If no crack is found, do the following steps.

- (d) Oversize the hole to the next nominal diameter in accordance with the drilling procedure given in the Repair Instruction Drawing R57102001 and use the drilling tool SOA-TL-R57102001B.

NOTE: To prevent any accidental damage, we recommend to train to oversize the taperlok holes on the test specimen.

98D57103002000 Test specimen 1

- (e) Depending on the removal, install :

Refer to [Fig. A-FCBAA Sheet 01](#), [Fig. A-FCBAA Sheet 02](#), [Fig. A-FCBAA Sheet 05](#), [Fig. A-FCBAA Sheet 06](#), [Fig. A-FCBAA Sheet 07](#)

6	Screw	ABS1418K8-17	Item 38
8	Screw	ABS1418K8-18	Item 33
9	Screw	ABS1418K8-19	Item 34
1	Screw	ABS1418K8-13	Item 31
7	Screw	ABS1418K8-14	Item 32
2	Screw	ABS1418K8-20	Item 78
1	Screw	ABS1418K8-22	Item 82
	with		
34	Nut	ASNA2532-8	Item 35
	and if necessary		
34	Washer	NSA5372-816AX	
	and/or		
34	Washer	NSA5372-816EX	
	and/or		

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34	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K8-25	Item 43
2	Screw	ABS1418K8-26	Item 37
2	Screw	ABS1418K8-21	Item 36
	with		
6	Nut	ASNA2531-8	Item 44
	and if necessary		
6	Washer	NSA5372-816AX	
	and/or		
6	Washer	NSA5372-816EX	
	and/or		
6	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K9-15	Item 73
8	Screw	ABS1418K9-16	Item 74
6	Screw	ABS1418K9-17	Item 75
	with		
16	Nut	ASNA2532-9	Item 76
	and if necessary		
16	Washer	NSA5372-916AX	
	and/or		
16	Washer	NSA5372-916EX	
	and/or		
16	Washer	NSA5372-916BX	
	and		
2	Screw	ABS1418K7-23	Item 80
	with		

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2 Nut ASNA2531-7 Item 11

and if necessary

2 Washer NSA5372-716AX

and/or

2 Washer NSA5372-716EX

and/or

2 Washer NSA5372-716BX

NOTE: Install the screw in accordance with the installation principle given in the repair instruction R57102001.

NOTE: If necessary, install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCDA Sheet 01](#).

NOTE: Wet assembly, apply :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

NOTE: Torque the nuts between 5.60daN and 6.70daN (493lbf.in. and 592lbf.in.).

NOTE: Outside the wing center box, apply on the screws :

Primer 04EAC2 As required
Polyurethane Paint
- Corrosion
Inhibiting

Wash Primer - 04CMA2 As required
Structure

Top Coat 04JAA3 As required
Polyurethane -
External Structure

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(f) If removed, install :

1 The structure of the belly fairing

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (171)	Retained at removal
---	----------------	------------	---------------------

with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

3	Bolt	EN6115T2-2	Item 62
---	------	------------	---------

10	Nut	ASNA2536-2	Item 61
----	-----	------------	---------

and/or

1	Web 4 assy	Item (170)	Retained at removal
---	------------	------------	---------------------

attached with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

7	Nut	ASNA2536-2	Item 61
---	-----	------------	---------

NOTE: Apply to the interface of the structure and all the parts :

Polysulfide Sealant- General Purpose Brushable	06AAA1	As required
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2 The cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (54)	Retained at removal
---	------	-----------	---------------------

1	Duct	Item (52)	Retained at removal
---	------	-----------	---------------------

with

1	Clamp	Item (63)	Retained at removal
---	-------	-----------	---------------------

8	Screw	Item (64)	Retained at removal
---	-------	-----------	---------------------

8	Washer	NAS1149D0332K	Item 65
---	--------	---------------	---------

****CONF 002**

(1) Subtask 571140-833-001-002 - Replace the Nuts on the LH Side

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	12.00
Minimum number of person	2
Subtask elapsed time	6.00
Skills	AIRFRAME

Material necessary to do the job

Kit 571140A02R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
10	ASNA2532-7	34	NUT	
11	ASNA2531-7	6	NUT	
13	ASNA2532-8	16	NUT	
15	ASNA2531-6	2	NUT	
	NSA5372-616AX	2	WASHER	
	NSA5372-616BX	2	WASHER	
	NSA5372-616CX	2	WASHER	
	NSA5372-616DX	2	WASHER	
	NSA5372-616EX	2	WASHER	
	NSA5372-716AX	40	WASHER	
	NSA5372-716BX	40	WASHER	
	NSA5372-716CX	40	WASHER	
	NSA5372-716DX	40	WASHER	
	NSA5372-716EX	40	WASHER	
	NSA5372-816AX	16	WASHER	
	NSA5372-816BX	16	WASHER	
	NSA5372-816CX	16	WASHER	
	NSA5372-816DX	16	WASHER	
	NSA5372-816EX	16	WASHER	

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	

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ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11 51-43-00
Fig. A-FCAAA LH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FCCAA Definition of the Washer Thickness for the Original Taperlok	Sheet 01 Sheet 02 Sheet 03

NOTE: The nuts are removed during the inspection.

(a) Install the nuts.

In accordance with SRM 51-42-11 and SRM 51-43-00.

Refer to AMM Task 20-21-11-911-001.

Refer to [Fig. A-FCAAA Sheet 01](#), [Fig. A-FCAAA Sheet 02](#), [Fig. A-FCAAA Sheet 05](#), [Fig. A-FCAAA Sheet 06](#), [Fig. A-FCAAA Sheet 07](#)

34	Nut	ASNA2532-7	Item 10
6	Nut	ASNA2531-7	Item 11
	with		
40	Washer	NSA5372-716AX	
	and/or		
40	Washer	NSA5372-716BX	
	and/or		
40	Washer	NSA5372-716CX	
	and/or		
40	Washer	NSA5372-716DX	
	and/or		

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40	Washer	NSA5372-716EX	
	and		
2	Nut	ASNA2531-6	Item 15
	with		
2	Washer	NSA5372-616AX	
	and/or		
2	Washer	NSA5372-616BX	
	and/or		
2	Washer	NSA5372-616CX	
	and/or		
2	Washer	NSA5372-616DX	
	and/or		
2	Washer	NSA5372-616EX	
	and		
16	Nut	ASNA2532-8	Item 13
	with		
16	Washer	NSA5372-816AX	
	and/or		
16	Washer	NSA5372-816BX	
	and/or		
16	Washer	NSA5372-816CX	
	and/or		
16	Washer	NSA5372-816DX	
	and/or		
16	Washer	NSA5372-816EX	

NOTE: Install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCCAA Sheet 01](#).

NOTE: If necessary, the installation of the three washers is approved in this condition.

NOTE: Torque the nuts between 3.80daN and 4.60daN (342lbf.in. and

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411lbf.in.).

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(2) Subtask 571140-833-002-002 - Replace the Nuts on the RH Side

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	12.00
Minimum number of person	2
Subtask elapsed time	6.00
Skills	AIRFRAME

Material necessary to do the job

Kit 571140A02R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
10	ASNA2532-7	34	NUT	
11	ASNA2531-7	6	NUT	
13	ASNA2532-8	16	NUT	
15	ASNA2531-6	2	NUT	
	NSA5372-616AX	2	WASHER	
	NSA5372-616BX	2	WASHER	
	NSA5372-616CX	2	WASHER	
	NSA5372-616DX	2	WASHER	
	NSA5372-616EX	2	WASHER	
	NSA5372-716AX	40	WASHER	
	NSA5372-716BX	40	WASHER	
	NSA5372-716CX	40	WASHER	
	NSA5372-716DX	40	WASHER	
	NSA5372-716EX	40	WASHER	
	NSA5372-816AX	16	WASHER	
	NSA5372-816BX	16	WASHER	
	NSA5372-816CX	16	WASHER	
	NSA5372-816DX	16	WASHER	

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ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
	NSA5372-816EX	16	WASHER	

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11 51-43-00
Fig. A-FCCAA Definition of the Washer Thickness for the Original Taperlok	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCBAA RH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

NOTE: The nuts are removed during the inspection.

(a) Install the nuts.

In accordance with SRM 51-42-11 and SRM 51-43-00.

Refer to AMM Task 20-21-11-911-001.

Refer to [Fig. A-FCBAA Sheet 01](#), [Fig. A-FCBAA Sheet 02](#), [Fig. A-FCBAA Sheet 05](#), [Fig. A-FCBAA Sheet 06](#), [Fig. A-FCBAA Sheet 07](#)

34	Nut	ASNA2532-7	Item 10
6	Nut	ASNA2531-7	Item 11
	with		
40	Washer	NSA5372-716AX	
	and/or		
40	Washer	NSA5372-716BX	
	and/or		

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I	40	Washer	NSA5372-716CX	
I		and/or		
I	40	Washer	NSA5372-716DX	
I		and/or		
I	40	Washer	NSA5372-716EX	
I		and		
	2	Nut	ASNA2531-6	Item 15
		with		
I	2	Washer	NSA5372-616AX	
		and/or		
I	2	Washer	NSA5372-616BX	
		and/or		
I	2	Washer	NSA5372-616CX	
		and/or		
I	2	Washer	NSA5372-616DX	
		and/or		
I	2	Washer	NSA5372-616EX	
I		and		
I	16	Nut	ASNA2532-8	Item 13
I		with		
I	16	Washer	NSA5372-816AX	
I		and/or		
I	16	Washer	NSA5372-816BX	
I		and/or		
I	16	Washer	NSA5372-816CX	
I		and/or		
I	16	Washer	NSA5372-816DX	
I		and/or		
I	16	Washer	NSA5372-816EX	

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NOTE: Install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCCAA Sheet 01](#).

NOTE: If necessary, the installation of the three washers is approved in this condition.

NOTE: Torque the nuts between 3.80daN and 4.60daN (342lbf.in. and 411lbf.in.).

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(3) Subtask 571140-833-003-002 - Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	16.00
Minimum number of person	2
Subtask elapsed time	10.00
Skills	AIRFRAME NON DESTRUCTIVE TESTING

Material necessary to do the job

Kit 571140A02R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
60	EN6115T2-3	14	BOLT	
61	ASNA2536-2	17	NUT	
62	EN6115T2-2	3	BOLT	
65	NAS1149D0332K	8	WASHER	

Component COMPA01				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
11	ASNA2531-7	2	NUT	

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ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
31	ABS1418K8-13	1	SCREW	
32	ABS1418K8-14	7	SCREW	
33	ABS1418K8-18	8	SCREW	
34	ABS1418K8-19	9	SCREW	
35	ASNA2532-8	34	NUT	
36	ABS1418K8-21	2	SCREW	
37	ABS1418K8-26	2	SCREW	
38	ABS1418K8-17	6	SCREW	
43	ABS1418K8-25	2	SCREW	
44	ASNA2531-8	6	NUT	
73	ABS1418K9-15	2	SCREW	
74	ABS1418K9-16	8	SCREW	
75	ABS1418K9-17	6	SCREW	
76	ASNA2532-9	16	NUT	
78	ABS1418K8-20	2	SCREW	
80	ABS1418K7-23	2	SCREW	
82	ABS1418K8-22	1	SCREW	
	NSA5372-716AX	2	WASHER	
	NSA5372-716BX	2	WASHER	
	NSA5372-716EX	2	WASHER	
	NSA5372-816AX	40	WASHER	
	NSA5372-816BX	40	WASHER	
	NSA5372-816EX	40	WASHER	
	NSA5372-916AX	16	WASHER	
	NSA5372-916BX	16	WASHER	
	NSA5372-916EX	16	WASHER	

NOTE: The quantities given in the component table are for all the taperloks replacement.

NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Wash Primer - Structure	04CMA2	As required	
	Primer Polyurethane Paint - Corrosion Inhibiting	04EAC2	As required	
	Top Coat Polyurethane - External Structure	04JAA3	As required	
	Polysulfide Sealant-General Purpose Brushable	06AAA1	As required	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Drawing	R57102001
Non Destructive Test Manual (NTM)	51-10-01 PART 6
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00
Fig. A-FCAAA LH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FCDA Definition of the Washer Thickness for the New Taperlok	Sheet 01
Fig. A-FDAAA Removal/Installation of the Cooling Duct	Sheet 01
Fig. A-FDBAA Removal/Installation of the Belly Fairing Structure	Sheet 01 Sheet 02 Sheet 03

1	Duct	Item (52)	Retain
	attached with		
1	Clamp	Item (63)	Retain
8	Screw	Item (64)	Retain
8	Washer	Item (65)	Discard

2 Remove the structure of the belly fairing.

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (51)	Retain
	attached with		
3	Rivet	Item (62)	Discard
7	Rivet	Item (60)	Discard
10	Bush	Item (61)	Discard
	and/or		
1	Web 4 assy	Item (50)	Retain
	attached with		
7	Rivet	Item (60)	Discard
7	Bush	Item (61)	Discard

(b) In accordance with the result of the inspection, remove the taperlok(s) :

Refer to [Fig. A-FCAAA Sheet 01](#) to [Fig. A-FCAAA Sheet 04](#)

6	Screw	Item (16)	Discard
10	Screw	Item (5)	Discard
9	Screw	Item (7)	Discard
7	Screw	Item (3)	Discard
2	Screw	Item (14)	Discard
1	Screw	Item (1)	Discard
2	Screw	Item (12)	Discard

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2	Screw	Item (70)	Discard
8	Screw	Item (71)	Discard
6	Screw	Item (72)	Discard
2	Screw	Item (77)	Discard
2	Screw	Item (79)	Discard
1	Screw	Item (81)	Discard

NOTE: The nuts are removed during the inspection.

- (c) Do a rototest inspection of the holes for cracks, refer to NTM 51-10-01 PART 6.

1 If cracks are found, contact AIRBUS before any further work.

2 If no crack is found, do the following steps.

- (d) Oversize the hole to the next nominal diameter in accordance with the drilling procedure given in the Repair Instruction Drawing R57102001 and use the drilling tool SOA-TL-R57102001B.

NOTE: To prevent any accidental damage, we recommend to train to oversize the taperlok holes on the test specimen.

98D57103002000 Test specimen 1

- (e) Depending on the removal, install :

Refer to [Fig. A-FCAAA Sheet 01](#), [Fig. A-FCAAA Sheet 02](#), [Fig. A-FCAAA Sheet 05](#), [Fig. A-FCAAA Sheet 06](#), [Fig. A-FCAAA Sheet 07](#)

6	Screw	ABS1418K8-17	Item 38
8	Screw	ABS1418K8-18	Item 33
9	Screw	ABS1418K8-19	Item 34
1	Screw	ABS1418K8-13	Item 31
7	Screw	ABS1418K8-14	Item 32
2	Screw	ABS1418K8-20	Item 78
1	Screw	ABS1418K8-22	Item 82
with			
34	Nut	ASNA2532-8	Item 35

	and if necessary		
34	Washer	NSA5372-816AX	
	and/or		
34	Washer	NSA5372-816EX	
	and/or		
34	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K8-25	Item 43
2	Screw	ABS1418K8-26	Item 37
2	Screw	ABS1418K8-21	Item 36
	with		
6	Nut	ASNA2531-8	Item 44
	and if necessary		
6	Washer	NSA5372-816AX	
	and/or		
6	Washer	NSA5372-816EX	
	and/or		
6	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K9-15	Item 73
8	Screw	ABS1418K9-16	Item 74
6	Screw	ABS1418K9-17	Item 75
	with		
16	Nut	ASNA2532-9	Item 76
	and if necessary		
16	Washer	NSA5372-916AX	
	and/or		
16	Washer	NSA5372-916EX	

	and/or		
16	Washer	NSA5372-916BX	
	and		
2	Screw	ABS1418K7-23	Item 80
	with		
2	Nut	ASNA2531-7	Item 11
	and if necessary		
2	Washer	NSA5372-716AX	
	and/or		
2	Washer	NSA5372-716EX	
	and/or		
2	Washer	NSA5372-716BX	

NOTE: Install the screw in accordance with the installation principle given in the repair instruction R57102001.

NOTE: If necessary, install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCDA Sheet 01](#).

NOTE: Wet assembly, apply :

Polysulfide Sealant-	06ABB1	As required
Fuel Tank Fillet		

NOTE: Torque the nuts between 5.60daN and 6.70daN (493lbf.in. and 592lbf.in.).

NOTE: Outside the wing center box, apply on the screws :

Primer	04EAC2	As required
Polyurethane Paint		
- Corrosion		
Inhibiting		

Wash Primer -	04CMA2	As required
Structure		

Top Coat	04JAA3	As required
Polyurethane -		
External Structure		

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(f) If removed, install :

1 The structure of the belly fairing

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (51)	Retained at removal
---	----------------	-----------	------------------------

with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

3	Bolt	EN6115T2-2	Item 62
---	------	------------	---------

10	Nut	ASNA2536-2	Item 61
----	-----	------------	---------

and/or

1	Web 4 assy	Item (50)	Retained at removal
---	------------	-----------	------------------------

attached with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

7	Nut	ASNA2536-2	Item 61
---	-----	------------	---------

NOTE: Apply to the interface of the structure and all the parts :

Polysulfide Sealant- 06AAA1 As required
General Purpose
Brushable

2 The cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (53)	Retained at removal
---	------	-----------	------------------------

1	Duct	Item (52)	Retained at removal
---	------	-----------	------------------------

with

1	Clamp	Item (63)	Retained at removal
8	Screw	Item (64)	Retained at removal
8	Washer	NAS1149D0332K	Item 65

(4) Subtask 571140-833-004-002 - Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	16.00
Minimum number of person	2
Subtask elapsed time	10.00
Skills	AIRFRAME NON DESTRUCTIVE TESTING

Material necessary to do the job

Kit 571140A02R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
60	EN6115T2-3	14	BOLT	
61	ASNA2536-2	17	NUT	
62	EN6115T2-2	3	BOLT	
65	NAS1149D0332K	8	WASHER	

Component COMPA01				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
11	ASNA2531-7	2	NUT	
31	ABS1418K8-13	1	SCREW	
32	ABS1418K8-14	7	SCREW	
33	ABS1418K8-18	8	SCREW	
34	ABS1418K8-19	9	SCREW	
35	ASNA2532-8	34	NUT	
36	ABS1418K8-21	2	SCREW	
37	ABS1418K8-26	2	SCREW	
38	ABS1418K8-17	6	SCREW	
43	ABS1418K8-25	2	SCREW	

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ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
44	ASNA2531-8	6	NUIT	
73	ABS1418K9-15	2	SCREW	
74	ABS1418K9-16	8	SCREW	
75	ABS1418K9-17	6	SCREW	
76	ASNA2532-9	16	NUT	
78	ABS1418K8-20	2	SCREW	
80	ABS1418K7-23	2	SCREW	
82	ABS1418K8-22	1	SCREW	
	NSA5372-716AX	2	WASHER	
	NSA5372-716BX	2	WASHER	
	NSA5372-716EX	2	WASHER	
	NSA5372-816AX	40	WASHER	
	NSA5372-816BX	40	WASHER	
	NSA5372-816EX	40	WASHER	
	NSA5372-916AX	16	WASHER	
	NSA5372-916BX	16	WASHER	
	NSA5372-916EX	16	WASHER	

NOTE: The quantities given in the component table are for all the taperloks replacement.

NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Wash Primer - Structure	04CMA2	As required	
	Primer Polyurethane Paint - Corrosion Inhibiting	04EAC2	As required	
	Top Coat Polyurethane - External Structure	04JAA3	As required	
	Polysulfide Sealant-General Purpose Brushable	06AAA1	As required	
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

Kit tool 571140T02R00				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103002000	Test	1	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Drawing	R57102001
Non Destructive Test Manual (NTM)	51-10-01 PART 6
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00
Fig. A-FCBAA RH Side, Replacement of the Lower Panel Fasteners	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FCDAA Definition of the Washer Thickness for the New Taperlok	Sheet 01
Fig. A-FDAAA Removal/Installation of the Cooling Duct	Sheet 01
Fig. A-FDBAA Removal/Installation of the Belly Fairing Structure	Sheet 01 Sheet 02 Sheet 03

In accordance with SRM 51-42-11, SRM 51-43-00 and SRM 51-24-00.

Refer to AMM Task 20-21-11-911-001.

Refer to NTM 51-10-01 PART 6

Refer to [Fig. A-FCBAA Sheet 01](#) to [Fig. A-FCBAA Sheet 07](#)

(a) Depending on the result of the inspection, remove for access :

1 Remove the cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (54)	Retain
1	Duct	Item (52)	Retain
	attached with		
1	Clamp	Item (63)	Retain
8	Screw	Item (64)	Retain
8	Washer	Item (65)	Discard

2 Remove the structure of the belly fairing.

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed attached with	Item (171)	Retain
3	Rivet	Item (62)	Discard
7	Rivet	Item (60)	Discard
10	Bush and/or	Item (61)	Discard
1	Web 4 assy attached with	Item (170)	Retain
7	Rivet	Item (60)	Discard
7	Bush	Item (61)	Discard

(b) In accordance with the result of the inspection, remove the taperlok(s) :

Refer to [Fig. A-FCBAA Sheet 01](#) to [Fig. A-FCBAA Sheet 04](#)

6	Screw	Item (16)	Discard
10	Screw	Item (5)	Discard
9	Screw	Item (7)	Discard
7	Screw	Item (3)	Discard
2	Screw	Item (14)	Discard
1	Screw	Item (1)	Discard
2	Screw	Item (12)	Discard
2	Screw	Item (70)	Discard
8	Screw	Item (71)	Discard
6	Screw	Item (72)	Discard
2	Screw	Item (77)	Discard

2	Screw	Item (79)	Discard
1	Screw	Item (81)	Discard

NOTE: The nuts are removed during the inspection.

- (c) Do a rototest inspection of the holes for cracks refer to NTM 51-10-01 PART 6.

1 If cracks are found, contact AIRBUS before any further work.

2 If no crack is found, do the following steps.

- (d) Oversize the hole to the next nominal diameter in accordance with the drilling procedure given in the Repair Instruction Drawing R57102001 and use the drilling tool SOA-TL-R57102001B.

NOTE: To prevent any accidental damage, we recommend to train to oversize the taperlok holes on the test specimen.

98D57103002000 Test specimen 1

- (e) Depending on the removal, install :

Refer to [Fig. A-FCBAA Sheet 01](#), [Fig. A-FCBAA Sheet 02](#), [Fig. A-FCBAA Sheet 05](#), [Fig. A-FCBAA Sheet 06](#), [Fig. A-FCBAA Sheet 07](#)

6	Screw	ABS1418K8-17	Item 38
8	Screw	ABS1418K8-18	Item 33
9	Screw	ABS1418K8-19	Item 34
1	Screw	ABS1418K8-13	Item 31
7	Screw	ABS1418K8-14	Item 32
2	Screw	ABS1418K8-20	Item 78
1	Screw	ABS1418K8-22	Item 82
	with		
34	Nut	ASNA2532-8	Item 35
	and if necessary		
34	Washer	NSA5372-816AX	
	and/or		
34	Washer	NSA5372-816EX	
	and/or		

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34	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K8-25	Item 43
2	Screw	ABS1418K8-26	Item 37
2	Screw	ABS1418K8-21	Item 36
	with		
6	Nut	ASNA2531-8	Item 44
	and if necessary		
6	Washer	NSA5372-816AX	
	and/or		
6	Washer	NSA5372-816EX	
	and/or		
6	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K9-15	Item 73
8	Screw	ABS1418K9-16	Item 74
6	Screw	ABS1418K9-17	Item 75
	with		
16	Nut	ASNA2532-9	Item 76
	and if necessary		
16	Washer	NSA5372-916AX	
	and/or		
16	Washer	NSA5372-916EX	
	and/or		
16	Washer	NSA5372-916BX	
	and		
2	Screw	ABS1418K7-23	Item 80
	with		

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2 Nut ASNA2531-7 Item 11

and if necessary

2 Washer NSA5372-716AX

and/or

2 Washer NSA5372-716EX

and/or

2 Washer NSA5372-716BX

NOTE: Install the screw in accordance with the installation principle given in the repair instruction R57102001.

NOTE: If necessary, install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCDA Sheet 01](#).

NOTE: Wet assembly, apply :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

NOTE: Torque the nuts between 5.60daN and 6.70daN (493lbf.in. and 592lbf.in.).

NOTE: Outside the wing center box, apply on the screws :

Primer 04EAC2 As required
Polyurethane Paint
- Corrosion
Inhibiting

Wash Primer - 04CMA2 As required
Structure

Top Coat 04JAA3 As required
Polyurethane -
External Structure

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(f) If removed, install :

1 The structure of the belly fairing

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed		Item (171)	Retained at removal
---	----------------	--	------------	---------------------

with

7	Bolt	EN6115T2-3	Item 60	
---	------	------------	---------	--

3	Bolt	EN6115T2-2	Item 62	
---	------	------------	---------	--

10	Nut	ASNA2536-2	Item 61	
----	-----	------------	---------	--

and/or

1	Web 4 assy		Item (170)	Retained at removal
---	------------	--	------------	---------------------

attached with

7	Bolt	EN6115T2-3	Item 60	
---	------	------------	---------	--

7	Nut	ASNA2536-2	Item 61	
---	-----	------------	---------	--

NOTE: Apply to the interface of the structure and all the parts :

Polysulfide Sealant- General Purpose Brushable	06AAA1	As required
--	--------	-------------

2 The cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct		Item (54)	Retained at removal
---	------	--	-----------	---------------------

1	Duct		Item (52)	Retained at removal
---	------	--	-----------	---------------------

with

1	Clamp		Item (63)	Retained at removal
---	-------	--	-----------	---------------------

8	Screw		Item (64)	Retained at removal
---	-------	--	-----------	---------------------

8	Washer	NAS1149D0332K	Item 65	
---	--------	---------------	---------	--

****CONF ALL****D. TEST******CONF 001****(1) Subtask 571140-710-001-001 - Test**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 20-28-00-720-005
	Task 20-28-00-869-002
	Task 28-11-00-280-002
	Task 28-21-00-710-006
	Task 57-17-11-400-001
	Task 57-27-11-400-001

NOTE: This test procedure is to be accomplished after the installation of the parts removed for the access SUBTASK 571137-410-001 001 Install/Close Items Removed/Opened for Access on the LH Side, SUBTASK 571137-410-002 001 Install/Close Items Removed/Opened for Access on the RH Side, SUBTASK 571137-410-005 001 Install the Fuel Strainer, LH Side and SUBTASK 571137-410-006 001 Install the Fuel Strainer, RH Side.

(a) Job Set-up :

- 1 Remove the safety clips and tags and close the circuit breakers as specified after the installation of the access covers 147AZ and 148AZ, refer to AMM Task 57-17-11-400-001.
- 2 Remove the safety clips and tags and close the circuit breakers as specified after the installation of the access panels 540AB and 640AB, refer to AMM Task 57-27-11-400-001.

(b) Test :

- 1 Do the test procedure as specified after the installation of the access covers 147AZ and 148AZ, refer to AMM Task 57-17-11-400-001.
- 2 Do the test procedure as specified after the installation of the access panels 540AB and 640AB, refer to AMM Task 57-27-11-400-001.
- 3 Do a fuel tanks leak checks (only leak test of the center tank or/and leak test of the wing tanks - without fuel - by the local blowing procedure), refer to AMM Task 28-11-00-280-002.

NOTE: Do this test on the taperlok bolts inspected by this Service Bulletin.

- 4 For all the access panels and covers removed during the preparation procedure, do a check of the electrical bonding (external metal surfaces/parts (hinges or fixed)) refer to AMM Task 20-28-00-720-005.

NOTE: For the maximum permitted resistance values : refer to the table AMM Task 20-28-00-869-002.

****CONF 002**

(1) Subtask 571140-710-001-001 - Test

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 20-28-00-720-005
	Task 20-28-00-869-002
	Task 28-11-00-280-002
	Task 28-21-00-710-006
	Task 57-17-11-400-001
	Task 57-27-11-400-001

NOTE: This test procedure is to be accomplished after the installation of the parts removed for the access SUBTASK 571137-410-001 001 Install/Close Items Removed/Opened for Access on the LH Side, SUBTASK 571137-410-002 001 Install/Close Items Removed/Opened for Access on the RH Side, SUBTASK 571137-410-005 001 Install the Fuel Strainer, LH Side and SUBTASK 571137-410-006 001 Install the Fuel Strainer, RH Side.

(a) Job Set-up :

- 1 Remove the safety clips and tags and close the circuit breakers as specified after the installation of the access covers 147AZ and 148AZ, refer to AMM Task 57-17-11-400-001.
- 2 Remove the safety clips and tags and close the circuit breakers as specified after the installation of the access panels 540AB and 640AB, refer to AMM Task 57-27-11-400-001.

(b) Test :

- 1 Do the test procedure as specified after the installation of the access covers 147AZ and 148AZ, refer to AMM Task 57-17-11-400-001.
- 2 Do the test procedure as specified after the installation of the access panels 540AB and 640AB, refer to AMM Task 57-27-11-400-001.
- 3 Do a fuel tanks leak checks (only leak test of the center tank or/and leak test of the wing tanks - without fuel - by the local blowing procedure), refer to AMM Task 28-11-00-280-002.

NOTE: Do this test on the taperlok bolts inspected by this Service

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- 4 For all the access panels and covers removed during the preparation procedure, do a check of the electrical bonding (external metal surfaces/parts (hinges or fixed)) refer to AMM Task 20-28-00-720-005.

NOTE: For the maximum permitted resistance values : refer to the table AMM Task 20-28-00-869-002.

****CONF ALL**

E. CLOSE-UP

****CONF 001**

- (1) **Subtask 571140-410-001-001 - Install/Close Items Removed/Opened for Access on the LH Side**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-400-001 Task 28-21-54-400-001 Task 53-35-11-400-001 Task 57-17-11-400-001 Task 57-27-11-400-001 Task 57-27-12-400-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-001 001 Testing.

- Make sure that the work areas are clean and clear of tools and other items of equipment.
- Install the covers Item (80) and (81) of the fuel pump strainer FIN 6QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-400-001.
- Install the fuel pump suction valve, refer to AMM Task 28-21-42-400-001.
- Close the fuel surge access door 540BZ, refer to AMM Task 57-27-12-400-001.
- Install the wing access panel 540AB, refer to AMM Task 57-27-11-400-001.
- Install the access cover 147AZ, refer to AMM Task 57-17-11-400-001.
- If removed, install the panels 191LB and 147CB on the wing to fuselage, refer to AMM Task 53-35-11-400-001.

(2) Subtask 571140-410-002-001 - Install/Close Items Removed/Opened for Access on the RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-400-001 Task 28-21-54-400-001 Task 53-35-11-400-001 Task 57-17-11-400-001 Task 57-27-11-400-001 Task 57-27-12-400-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-001 001 Testing.

- (a) Make sure that the work areas are clean and clear of tools and other items of equipment.
- (b) Install the covers Item (80) and (81) of the fuel pump strainer FIN 8QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-400-001.
- (c) Install the fuel pump suction valve, refer to AMM Task 28-21-42-400-001.
- (d) Close the fuel surge access door 640BZ, refer to AMM Task 57-27-12-400-001.
- (e) Install the wing access panel 640AB, refer to AMM Task 57-27-11-400-001.
- (f) Install the access cover 148AZ, refer to AMM Task 57-17-11-400-001.
- (g) If removed, install the panels , 192LB and 148DB on the wing to fuselage, refer to AMM Task 53-35-11-400-001.

(3) Subtask 571140-410-005-001 - Install the Fuel Strainer, LH Side

Manpower Resources	
Skills	NON SPECIFIC

Material necessary to do the job	
----------------------------------	--

Kit 571140A01R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
69	NAS1149F0332P	2	WASHER	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11

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References	
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-001 001 Testing.

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

(a) If removed, install the fuel retainer for LH side:

Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)

In accordance with SRM 51-42-11.

Refer to AMM Task 20-21-11-911-001.

1	Strainer Assembly	Item (55)	Retained at removal
1	Strainer Assembly	Item (57)	Retained at removal
	with		
8	Bolt	Item (66)	Retained at removal
2	Bolt	Item (67)	Retained at removal
2	Bolt	Item (68)	Retained at removal
2	Washer	NAS1149F0332P	Item 69

(4) Subtask 571140-410-006-001 - Install the Fuel Strainer, RH Side

Manpower Resources	
Skills	NON SPECIFIC

Material necessary to do the job

Kit 571140A01R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
69	NAS1149F0332P	2	WASHER	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11

References	
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-001 001 Testing.

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

(a) If removed, install the fuel retainer for RH side:

Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)

In accordance with SRM 51-42-11.

Refer to AMM Task 20-21-11-911-001.

1	Strainer Assembly	Item (56)	Retained at removal
1	Strainer Assembly	Item (58)	Retained at removal
	with		
8	Bolt	Item (66)	Retained at removal
2	Bolt	Item (67)	Retained at removal
2	Bolt	Item (68)	Retained at removal
2	Washer	NAS1149F0332P	Item 69

(5) Subtask 571140-942-001-001 - Close-up

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-25-00-650-001 Task 32-12-00-410-001

- (a) Refuel the LH and RH wing tanks, refer to AMM Task 28-25-00-650-001.
- (b) Refuel the center tank, refer to AMM Task 28-25-00-650-001.
- (c) Close the main landing gear doors, refer to AMM Task 32-12-00-410-001.
- (d) Remove the access platform(s).

- (e) Put the aircraft back to its initial configuration.

****CONF 002****(1) Subtask 571140-410-001-001 - Install/Close Items Removed/Opened for Access on the LH Side**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-400-001 Task 28-21-54-400-001 Task 53-35-11-400-001 Task 57-17-11-400-001 Task 57-27-11-400-001 Task 57-27-12-400-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-001 001 Testing.

- (a) Make sure that the work areas are clean and clear of tools and other items of equipment.
- (b) Install the covers Item (80) and (81) of the fuel pump strainer FIN 6QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-400-001.
- (c) Install the fuel pump suction valve, refer to AMM Task 28-21-42-400-001.
- (d) Close the fuel surge access door 540BZ, refer to AMM Task 57-27-12-400-001.
- (e) Install the wing access panel 540AB, refer to AMM Task 57-27-11-400-001.
- (f) Install the access cover 147AZ, refer to AMM Task 57-17-11-400-001.
- (g) If removed, install the panels 191LB and 147CB on the wing to fuselage, refer to AMM Task 53-35-11-400-001.

(2) Subtask 571140-410-002-001 - Install/Close Items Removed/Opened for Access on the RH Side

Manpower Resources	
Skills	NON SPECIFIC

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References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-400-001 Task 28-21-54-400-001 Task 53-35-11-400-001 Task 57-17-11-400-001 Task 57-27-11-400-001 Task 57-27-12-400-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-001 001 Testing.

- (a) Make sure that the work areas are clean and clear of tools and other items of equipment.
- (b) Install the covers Item (80) and (81) of the fuel pump strainer FIN 8QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-400-001.
- (c) Install the fuel pump suction valve, refer to AMM Task 28-21-42-400-001.
- (d) Close the fuel surge access door 640BZ, refer to AMM Task 57-27-12-400-001.
- (e) Install the wing access panel 640AB, refer to AMM Task 57-27-11-400-001.
- (f) Install the access cover 148AZ, refer to AMM Task 57-17-11-400-001.
- (g) If removed, install the panels , 192LB and 148DB on the wing to fuselage, refer to AMM Task 53-35-11-400-001.

(3) Subtask 571140-410-005-002 - Install the Fuel Strainer, LH Side

Manpower Resources	
Skills	NON SPECIFIC

Material necessary to do the job

Kit 571140A02R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
69	NAS1149F0332P	2	WASHER	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-001 001 Testing.

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NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

(a) If removed, install the fuel retainer for LH side:

Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)

In accordance with SRM 51-42-11.

Refer to AMM Task 20-21-11-911-001.

1	Strainer Assembly	Item (55)	Retained at removal
1	Strainer Assembly	Item (57)	Retained at removal
	with		
8	Bolt	Item (66)	Retained at removal
2	Bolt	Item (67)	Retained at removal
2	Bolt	Item (68)	Retained at removal
2	Washer	NAS1149F0332P Item 69	

(4) Subtask 571140-410-006-002 - Install the Fuel Strainer, RH Side

Manpower Resources	
Skills	NON SPECIFIC

Material necessary to do the job

Kit 571140A02R04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
69	NAS1149F0332P	2	WASHER	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-001 001 Testing.

NOTE: This step is only valid for aircraft on which Service Bulletin No.

A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- (a) If removed, install the fuel retainer for RH side:

Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)

In accordance with SRM 51-42-11.

Refer to AMM Task 20-21-11-911-001.

1	Strainer Assembly	Item (56)	Retained at removal
1	Strainer Assembly	Item (58)	Retained at removal
	with		
8	Bolt	Item (66)	Retained at removal
2	Bolt	Item (67)	Retained at removal
2	Bolt	Item (68)	Retained at removal
2	Washer	NAS1149F0332P	Item 69

(5) Subtask 571140-942-001-001 - Close-up

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-25-00-650-001 Task 32-12-00-410-001

- (a) Refuel the LH and RH wing tanks, refer to AMM Task 28-25-00-650-001.
- (b) Refuel the center tank, refer to AMM Task 28-25-00-650-001.
- (c) Close the main landing gear doors, refer to AMM Task 32-12-00-410-001.
- (d) Remove the access platform(s).
- (e) Put the aircraft back to its initial configuration.

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Task 571140-832-802-001 - INSPECTION ADDITIONAL WORK

****CONF ALL**

WARNING: MAKE SURE THAT YOU OBEY ALL THE WARNINGS AND ALL THE CAUTIONS INCLUDED IN THE REFERENCED PROCEDURES.

CAUTION: ALWAYS OBEY THE PRECAUTIONS THAT FOLLOW TO KEEP ELECTRICAL WIRING IN A SATISFACTORY CONDITION (ELECTRICALLY AND MECHANICALLY SERVICEABLE). WHEN YOU DO MAINTENANCE WORK, REPAIRS OR MODIFICATIONS, ALWAYS KEEP ELECTRICAL WIRING, COMPONENTS AND THE WORK AREA AS CLEAN AS POSSIBLE. TO DO THIS :

- PUT PROTECTION, SUCH AS PLASTIC SHEETING, CLOTHS, ETC; AS NECESSARY ON WIRING AND COMPONENTS.
- REGULARLY REMOVE ALL SHAVINGS, UNWANTED MATERIAL AND OTHER CONTAMINATION.

THESE PRECAUTIONS WILL DECREASE THE RISK OF CONTAMINATION AND DAMAGE TO THE ELECTRICAL WIRING INSTALLATION.

IN CASE OF CONTAMINATION REFER TO ESPM 20-55-00.

NOTE: The accomplishment instructions of this Service Bulletin include procedures given in other documents or in other sections of the Service Bulletin. When the words 'refer to' are used and the operator has a procedure accepted by the local authority he belongs to, the accepted alternative procedure can be used. When the words 'in accordance with' are used then the given procedure must be followed.

NOTE: The access and close-up instructions, not comprising return to service tests, in this Service Bulletin do not constitute or affect the technical intent of the Service Bulletin. Operators can therefore, as deemed necessary, omit or add access and/or close-up steps to add flexibility to their maintenance operations as long as the technical intent of the Service Bulletin is met within the set parameters.

NOTE: Manual titles given in the accomplishment instructions are referred to by acronyms. Refer to paragraph 1.J., References, for the definition of acronyms.

NOTE: Discarded parts must be managed at the operator's discretion.

NOTE: This Service Bulletin contains the instructions for the on-aircraft maintenance necessary to ensure the continued airworthiness of the aircraft.

For any deviations to the instructions contained herein, contact AIRBUS for further instructions and approval.

NOTE: The purpose of flowcharts is to supplement the information given in the Procedure and Compliance paragraphs and not to serve as the primary source for tasks or compliance times given in this Service Bulletin.

Task Associated Data****CONF 001**

Zone	
From 141 to 540	
From 142 to 640	
Access	
Panel	147AZ 148AZ 540AB 540BZ 640AB 640BZ
Manpower	
TOTAL MANHOURS	59.00
ELAPSED TIME (HOURS)	15.50

****CONF 002**

Zone	
From 141 to 540	
From 142 to 640	
Access	
Panel	147AZ 148AZ 540AB 540BZ 640AB 640BZ
Manpower	
TOTAL MANHOURS	37.00
ELAPSED TIME (HOURS)	10.00

****CONF ALL****A. GENERAL******CONF 001****(1) Subtask 571140-910-003-001 - Standard Practices**

Manpower Resources	
Skills	NON SPECIFIC

References	
Consumable Material List (CML)	
Structural Repair Manual (SRM)	51-42-11

(a) For the specification of Material Numbers (Mat. No.) given in this Service Bulletin, refer to the Consumable Material List (CML).

(b) Remove and install fasteners in accordance with SRM 51-42-11.

(2) Subtask 571140-839-003-001 - Administrative

Manpower Resources	
Skills	NON SPECIFIC

Write in the applicable aircraft records that you have done all the work given in the Service Bulletin.

****CONF 002****(1) Subtask 571140-910-003-001 - Standard Practices**

Manpower Resources	
Skills	NON SPECIFIC

References	
Consumable Material List (CML)	
Structural Repair Manual (SRM)	51-42-11

- (a) For the specification of Material Numbers (Mat. No.) given in this Service Bulletin, refer to the Consumable Material List (CML).
- (b) Remove and install fasteners in accordance with SRM 51-42-11.

(2) Subtask 571140-839-003-001 - Administrative

Manpower Resources	
Skills	NON SPECIFIC

Write in the applicable aircraft records that you have done all the work given in the Service Bulletin.

****CONF ALL****B. PREPARATION******CONF 001****(1) Subtask 571140-941-002-001 - Job Set-up**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 12-34-24-869-002 Task 28-00-00-910-001 Task 28-10-00-910-004 Task 28-25-00-650-001 Task 32-12-00-010-001

NOTE: The items given in this note shall be considered as the basic Aircraft configuration before you start a maintenance task:

- Aircraft on the ground resting on landing gear (the ground safety locks and the wheel chocks are in position on the landing gear),
- Engine shut down, thrust reversers closed and locked,
- Aircraft in clean configuration,
- Parking brake applied,

- Aircraft electrical network de-energized,
 - Hydraulic systems depressurized,
 - Access to the cockpit and cabin is available,
 - All circuit breakers are in closed position,
 - All controls in NORM, AUTO or OFF position.
- (a) Make sure that the aircraft is electrically grounded, refer to AMM Task 12-34-24-869-002.
- (b) Put an access platform in position at zones Z540 and Z640.
- (c) Defuel the center tank, refer to AMM Task 28-25-00-650-001.
- (d) Defuel the LH and RH wing tanks, refer to AMM Task 28-25-00-650-001.
- (e) Degas and ventilate the center tank and the wing tanks, refer to AMM Task 28-00-00-910-001 and AMM Task 28-10-00-910-004.
- (f) Open the main landing gear doors, refer to AMM Task 32-12-00-010-001.

(2) Subtask 571140-010-003-001 - Remove/Open for Access on the LH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-000-001 Task 28-21-54-000-001 Task 53-35-11-000-001 Task 57-17-11-000-001 Task 57-27-11-000-001 Task 57-27-12-000-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

- (a) Remove the access cover 147AZ, refer to AMM Task 57-17-11-000-001.
- (b) Remove the wing access panel 540AB, refer to AMM Task 57-27-11-000-001.
- (c) Open the fuel surge access door 540BZ, refer to AMM Task 57-27-12-000-001.
- (d) Remove the fuel pump suction valve, refer to AMM Task 28-21-42-000-001.
- (e) Remove the covers Item (80) and (81) of the fuel pump strainer FIN 6QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-000-001.
- (f) If necessary, remove the panels 191LB and 147CB on the wing to fuselage, refer to AMM Task 53-35-11-000-001.

(3) Subtask 571140-010-004-001 - Remove/Open for Access on the RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-000-001 Task 28-21-54-000-801 Task 53-35-11-000-001 Task 57-17-11-000-001 Task 57-27-11-000-001 Task 57-27-12-000-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

- (a) Remove the access cover 148AZ, refer to AMM Task 57-17-11-000-001.
- (b) Remove the wing access panel 640AB, refer to AMM Task 57-27-11-000-001.
- (c) Open the fuel surge access door 640BZ, refer to AMM Task 57-27-12-000-001.
- (d) Remove the fuel pump suction valve, refer to AMM Task 28-21-42-000-001.
- (e) Remove the covers Item (80) and (81) of the fuel pump strainer FIN 8QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-000-801.
- (f) If necessary, remove the panels 192LB and 148DB on the wing to fuselage, refer to AMM Task 53-35-11-000-001.

(4) Subtask 571140-010-007-001 - Remove the Fuel Strainer, LH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- Remove for access the fuel strainer for LH side:
Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)
In accordance with SRM 51-42-11.

1	Strainer Assembly	Item (55)	Retain
1	Strainer Assembly	Item (57)	Retain

attached with:

8	Bolt	Item (66)	Retain
2	Bolt	Item (67)	Retain
2	Bolt	Item (68)	Retain
2	Washer	Item (69)	Discard

(5) Subtask 571140-010-008-001 - Remove the Fuel Strainer, RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- Remove for access the fuel strainer for RH side:
Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)
In accordance with SRM 51-42-11.

1	Strainer Assembly	Item (56)	Retain
1	Strainer Assembly	Item (58)	Retain
attached with:			
8	Bolt	Item (66)	Retain
2	Bolt	Item (67)	Retain
2	Bolt	Item (68)	Retain
2	Washer	Item (69)	Discard

****CONF 002**

(1) Subtask 571140-941-002-001 - Job Set-up

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 12-34-24-869-002 Task 28-00-00-910-001 Task 28-10-00-910-004 Task 28-25-00-650-001 Task 32-12-00-010-001

NOTE: The items given in this note shall be considered as the basic Aircraft configuration before you start a maintenance task:

- Aircraft on the ground resting on landing gear (the ground safety locks and the wheel chocks are in position on the landing gear),
- Engine shut down, thrust reversers closed and locked,
- Aircraft in clean configuration,
- Parking brake applied,
- Aircraft electrical network de-energized,
- Hydraulic systems depressurized,
- Access to the cockpit and cabin is available,
- All circuit breakers are in closed position,
- All controls in NORM, AUTO or OFF position.

- (a) Make sure that the aircraft is electrically grounded, refer to AMM Task 12-34-24-869-002.
- (b) Put an access platform in position at zones Z540 and Z640.
- (c) Defuel the center tank, refer to AMM Task 28-25-00-650-001.
- (d) Defuel the LH and RH wing tanks, refer to AMM Task 28-25-00-650-001.
- (e) Degas and ventilate the center tank and the wing tanks, refer to AMM Task 28-00-00-910-001 and AMM Task 28-10-00-910-004.
- (f) Open the main landing gear doors, refer to AMM Task 32-12-00-010-001.

(2) Subtask 571140-010-003-001 - Remove/Open for Access on the LH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-000-001 Task 28-21-54-000-001 Task 53-35-11-000-001 Task 57-17-11-000-001 Task 57-27-11-000-001 Task 57-27-12-000-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

- (a) Remove the access cover 147AZ, refer to AMM Task 57-17-11-000-001.
- (b) Remove the wing access panel 540AB, refer to AMM Task 57-27-11-000-001.
- (c) Open the fuel surge access door 540BZ, refer to AMM Task 57-27-12-000-001.
- (d) Remove the fuel pump suction valve, refer to AMM Task 28-21-42-000-001.
- (e) Remove the covers Item (80) and (81) of the fuel pump strainer FIN 6QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-000-001.
- (f) If necessary, remove the panels 191LB and 147CB on the wing to fuselage, refer to AMM Task 53-35-11-000-001.

(3) Subtask 571140-010-004-001 - Remove/Open for Access on the RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-000-001 Task 28-21-54-000-801 Task 53-35-11-000-001 Task 57-17-11-000-001 Task 57-27-11-000-001 Task 57-27-12-000-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

- (a) Remove the access cover 148AZ, refer to AMM Task 57-17-11-000-001.
- (b) Remove the wing access panel 640AB, refer to AMM Task 57-27-11-000-001.
- (c) Open the fuel surge access door 640BZ, refer to AMM Task 57-27-12-000-001.
- (d) Remove the fuel pump suction valve, refer to AMM Task 28-21-42-000-001.
- (e) Remove the covers Item (80) and (81) of the fuel pump strainer FIN 8QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-000-801.

- (f) If necessary, remove the panels 192LB and 148DB on the wing to fuselage, refer to AMM Task 53-35-11-000-001.

(4) Subtask 571140-010-007-001 - Remove the Fuel Strainer, LH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- Remove for access the fuel strainer for LH side:
Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)
In accordance with SRM 51-42-11.

1	Strainer Assembly	Item (55)	Retain
1	Strainer Assembly	Item (57)	Retain
attached with:			
8	Bolt	Item (66)	Retain
2	Bolt	Item (67)	Retain
2	Bolt	Item (68)	Retain
2	Washer	Item (69)	Discard

(5) Subtask 571140-010-008-001 - Remove the Fuel Strainer, RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- Remove for access the fuel strainer for RH side:
Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)
In accordance with SRM 51-42-11.

1	Strainer Assembly	Item (56)	Retain
1	Strainer Assembly	Item (58)	Retain
attached with:			
8	Bolt	Item (66)	Retain
2	Bolt	Item (67)	Retain
2	Bolt	Item (68)	Retain
2	Washer	Item (69)	Discard

****CONF ALL**

C. PROCEDURE

****CONF 001**

- (1) **Subtask 571140-832-003-001 - LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK**

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	21.00
Minimum number of person	2
Subtask elapsed time	10.50
Skills	AIRFRAME

Material necessary to do the job

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Non Aqueous Cleaner	08BAA9	As required	

Kit tool 571140T01R01				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103001000	Gage	1	

References	
In Service Information (ISI)	00.00.00179
Structural Repair Manual (SRM)	51-42-11
Fig. A-FBCAA Thread Damage	Sheet 01
Fig. A-FCEAA Inspection of the Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCFAA LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FFBAA Flow Chart for the ADDITIONAL WORK	Sheet 01

(a) At threshold given in the PLANNING INFORMATION Paragraph E.(2):

Do a detailed inspection of lower panels in center and outer wing box at level of rib1 junction:

In accordance with SRM 51-42-11.

Refer to [Fig. A-FFBAA Sheet 01](#)

Refer to [Fig. A-FBCAA Sheet 01](#).

Refer to [Fig. A-FCEAA Sheet 01](#) to [Fig. A-FCEAA Sheet 03](#)

Refer to [Fig. A-FCFAA Sheet 01](#) to [Fig. A-FCFAA Sheet 04](#)

1 Remove the sealant on the nuts with:

Non Aqueous	08BAA9	As required
Cleaner		

2 Remove the nuts :

Refer to [Fig. A-FCFAA Sheet 01](#) to [Fig. A-FCFAA Sheet 04](#)

11	Nut	NSA5457-7E	Item (10)	Discard
	or			
11	Nut	NAS1727-7E	Item (10)	Discard
	or			
11	Nut	NSA5151-7	Item (10)	Discard
	and			

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2	Nut	NAS1726-7E	Item (11)	Discard
	or			
2	Nut	NSA5050-7	Item (11)	Discard
	and			
16	Nut	NSA5457-8E	Item (13)	Discard
	or			
16	Nut	NAS1727-8E	Item (13)	Discard
	or			
16	Nut	NSA5151-8	Item (13)	Discard
	and			
2	Nut	NAS1726-6E	Item (15)	Discard
	or			
2	Nut	NSA5050-6	Item (15)	Discard

NOTE: Identify the location of each removed nut.

- (b) Check if the taperlok gage fit and move as described in the gage set procedure:

98D57103001000 Gage set 1

- 1 If taperlok gage fit and move:

- a Do a visual inspection of the thread to determine the cause of thread marks found:

<1> If thread marks are found caused by an incorrect installation:

<a> Apply SUBTASK 571140-833-007 001 Replace Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

<2> If no thread marks are found caused by an incorrect installation but thread was damaged during the nut removal:

<a> Apply SUBTASK 571140-833-007 001 Replace Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

<3> If no thread marks are found caused by an incorrect installation and thread was not damaged during the nut removal:

<a> Apply SUBTASK 571140-833-005 001 Replace the Nuts on the LH Side before next flight.

2 If the taperlok gage do not fit and move but thread was damaged during the nut removal:

a Apply SUBTASK 571140-833-007 001 Replace Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

3 If the taperlok gage do not fit and move and thread was not damaged during the nut removal:

a Apply SUBTASK 571140-833-005 001 Replace the Nuts on the LH Side before next flight.

(c) Upload the completed Inspection Report sheet given in APPENDIX 04 - Inspection Report in accordance with ISI 00.00.00179.

NOTE: Uploading the Inspection Report sheet is not Required for Compliance.

(2) Subtask 571140-832-004-001 - RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	21.00
Minimum number of person	2
Subtask elapsed time	10.50
Skills	AIRFRAME

Material necessary to do the job

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Non Aqueous Cleaner	08BAA9	As required	

Kit tool 571140T01R01				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103001000	Gage	1	

References	
In Service Information (ISI)	00.00.00179
Structural Repair Manual (SRM)	51-42-11

References	
Fig. A-FBCAA Thread Damage	Sheet 01
Fig. A-FCEAA Inspection of the Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCGAA RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FFBAA Flow Chart for the ADDITIONAL WORK	Sheet 01

(a) At threshold given in the PLANNING INFORMATION Paragraph E.(2):

Do a detailed inspection of lower panels in center and outer wing box at level of rib1 junction:

In accordance with SRM 51-42-11.

Refer to [Fig. A-FFBAA Sheet 01](#)

Refer to [Fig. A-FBCAA Sheet 01](#).

Refer to [Fig. A-FCEAA Sheet 01](#) to [Fig. A-FCEAA Sheet 03](#)

Refer to [Fig. A-FCGAA Sheet 01](#) to [Fig. A-FCGAA Sheet 04](#)

1 Remove the sealant on the nuts with:

Non Aqueous	08BAA9	As required
Cleaner		

2 Remove the nuts :

Refer to [Fig. A-FCGAA Sheet 01](#) to [Fig. A-FCGAA Sheet 04](#)

11	Nut	NSA5457-7E	Item (10)	Discard
	or			
11	Nut	NAS1727-7E	Item (10)	Discard
	or			
11	Nut	NSA5151-7	Item (10)	Discard
	and			
2	Nut	NAS1726-7E	Item (11)	Discard

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	or			
2	Nut	NSA5050-7	Item (11)	Discard
	and			
16	Nut	NSA5457-8E	Item (13)	Discard
	or			
16	Nut	NAS1727-8E	Item (13)	Discard
	or			
16	Nut	NSA5151-8	Item (13)	Discard
	and			
2	Nut	NAS1726-6E	Item (15)	Discard
	or			
2	Nut	NSA5050-6	Item (15)	Discard

NOTE: Identify the location of each removed nut.

- (b) Check if the taperlok gage fit and move as described in the gage set procedure:

98D57103001000 Gage set 1

- 1 If taperlok gage fit and move:

- a Do a visual inspection of the thread to determine the cause of thread marks found:

<1> If thread marks are found caused by an incorrect installation:

<a> Apply SUBTASK 571140-833-008 001 Replace Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

<2> If no thread marks are found caused by an incorrect installation but thread was damaged during the nut removal:

<a> Apply SUBTASK 571140-833-008 001 Replace Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

<3> If no thread marks are found caused by an incorrect installation and thread was not damaged during the nut removal:

<a> Apply SUBTASK 571140-833-006 001 Replace the Nuts on the RH Side before next flight.

2 If the taperlok gage do not fit and move but thread was damaged during the nut removal:

a Apply SUBTASK 571140-833-008 001 Replace Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

3 If the taperlok gage do not fit and move and thread was not damaged during the nut removal:

a Apply SUBTASK 571140-833-006 001 Replace the Nuts on the RH Side before next flight.

(c) Upload the completed Inspection Report sheet given in APPENDIX 04 - Inspection Report in accordance with ISI 00.00.00179.

NOTE: Uploading the Inspection Report sheet is not Required for Compliance.

****CONF 002**

(1) Subtask 571140-832-003-002 - LH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	10.00
Minimum number of person	2
Subtask elapsed time	5.00
Skills	AIRFRAME

Material necessary to do the job

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Non Aqueous Cleaner	08BAA9	As required	

Kit tool 571140T01R01				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103001000	Gage	1	

References	
In Service Information (ISI)	00.00.00179

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FBCAA Thread Damage	Sheet 01
Fig. A-FCEAB Inspection of the Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCFAB LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FFBAB Flow Chart for the ADDITIONAL WORK	Sheet 01

(a) At threshold given in the PLANNING INFORMATION Paragraph E.(2):

Do a detailed inspection of lower panels in center and outer wing box at level of rib1 junction:

In accordance with SRM 51-42-11.

Refer to [Fig. A-FFBAB Sheet 01](#)

Refer to [Fig. A-FBCAA Sheet 01](#).

Refer to [Fig. A-FCEAB Sheet 01](#) to [Fig. A-FCEAB Sheet 03](#)

Refer to [Fig. A-FCFAB Sheet 01](#) to [Fig. A-FCFAB Sheet 04](#)

1 Remove the sealant on the nuts with:

Non Aqueous Cleaner	08BAA9	As required
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2 Remove the nuts :

Refer to [Fig. A-FCFAB Sheet 01](#) to [Fig. A-FCFAB Sheet 04](#)

7	Nut	NSA5457-7E	Item (10)	Discard
	or			
7	Nut	NAS1727-7E	Item (10)	Discard
	or			
7	Nut	NSA5151-7	Item (10)	Discard
	and			

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2	Nut	NAS1726-7E	Item (11)	Discard
	or			
2	Nut	NSA5050-7	Item (11)	Discard
	and			
16	Nut	NSA5457-8E	Item (13)	Discard
	or			
16	Nut	NAS1727-8E	Item (13)	Discard
	or			
16	Nut	NSA5151-8	Item (13)	Discard
	and			
2	Nut	NAS1726-6E	Item (15)	Discard
	or			
2	Nut	NSA5050-6	Item (15)	Discard

NOTE: Identify the location of each removed nut.

- (b) Check if the taperlok gage fit and move as described in the gage set procedure:

98D57103001000 Gage set 1

- 1 If taperlok gage fit and move:

- a Do a visual inspection of the thread to determine the cause of thread marks found:

<1> If thread marks are found caused by an incorrect installation:

<a> Apply SUBTASK 571140-833-007 002 Replace Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

<2> If no thread marks are found caused by an incorrect installation but thread was damaged during the nut removal:

<a> Apply SUBTASK 571140-833-007 002 Replace Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

<3> If no thread marks are found caused by an incorrect installation and thread was not damaged during the nut removal:

<a> Apply SUBTASK 571140-833-005 002 Replace the the Nuts on the LH Side before next flight.

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2 If the taperlok gage do not fit and move but thread was damaged during the nut removal:

a Apply SUBTASK 571140-833-007 002 Replace Taperlok with the Next Nominal Diameter Taperlok on the LH Side before next flight.

3 If the taperlok gage do not fit and move and thread was not damaged during the nut removal:

a Apply SUBTASK 571140-833-005 002 Replace the Nuts on the LH Side before next flight.

(c) Upload the completed Inspection Report sheet given in APPENDIX 04 - Inspection Report in accordance with ISI 00.00.00179.

NOTE: Uploading the Inspection Report sheet is not Required for Compliance.

(2) Subtask 571140-832-004-002 - RH Side, do a Detailed Inspection in Center and Outer Wing Box at Level of Rib1 junction, ADDITIONAL WORK

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	10.00
Minimum number of person	2
Subtask elapsed time	5.00
Skills	AIRFRAME

Material necessary to do the job

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Non Aqueous Cleaner	08BAA9	As required	

Kit tool 571140T01R01				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103001000	Gage	1	

References	
In Service Information (ISI)	00.00.00179
Structural Repair Manual (SRM)	51-42-11

References	
Fig. A-FBCAA Thread Damage	Sheet 01
Fig. A-FCGAB RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07
Fig. A-FCEAB Inspection of the Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03
Fig. A-FFBAB Flow Chart for the ADDITIONAL WORK	Sheet 01

(a) At threshold given in the PLANNING INFORMATION Paragraph E.(2):

Do a detailed inspection of lower panels in center and outer wing box at level of rib1 junction:

In accordance with SRM 51-42-11.

Refer to [Fig. A-FFBAB Sheet 01](#)

Refer to [Fig. A-FBCAA Sheet 01](#).

Refer to [Fig. A-FCEAB Sheet 01](#) to [Fig. A-FCEAB Sheet 03](#)

Refer to [Fig. A-FCGAB Sheet 01](#) to [Fig. A-FCGAB Sheet 04](#)

1 Remove the sealant on the nuts with:

Non Aqueous	08BAA9	As required
Cleaner		

2 Remove the nuts :

Refer to [Fig. A-FCGAB Sheet 01](#) to [Fig. A-FCGAB Sheet 04](#)

7	Nut	NSA5457-7E	Item (10)	Discard
	or			
7	Nut	NAS1727-7E	Item (10)	Discard
	or			
7	Nut	NSA5151-7	Item (10)	Discard
	and			
2	Nut	NAS1726-7E	Item (11)	Discard

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	or			
2	Nut	NSA5050-7	Item (11)	Discard
	and			
16	Nut	NSA5457-8E	Item (13)	Discard
	or			
16	Nut	NAS1727-8E	Item (13)	Discard
	or			
16	Nut	NSA5151-8	Item (13)	Discard
	and			
2	Nut	NAS1726-6E	Item (15)	Discard
	or			
2	Nut	NSA5050-6	Item (15)	Discard

NOTE: Identify the location of each removed nut.

(b) Check if the taperlok gage fit and move as described in the gage set procedure:

98D57103001000 Gage set 1

1 If taperlok gage fit and move:

a Do a visual inspection of the thread to determine the cause of thread marks found:

<1> If thread marks are found caused by an incorrect installation:

<a> Apply SUBTASK 571140-833-008 002 Replace Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

<2> If no thread marks are found caused by an incorrect installation but thread was damaged during the nut removal:

<a> Apply SUBTASK 571140-833-008 002 Replace Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.

<3> If no thread marks are found caused by an incorrect installation and thread was not damaged during the nut removal:

<a> Apply SUBTASK 571140-833-006 002 Replace the Nuts on the RH Side before next flight.

- 2 If the taperlok gage do not fit and move but thread was damaged during the nut removal:
- a Apply SUBTASK 571140-833-008 002 Replace Taperlok with the Next Nominal Diameter Taperlok on the RH Side before next flight.
- 3 If the taperlok gage do not fit and move and thread was not damaged during the nut removal:
- a Apply SUBTASK 571140-833-006 002 Replace the Nuts on the RH Side before next flight.

- (c) Upload the completed Inspection Report sheet given in APPENDIX 04 - Inspection Report in accordance with ISI 00.00.00179.

NOTE: Uploading the Inspection Report sheet is not Required for Compliance.

****CONF ALL**

D. TEST

****CONF 001**

None

****CONF 002**

None

****CONF ALL**

E. CLOSE-UP

****CONF 001**

None

****CONF 002**

None

Task 571140-833-802-001 - REPAIR ADDITIONAL WORK

****CONF ALL**

WARNING: MAKE SURE THAT YOU OBEY ALL THE WARNINGS AND ALL THE CAUTIONS INCLUDED IN THE REFERENCED PROCEDURES.

CAUTION: ALWAYS OBEY THE PRECAUTIONS THAT FOLLOW TO KEEP ELECTRICAL WIRING IN A SATISFACTORY CONDITION (ELECTRICALLY AND MECHANICALLY SERVICEABLE). WHEN YOU DO MAINTENANCE WORK, REPAIRS OR MODIFICATIONS, ALWAYS KEEP ELECTRICAL WIRING, COMPONENTS AND THE WORK AREA AS CLEAN AS POSSIBLE. TO DO THIS :

- PUT PROTECTION, SUCH AS PLASTIC SHEETING, CLOTHS, ETC; AS NECESSARY ON WIRING AND COMPONENTS.
- REGULARLY REMOVE ALL SHAVINGS, UNWANTED MATERIAL AND OTHER CONTAMINATION.

THESE PRECAUTIONS WILL DECREASE THE RISK OF CONTAMINATION AND DAMAGE TO THE ELECTRICAL WIRING INSTALLATION.

IN CASE OF CONTAMINATION REFER TO ESPM 20-55-00.

NOTE: The accomplishment instructions of this Service Bulletin include procedures given in other documents or in other sections of the Service Bulletin. When the words 'refer to' are used and the operator has a procedure accepted by the local authority he belongs to, the accepted alternative procedure can be used. When the words 'in accordance with' are used then the given procedure must be followed.

NOTE: The access and close-up instructions, not comprising return to service tests, in this Service Bulletin do not constitute or affect the technical intent of the Service Bulletin. Operators can therefore, as deemed necessary, omit or add access and/or close-up steps to add flexibility to their maintenance operations as long as the technical intent of the Service Bulletin is met within the set parameters.

NOTE: Manual titles given in the accomplishment instructions are referred to by acronyms. Refer to paragraph 1.J., References, for the definition of acronyms.

NOTE: This Service Bulletin contains the instructions for the on-aircraft maintenance necessary to ensure the continued airworthiness of the aircraft.

For any deviations to the instructions contained herein, contact AIRBUS for further instructions and approval.

Task Associated Data

****CONF 001**

Zone
From 141 to 540
From 142 to 640

Access	
Panel	147AZ 148AZ 540AB 540BZ 640AB 640BZ
Manpower	
TOTAL MANHOURS	65.50
ELAPSED TIME (HOURS)	20.50

****CONF 002**

Zone	
From 141 to 540 From 142 to 640	
Access	
Panel	147AZ 148AZ 540AB 540BZ 640AB 640BZ
Manpower	
TOTAL MANHOURS	41.50
ELAPSED TIME (HOURS)	14.50

****CONF ALL****A. GENERAL******CONF 001****(1) Subtask 571140-910-004-001 - Standard Practices**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Consumable Material List (CML)	
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00 51-75-12

- (a) Apply the sealant in accordance with SRM 51-24-00.
- (b) For the specification of Material Numbers (Mat. No.) given in this Service Bulletin, refer to the Consumable Material List (CML).
- (c) Remove and install fasteners in accordance with SRM 51-42-11.
- (d) Get alternative and substitute fastener data in accordance with SRM 51-43-00.
- (e) Renew the protective finish in accordance with SRM 51-75-12.
- (f) To torque tighten the standard threaded fasteners, refer to AMM Task 20-21-11-911-001.

(2) Subtask 571140-839-004-001 - Administrative

Manpower Resources	
Skills	NON SPECIFIC

Write in the applicable aircraft records that you have done all the work given in the Service Bulletin.

****CONF 002****(1) Subtask 571140-910-004-001 - Standard Practices**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Consumable Material List (CML)	
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00 51-75-12

- (a) Apply the sealant in accordance with SRM 51-24-00.
- (b) For the specification of Material Numbers (Mat. No.) given in this Service Bulletin, refer to the Consumable Material List (CML).
- (c) Remove and install fasteners in accordance with SRM 51-42-11.
- (d) Get alternative and substitute fastener data in accordance with SRM 51-43-00.
- (e) Renew the protective finish in accordance with SRM 51-75-12.
- (f) To torque tighten the standard threaded fasteners, refer to AMM Task 20-21-11-911-001.

(2) Subtask 571140-839-004-001 - Administrative

Manpower Resources	
Skills	NON SPECIFIC

Write in the applicable aircraft records that you have done all the work given in the Service Bulletin.

****CONF ALL****B. PREPARATION******CONF 001**

None

****CONF 002**

None

****CONF ALL**

C. PROCEDURE

****CONF 001**

(1) Subtask 571140-833-005-001 - Replace the Nuts on the LH Side, ADDITIONAL WORK

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	10.00
Minimum number of person	2
Subtask elapsed time	5.00
Skills	AIRFRAME

Material necessary to do the job

Supplementary kit 571140A01S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
10	ASNA2532-7	11	NUT	
11	ASNA2531-7	2	NUT	
13	ASNA2532-8	16	NUT	
15	ASNA2531-6	2	NUT	
	NSA5372-716AX	31	WASHER	
	NSA5372-716BX	31	WASHER	
	NSA5372-716CX	31	WASHER	
	NSA5372-716DX	31	WASHER	
	NSA5372-716EX	31	WASHER	

Supplementary kit 571140A01S03				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
	NSA5372-616AX	2	WASHER	
	NSA5372-616BX	2	WASHER	
	NSA5372-616CX	2	WASHER	
	NSA5372-616DX	2	WASHER	
	NSA5372-616EX	2	WASHER	

ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
	NSA5372-816AX	16	WASHER	
	NSA5372-816BX	16	WASHER	
	NSA5372-816CX	16	WASHER	
	NSA5372-816DX	16	WASHER	
	NSA5372-816EX	16	WASHER	

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11 51-43-00
Fig. A-FCCAA Definition of the Washer Thickness for the Original Taperlok	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCFAA LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

NOTE: The nuts are removed during the inspection.

(a) Install the nuts.

In accordance with SRM 51-42-11 and SRM 51-43-00.

Refer to AMM Task 20-21-11-911-001.

Refer to [Fig. A-FCFAA Sheet 01](#), [Fig. A-FCFAA Sheet 02](#), [Fig. A-FCFAA Sheet 05](#), [Fig. A-FCFAA Sheet 06](#), [Fig. A-FCFAA Sheet 07](#)

- The supplementary kit 538096A01S02 includes:

11	Nut	ASNA2532-7	Item 10
2	Nut	ASNA2531-7	Item 11
16	Nut	ASNA2532-8	Item 13
2	Nut	ASNA2531-6	Item 15

with

31 Washer NSA5372-716AX

and/or

31 Washer NSA5372-716BX

and/or

31 Washer NSA5372-716CX

and/or

31 Washer NSA5372-716DX

and/or

31 Washer NSA5372-716EX

- The supplementary kit 538096A01S03 includes:

2 Washer NSA5372-616AX

and/or

2 Washer NSA5372-616BX

and/or

2 Washer NSA5372-616CX

and/or

2 Washer NSA5372-616DX

and/or

2 Washer NSA5372-616EX

and

16 Washer NSA5372-816AX

and/or

16 Washer NSA5372-816BX

and/or

16 Washer NSA5372-816CX

and/or

16 Washer NSA5372-816DX

and/or

16 Washer NSA5372-816EX

NOTE: Install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCCAA Sheet 01](#).

NOTE: If necessary, the installation of the three washers is approved in this condition.

NOTE: Torque the nuts between 3.80daN and 4.60daN (342lbf.in. and 411lbf.in.).

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(2) Subtask 571140-833-006-001 - Replace the Nuts on the RH Side, ADDITIONAL WORK

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	10.00
Minimum number of person	2
Subtask elapsed time	5.00
Skills	AIRFRAME

Material necessary to do the job

Supplementary kit 571140A01S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
10	ASNA2532-7	11	NUT	
11	ASNA2531-7	2	NUT	
13	ASNA2532-8	16	NUT	
15	ASNA2531-6	2	NUT	
	NSA5372-716AX	31	WASHER	
	NSA5372-716BX	31	WASHER	
	NSA5372-716CX	31	WASHER	
	NSA5372-716DX	31	WASHER	
	NSA5372-716EX	31	WASHER	

Supplementary kit 571140A01S03				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
	NSA5372-616AX	2	WASHER	
	NSA5372-616BX	2	WASHER	
	NSA5372-616CX	2	WASHER	
	NSA5372-616DX	2	WASHER	
	NSA5372-616EX	2	WASHER	
	NSA5372-816AX	16	WASHER	
	NSA5372-816BX	16	WASHER	
	NSA5372-816CX	16	WASHER	
	NSA5372-816DX	16	WASHER	
	NSA5372-816EX	16	WASHER	

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11 51-43-00
Fig. A-FCCAA Definition of the Washer Thickness for the Original Taperlok	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCGAA RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

NOTE: The nuts are removed during the inspection.

(a) Install the nuts.

In accordance with SRM 51-42-11 and SRM 51-43-00.

Refer to AMM Task 20-21-11-911-001.

Refer to [Fig. A-FCGAA Sheet 01](#), [Fig. A-FCGAA Sheet 02](#), [Fig. A-FCGAA Sheet 05](#), [Fig. A-FCGAA Sheet 06](#), [Fig. A-FCGAA Sheet 07](#)

- The supplementary kit 538096A01S02 includes:

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11	Nut	ASNA2532-7	Item 10
2	Nut	ASNA2531-7	Item 11
16	Nut	ASNA2532-8	Item 13
2	Nut	ASNA2531-6	Item 15

with

31	Washer	NSA5372-716AX
----	--------	---------------

and/or

31	Washer	NSA5372-716BX
----	--------	---------------

and/or

31	Washer	NSA5372-716CX
----	--------	---------------

and/or

31	Washer	NSA5372-716DX
----	--------	---------------

and/or

31	Washer	NSA5372-716EX
----	--------	---------------

- The supplementary kit 538096A01S03 includes:

2	Washer	NSA5372-616AX
---	--------	---------------

and/or

2	Washer	NSA5372-616BX
---	--------	---------------

and/or

2	Washer	NSA5372-616CX
---	--------	---------------

and/or

2	Washer	NSA5372-616DX
---	--------	---------------

and/or

2	Washer	NSA5372-616EX
---	--------	---------------

and

16	Washer	NSA5372-816AX
----	--------	---------------

and/or

16	Washer	NSA5372-816BX
----	--------	---------------

and/or

16 Washer NSA5372-816CX

and/or

16 Washer NSA5372-816DX

and/or

16 Washer NSA5372-816EX

NOTE: Install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCCAA Sheet 01](#).

NOTE: Install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCCAA Sheet 01](#).

NOTE: If necessary, the installation of the three washers is approved in this condition.

NOTE: Torque the nuts between 3.80daN and 4.60daN (342lbf.in. and 411lbf.in.).

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(3) Subtask 571140-833-007-001 - Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side, **ADDITIONAL WORK**

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

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Manpower Resources	
Manhours	13.00
Minimum number of person	2
Subtask elapsed time	6.50
Skills	AIRFRAME NON DESTRUCTIVE TESTING

Material necessary to do the job

Supplementary kit 571140A01S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
60	EN6115T2-3	14	BOLT	
61	ASNA2536-2	17	NUT	
62	EN6115T2-2	3	BOLT	
65	NAS1149D0332K	8	WASHER	

Component COMPA03				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
11	ASNA2531-7	2	NUT	
32	ABS1418K8-14	4	SCREW	
34	ABS1418K8-19	4	SCREW	
35	ASNA2532-8	11	NUT	
37	ABS1418K8-26	1	SCREW	
43	ABS1418K8-25	1	SCREW	
44	ASNA2531-8	2	NUT	
73	ABS1418K9-15	2	SCREW	
74	ABS1418K9-16	8	SCREW	
75	ABS1418K9-17	6	SCREW	
76	ASNA2532-9	16	NUT	
78	ABS1418K8-20	2	SCREW	
80	ABS1418K7-23	2	SCREW	
82	ABS1418K8-22	1	SCREW	
	NSA5372-716AX	2	WASHER	
	NSA5372-716BX	2	WASHER	
	NSA5372-716EX	2	WASHER	
	NSA5372-816AX	13	WASHER	
	NSA5372-816BX	13	WASHER	
	NSA5372-816EX	13	WASHER	
	NSA5372-916AX	16	WASHER	
	NSA5372-916BX	16	WASHER	
	NSA5372-916EX	16	WASHER	

NOTE: The quantities given in the component table are for all the taperlocks replacement.

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NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Wash Primer - Structure	04CMA2	As required	
	Primer Polyurethane Paint - Corrosion Inhibiting	04EAC2	As required	
	Top Coat Polyurethane - External Structure	04JAA3	As required	
	Polysulfide Sealant-General Purpose Brushable	06AAA1	As required	
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

Kit tool 571140T02R00				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103002000	Test	1	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Drawing	R57102001
Non Destructive Test Manual (NTM)	51-10-01 PART 6
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00
Fig. A-FCDA Definition of the Washer Thickness for the New Taperlok	Sheet 01
Fig. A-FDAAA Removal/Installation of the Cooling Duct	Sheet 01
Fig. A-FDBAA Removal/Installation of the Belly Fairing Structure	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCFAA LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

In accordance with SRM 51-42-11, SRM 51-43-00 and SRM 51-24-00.

Refer to AMM Task 20-21-11-911-001.

Refer to NTM 51-10-01 PART 6

Refer to [Fig. A-FCFAA Sheet 01](#) to [Fig. A-FCFAA Sheet 07](#)

(a) Depending on the result of the inspection, remove for access :

1 Remove the cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (53)	Retain
1	Duct	Item (52)	Retain
	attached with		
1	Clamp	Item (63)	Retain
8	Screw	Item (64)	Retain
8	Washer	Item (65)	Discard

2 Remove the structure of the belly fairing.

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (51)	Retain
	attached with		
3	Rivet	Item (62)	Discard
7	Rivet	Item (60)	Discard
10	Bush	Item (61)	Discard
	and/or		
1	Web 4 assy	Item (50)	Retain
	attached with		
7	Rivet	Item (60)	Discard
7	Bush	Item (61)	Discard

(b) In accordance with the result of the inspection, remove the taperlok(s) :

Refer to [Fig. A-FCFAA Sheet 01](#) to [Fig. A-FCFAA Sheet 04](#)

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4	Screw	Item (7)	Discard
4	Screw	Item (3)	Discard
1	Screw	Item (14)	Discard
1	Screw	Item (12)	Discard
2	Screw	Item (70)	Discard
8	Screw	Item (71)	Discard
6	Screw	Item (72)	Discard
2	Screw	Item (77)	Discard
2	Screw	Item (79)	Discard
1	Screw	Item (81)	Discard

NOTE: The nuts are removed during the inspection.

- (c) Do a rototest inspection of the holes for cracks, refer to NTM 51-10-01 PART 6.

1 If cracks are found, contact AIRBUS before any further work.

2 If no crack is found, do the following steps.

- (d) Oversize the hole to the next nominal diameter in accordance with the drilling procedure given in the Repair Instruction Drawing R57102001 and use the drilling tool SOA-TL-R57102001B.

NOTE: To prevent any accidental damage, we recommend to train to oversize the taperlok holes on the test specimen.

98D57103002000 Test specimen 1

- (e) Depending on the removal, install :

Refer to [Fig. A-FCFAA Sheet 01](#), [Fig. A-FCFAA Sheet 02](#), [Fig. A-FCFAA Sheet 05](#), [Fig. A-FCFAA Sheet 06](#), [Fig. A-FCFAA Sheet 07](#)

4	Screw	ABS1418K8-19	Item 34
4	Screw	ABS1418K8-14	Item 32
2	Screw	ABS1418K8-20	Item 78
1	Screw	ABS1418K8-22	Item 82

with

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11	Nut	ASNA2532-8	Item 35
	and if necessary		
11	Washer	NSA5372-816AX	
	and/or		
11	Washer	NSA5372-816EX	
	and/or		
11	Washer	NSA5372-816BX	
	and		
1	Screw	ABS1418K8-25	Item 43
1	Screw	ABS1418K8-26	Item 37
	with		
2	Nut	ASNA2531-8	Item 44
	and if necessary		
2	Washer	NSA5372-816AX	
	and/or		
2	Washer	NSA5372-816EX	
	and/or		
2	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K9-15	Item 73
8	Screw	ABS1418K9-16	Item 74
6	Screw	ABS1418K9-17	Item 75
	with		
16	Nut	ASNA2532-9	Item 76
	and if necessary		
16	Washer	NSA5372-916AX	
	and/or		
16	Washer	NSA5372-916EX	

	and/or		
16	Washer	NSA5372-916BX	
	and		
2	Screw	ABS1418K7-23	Item 80
	with		
2	Nut	ASNA2531-7	Item 11
	and if necessary		
2	Washer	NSA5372-716AX	
	and/or		
2	Washer	NSA5372-716EX	
	and/or		
2	Washer	NSA5372-716BX	

NOTE: Install the screw in accordance with the installation principle given in the repair instruction R57102001.

NOTE: If necessary, install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCDA Sheet 01](#).

NOTE: Wet assembly, apply :

Polysulfide Sealant-	06ABB1	As required
Fuel Tank Fillet		

NOTE: Torque the nuts between 5.60daN and 6.70daN (493lbf.in. and 592lbf.in.).

NOTE: Outside the wing center box, apply on the screws :

Primer	04EAC2	As required
Polyurethane Paint		
- Corrosion		
Inhibiting		

Wash Primer -	04CMA2	As required
Structure		

Top Coat	04JAA3	As required
Polyurethane -		
External Structure		

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(f) If removed, install :

1 The structure of the belly fairing

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (51)	Retained at removal
---	----------------	-----------	---------------------

with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

3	Bolt	EN6115T2-2	Item 62
---	------	------------	---------

10	Nut	ASNA2536-2	Item 61
----	-----	------------	---------

and/or

1	Web 4 assy	Item (50)	Retained at removal
---	------------	-----------	---------------------

attached with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

7	Nut	ASNA2536-2	Item 61
---	-----	------------	---------

NOTE: Apply to the interface of the structure and all the parts :

Polysulfide Sealant- 06AAA1 As required
General Purpose
Brushable

2 The cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (53)	Retained at removal
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1	Duct	Item (52)	Retained at removal
---	------	-----------	---------------------

with

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1	Clamp	Item (63)	Retained at removal
8	Screw	Item (64)	Retained at removal
8	Washer	NAS1149D0332K	Item 65

(4) Subtask 571140-833-008-001 - Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side, ADDITIONAL WORK

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	13.00
Minimum number of person	2
Subtask elapsed time	6.50
Skills	AIRFRAME NON DESTRUCTIVE TESTING

Material necessary to do the job

Supplementary kit 571140A01S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
60	EN6115T2-3	14	BOLT	
61	ASNA2536-2	17	NUT	
62	EN6115T2-2	3	BOLT	
65	NAS1149D0332K	8	WASHER	

Component COMPA03				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
11	ASNA2531-7	2	NUT	
32	ABS1418K8-14	4	SCREW	
34	ABS1418K8-19	4	SCREW	
35	ASNA2532-8	11	NUT	
37	ABS1418K8-26	1	SCREW	
43	ABS1418K8-25	1	SCREW	
44	ASNA2531-8	2	NUT	
73	ABS1418K9-15	2	SCREW	
74	ABS1418K9-16	8	SCREW	

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ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
75	ABS1418K9-17	6	SCREW	
76	ASNA2532-9	16	NUT	
78	ABS1418K8-20	2	SCREW	
80	ABS1418K7-23	2	SCREW	
82	ABS1418K8-22	1	SCREW	
	NSA5372-716AX	2	WASHER	
	NSA5372-716BX	2	WASHER	
	NSA5372-716EX	2	WASHER	
	NSA5372-816AX	13	WASHER	
	NSA5372-816BX	13	WASHER	
	NSA5372-816EX	13	WASHER	
	NSA5372-916AX	16	WASHER	
	NSA5372-916BX	16	WASHER	
	NSA5372-916EX	16	WASHER	

NOTE: The quantities given in the component table are for all the taperlocks replacement.

NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Wash Primer - Structure	04CMA2	As required	
	Primer Polyurethane Paint - Corrosion Inhibiting	04EAC2	As required	
	Top Coat Polyurethane - External Structure	04JAA3	As required	
	Polysulfide Sealant-General Purpose Brushable	06AAA1	As required	
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

Kit tool 571140T02R00				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103002000	Test	1	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Drawing	R57102001

References	
Non Destructive Test Manual (NTM)	51-10-01 PART 6
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00
Fig. A-FCDA Definition of the Washer Thickness for the New Taperlok	Sheet 01
Fig. A-FDAAA Removal/Installation of the Cooling Duct	Sheet 01
Fig. A-FDBAA Removal/Installation of the Belly Fairing Structure	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCGAA RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

In accordance with SRM 51-42-11, SRM 51-43-00 and SRM 51-24-00.

Refer to AMM Task 20-21-11-911-001.

Refer to NTM 51-10-01 PART 6

Refer to [Fig. A-FCGAA Sheet 01](#) to [Fig. A-FCGAA Sheet 07](#)

(a) Depending on the result of the inspection, remove for access :

1 Remove the cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (54)	Retain
1	Duct	Item (52)	Retain
	attached with		
1	Clamp	Item (63)	Retain
8	Screw	Item (64)	Retain
8	Washer	Item (65)	Discard

2 Remove the structure of the belly fairing.

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

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1	Channel-Formed	Item (171)	Retain
	attached with		
3	Rivet	Item (62)	Discard
7	Rivet	Item (60)	Discard
10	Bush	Item (61)	Discard
	and/or		
1	Web 4 assy	Item (170)	Retain
	attached with		
7	Rivet	Item (60)	Discard
7	Bush	Item (61)	Discard

(b) In accordance with the result of the inspection, remove the taperlok(s) :

Refer to [Fig. A-FCGAA Sheet 01](#) to [Fig. A-FCGAA Sheet 04](#)

4	Screw	Item (7)	Discard
4	Screw	Item (3)	Discard
1	Screw	Item (14)	Discard
1	Screw	Item (12)	Discard
2	Screw	Item (70)	Discard
8	Screw	Item (71)	Discard
6	Screw	Item (72)	Discard
2	Screw	Item (77)	Discard
2	Screw	Item (79)	Discard
1	Screw	Item (81)	Discard

NOTE: The nuts are removed during the inspection.

(c) Do a rototest inspection of the holes for cracks refer to NTM 51-10-01 PART 6.

1 If cracks are found, contact AIRBUS before any further work.

2 If no crack is found, do the following steps.

- (d) Oversize the hole to the next nominal diameter in accordance with the drilling procedure given in the Repair Instruction Drawing R57102001 and use the drilling tool SOA-TL-R57102001B.

NOTE: To prevent any accidental damage, we recommend to train to oversize the taperlok holes on the test specimen.

98D57103002000 Test specimen 1

- (e) Depending on the removal, install :

Refer to [Fig. A-FCGAA Sheet 01](#), [Fig. A-FCGAA Sheet 02](#), [Fig. A-FCGAA Sheet 05](#), [Fig. A-FCGAA Sheet 06](#), [Fig. A-FCGAA Sheet 07](#)

4	Screw	ABS1418K8-19	Item 34
4	Screw	ABS1418K8-14	Item 32
2	Screw	ABS1418K8-20	Item 78
1	Screw	ABS1418K8-22	Item 82
	with		
11	Nut	ASNA2532-8	Item 35
	and if necessary		
11	Washer	NSA5372-816AX	
	and/or		
11	Washer	NSA5372-816EX	
	and/or		
11	Washer	NSA5372-816BX	
	and		
1	Screw	ABS1418K8-25	Item 43
1	Screw	ABS1418K8-26	Item 37
	with		
2	Nut	ASNA2531-8	Item 44
	and if necessary		
2	Washer	NSA5372-816AX	
	and/or		
2	Washer	NSA5372-816EX	

	and/or		
2	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K9-15	Item 73
8	Screw	ABS1418K9-16	Item 74
6	Screw	ABS1418K9-17	Item 75
	with		
16	Nut	ASNA2532-9	Item 76
	and if necessary		
16	Washer	NSA5372-916AX	
	and/or		
16	Washer	NSA5372-916EX	
	and/or		
16	Washer	NSA5372-916BX	
	and		
2	Screw	ABS1418K7-23	Item 80
	with		
2	Nut	ASNA2531-7	Item 11
	and if necessary		
2	Washer	NSA5372-716AX	
	and/or		
2	Washer	NSA5372-716EX	
	and/or		
2	Washer	NSA5372-716BX	

NOTE: Install the screw in accordance with the installation principle given in the repair instruction R57102001.

NOTE: If necessary, install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCDA Sheet 01](#).

NOTE: Wet assembly, apply :

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Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

NOTE: Torque the nuts between 5.60daN and 6.70daN (493lbf.in. and 592lbf.in.).

NOTE: Outside the wing center box, apply on the screws :

Primer 04EAC2 As required
Polyurethane Paint
- Corrosion
Inhibiting

Wash Primer - 04CMA2 As required
Structure

Top Coat 04JAA3 As required
Polyurethane -
External Structure

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(f) If removed, install :

1 The structure of the belly fairing

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (171)	Retained at removal
---	----------------	------------	------------------------

with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

3	Bolt	EN6115T2-2	Item 62
---	------	------------	---------

10	Nut	ASNA2536-2	Item 61
----	-----	------------	---------

and/or

1	Web 4 assy	Item (170)	Retained at removal
---	------------	------------	------------------------

attached with

7	Bolt	EN6115T2-3	Item 60
---	------	------------	---------

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7 Nut ASNA2536-2 Item 61

NOTE: Apply to the interface of the structure and all the parts :

Polysulfide Sealant- 06AAA1 As required
General Purpose
Brushable

2 The cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (54)	Retained at removal
1	Duct	Item (52)	Retained at removal
	with		
1	Clamp	Item (63)	Retained at removal
8	Screw	Item (64)	Retained at removal
8	Washer	NAS1149D0332K Item 65	

****CONF 002**

(1) Subtask 571140-833-005-002 - Replace the Nuts on the LH Side, ADDITIONAL WORK

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	5.00
Minimum number of person	2
Subtask elapsed time	2.50
Skills	AIRFRAME

Material necessary to do the job

Supplementary kit 571140A02S02

ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
10	ASNA2532-7	7	NUT	
11	ASNA2531-7	2	NUT	
13	ASNA2532-8	16	NUT	
15	ASNA2531-6	2	NUT	
	NSA5372-716AX	27	WASHER	
	NSA5372-716BX	27	WASHER	
	NSA5372-716CX	27	WASHER	
	NSA5372-716DX	27	WASHER	
	NSA5372-716EX	27	WASHER	

Supplementary kit 571140A02S03

ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
	NSA5372-616AX	2	WASHER	
	NSA5372-616BX	2	WASHER	
	NSA5372-616CX	2	WASHER	
	NSA5372-616DX	2	WASHER	
	NSA5372-616EX	2	WASHER	
	NSA5372-816AX	16	WASHER	
	NSA5372-816BX	16	WASHER	
	NSA5372-816CX	16	WASHER	
	NSA5372-816DX	16	WASHER	
	NSA5372-816EX	16	WASHER	

Consumable CMLA01

ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

References

Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11 51-43-00
Fig. A-FCCAA Definition of the Washer Thickness for the Original Taperlok	Sheet 01 Sheet 02 Sheet 03

References	
Fig. A-FCFAB LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01
	Sheet 02
	Sheet 03
	Sheet 04
	Sheet 05
	Sheet 06
	Sheet 07

NOTE: The nuts are removed during the inspection.

(a) Install the nuts.

In accordance with SRM 51-42-11 and SRM 51-43-00.

Refer to AMM Task 20-21-11-911-001.

Refer to [Fig. A-FCFAB Sheet 01](#), [Fig. A-FCFAB Sheet 02](#), [Fig. A-FCFAB Sheet 05](#),
[Fig. A-FCFAB Sheet 06](#), [Fig. A-FCFAB Sheet 07](#)

- The supplementary kit 538096A02S02 includes:

7	Nut	ASNA2532-7	Item 10
2	Nut	ASNA2531-7	Item 11
16	Nut	ASNA2532-8	Item 13
2	Nut	ASNA2531-6	Item 15
	with		
27	Washer	NSA5372-716AX	
	and/or		
27	Washer	NSA5372-716BX	
	and/or		
27	Washer	NSA5372-716CX	
	and/or		
27	Washer	NSA5372-716DX	
	and/or		
27	Washer	NSA5372-716EX	

- The supplementary kit 538096A02S03 includes:

NOTE: The additional kit is valid for operators which have ordered Kit 571140A02R03 and have not accomplished this Service Bulletin yet.

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2	Washer	NSA5372-616AX
	and/or	
2	Washer	NSA5372-616BX
	and/or	
2	Washer	NSA5372-616CX
	and/or	
2	Washer	NSA5372-616DX
	and/or	
2	Washer	NSA5372-616EX
	and	
16	Washer	NSA5372-816AX
	and/or	
16	Washer	NSA5372-816BX
	and/or	
16	Washer	NSA5372-816CX
	and/or	
16	Washer	NSA5372-816DX
	and/or	
16	Washer	NSA5372-816EX

NOTE: Install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCCAA Sheet 01](#).

NOTE: If necessary, the installation of the three washers is approved in this condition.

NOTE: Torque the nuts between 3.80daN and 4.60daN (342lbf.in. and 411lbf.in.).

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1	As required
Fuel Tank Fillet	

Adhesion Promoter- 06PAG1	As required
For Polysulfide Sealant	

(2) Subtask 571140-833-006-002 - Replace the Nuts on the RH Side, ADDITIONAL WORK

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

Manpower Resources	
Manhours	5.00
Minimum number of person	2
Subtask elapsed time	2.50
Skills	AIRFRAME

Material necessary to do the job

Supplementary kit 571140A02S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
10	ASNA2532-7	7	NUT	
11	ASNA2531-7	2	NUT	
13	ASNA2532-8	16	NUT	
15	ASNA2531-6	2	NUT	
	NSA5372-716AX	27	WASHER	
	NSA5372-716BX	27	WASHER	
	NSA5372-716CX	27	WASHER	
	NSA5372-716DX	27	WASHER	
	NSA5372-716EX	27	WASHER	

Supplementary kit 571140A02S03				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
	NSA5372-616AX	2	WASHER	
	NSA5372-616BX	2	WASHER	
	NSA5372-616CX	2	WASHER	
	NSA5372-616DX	2	WASHER	
	NSA5372-616EX	2	WASHER	
	NSA5372-816AX	16	WASHER	
	NSA5372-816BX	16	WASHER	
	NSA5372-816CX	16	WASHER	
	NSA5372-816DX	16	WASHER	
	NSA5372-816EX	16	WASHER	

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11 51-43-00
Fig. A-FCCAA Definition of the Washer Thickness for the Original Taperlok	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCGAB RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

NOTE: The nuts are removed during the inspection.

(a) Install the nuts.

In accordance with SRM 51-42-11 and SRM 51-43-00.

Refer to AMM Task 20-21-11-911-001.

Refer to [Fig. A-FCGAB Sheet 01](#), [Fig. A-FCGAB Sheet 02](#), [Fig. A-FCGAB Sheet 05](#), [Fig. A-FCGAB Sheet 06](#), [Fig. A-FCGAB Sheet 07](#)

- The supplementary kit 538096A02S02 includes:

7	Nut	ASNA2532-7	Item 10
2	Nut	ASNA2531-7	Item 11
16	Nut	ASNA2532-8	Item 13
2	Nut	ASNA2531-6	Item 15
	with		
27	Washer	NSA5372-716AX	
	and/or		
27	Washer	NSA5372-716BX	

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and/or

27 Washer NSA5372-716CX

and/or

27 Washer NSA5372-716DX

and/or

27 Washer NSA5372-716EX

- The supplementary kit 538096A02S03 includes:

NOTE: The additional kit is valid for operators which have ordered Kit 571140A02R03 and have not accomplished this Service Bulletin yet.

2 Washer NSA5372-616AX

and/or

2 Washer NSA5372-616BX

and/or

2 Washer NSA5372-616CX

and/or

2 Washer NSA5372-616DX

and/or

2 Washer NSA5372-616EX

and

16 Washer NSA5372-816AX

and/or

16 Washer NSA5372-816BX

and/or

16 Washer NSA5372-816CX

and/or

16 Washer NSA5372-816DX

and/or

16 Washer NSA5372-816EX

NOTE: Install the washer in accordance with the definition of the washer

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thickness given in the [Fig. A-FCCAA Sheet 01](#).

NOTE: If necessary, the installation of the three washers is approved in this condition.

NOTE: Torque the nuts between 3.80daN and 4.60daN (342lbf.in. and 411lbf.in.).

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(3) Subtask 571140-833-007-002 - Replace the Taperlok with the Next Nominal Diameter Taperlok on the LH Side, **ADDITIONAL WORK**

Work Zones and Access Panels			
	Zone	Access/Work location	
From	141	Access	Panel 147AZ
		Work location	Rib 1
To	540	Access	Panel 540AB, Panel 540BZ
		Work location	Rib 1

Manpower Resources	
Manhours	6.00
Minimum number of person	2
Subtask elapsed time	3.00
Skills	AIRFRAME NON DESTRUCTIVE TESTING

Material necessary to do the job

Supplementary kit 571140A02S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
60	EN6115T2-3	14	BOLT	
61	ASNA2536-2	17	NUT	
62	EN6115T2-2	3	BOLT	
65	NAS1149D0332K	8	WASHER	

Component COMPA04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
11	ASNA2531-7	2	NUT	
34	ABS1418K8-19	4	SCREW	

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ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
35	ASNA2532-8	7	NUT	
37	ABS1418K8-26	1	SCREW	
43	ABS1418K8-25	1	SCREW	
44	ASNA2531-8	2	NUT	
73	ABS1418K9-15	2	SCREW	
74	ABS1418K9-16	8	SCREW	
75	ABS1418K9-17	6	SCREW	
76	ASNA2532-9	16	NUT	
78	ABS1418K8-20	2	SCREW	
80	ABS1418K7-23	2	SCREW	
82	ABS1418K8-22	1	SCREW	
	NSA5372-716AX	2	WASHER	
	NSA5372-716BX	2	WASHER	
	NSA5372-716EX	2	WASHER	
	NSA5372-816AX	9	WASHER	
	NSA5372-816BX	9	WASHER	
	NSA5372-816EX	9	WASHER	
	NSA5372-916AX	16	WASHER	
	NSA5372-916BX	16	WASHER	
	NSA5372-916EX	16	WASHER	

NOTE: The quantities given in the component table are for all the taperloks replacement.

NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Wash Primer - Structure	04CMA2	As required	
	Primer Polyurethane Paint - Corrosion Inhibiting	04EAC2	As required	
	Top Coat Polyurethane - External Structure	04JAA3	As required	
	Polysulfide Sealant-General Purpose Brushable	06AAA1	As required	
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

Kit tool 571140T02R00				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103002000	Test	1	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Drawing	R57102001
Non Destructive Test Manual (NTM)	51-10-01 PART 6
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00
Fig. A-FCDA Definition of the Washer Thickness for the New Taperlok	Sheet 01
Fig. A-FDAAA Removal/Installation of the Cooling Duct	Sheet 01
Fig. A-FDBAA Removal/Installation of the Belly Fairing Structure	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCFAB LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

In accordance with SRM 51-42-11, SRM 51-43-00 and SRM 51-24-00.

Refer to AMM Task 20-21-11-911-001.

Refer to NTM 51-10-01 PART 6

Refer to [Fig. A-FCFAB Sheet 01](#) to [Fig. A-FCFAB Sheet 07](#)

(a) Depending on the result of the inspection, remove for access :

1 Remove the cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (53)	Retain
1	Duct	Item (52)	Retain
	attached with		
1	Clamp	Item (63)	Retain
8	Screw	Item (64)	Retain

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8	Washer	Item (65)	Discard
---	--------	-----------	---------

2 Remove the structure of the belly fairing.

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (51)	Retain
---	----------------	-----------	--------

attached with

3	Rivet	Item (62)	Discard
---	-------	-----------	---------

7	Rivet	Item (60)	Discard
---	-------	-----------	---------

10	Bush	Item (61)	Discard
----	------	-----------	---------

and/or

1	Web 4 assy	Item (50)	Retain
---	------------	-----------	--------

attached with

7	Rivet	Item (60)	Discard
---	-------	-----------	---------

7	Bush	Item (61)	Discard
---	------	-----------	---------

(b) In accordance with the result of the inspection, remove the taperlok(s) :

Refer to [Fig. A-FCFAB Sheet 01](#) to [Fig. A-FCFAB Sheet 04](#)

4	Screw	Item (7)	Discard
---	-------	----------	---------

1	Screw	Item (14)	Discard
---	-------	-----------	---------

1	Screw	Item (12)	Discard
---	-------	-----------	---------

2	Screw	Item (70)	Discard
---	-------	-----------	---------

8	Screw	Item (71)	Discard
---	-------	-----------	---------

6	Screw	Item (72)	Discard
---	-------	-----------	---------

2	Screw	Item (77)	Discard
---	-------	-----------	---------

2	Screw	Item (79)	Discard
---	-------	-----------	---------

1	Screw	Item (81)	Discard
---	-------	-----------	---------

NOTE: The nuts are removed during the inspection.

- (c) Do a rototest inspection of the holes for cracks, refer to NTM 51-10-01 PART 6.

1 If cracks are found, contact AIRBUS before any further work.

2 If no crack is found, do the following steps.

- (d) Oversize the hole to the next nominal diameter in accordance with the drilling procedure given in the Repair Instruction Drawing R57102001 and use the drilling tool SOA-TL-R57102001B.

NOTE: To prevent any accidental damage, we recommend to train to oversize the taperlok holes on the test specimen.

98D57103002000 Test specimen 1

- (e) Depending on the removal, install :

Refer to [Fig. A-FCFAB Sheet 01](#), [Fig. A-FCFAB Sheet 02](#), [Fig. A-FCFAB Sheet 05](#), [Fig. A-FCFAB Sheet 06](#), [Fig. A-FCFAB Sheet 07](#)

4	Screw	ABS1418K8-19	Item 34
2	Screw	ABS1418K8-20	Item 78
1	Screw	ABS1418K8-22	Item 82
	with		
7	Nut	ASNA2532-8	Item 35
	and if necessary		
7	Washer	NSA5372-816AX	
	and/or		
7	Washer	NSA5372-816EX	
	and/or		
7	Washer	NSA5372-816BX	
	and		
1	Screw	ABS1418K8-25	Item 43
1	Screw	ABS1418K8-26	Item 37
	with		
2	Nut	ASNA2531-8	Item 44
	and if necessary		
2	Washer	NSA5372-816AX	

	and/or		
2	Washer	NSA5372-816EX	
	and/or		
2	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K9-15	Item 73
8	Screw	ABS1418K9-16	Item 74
6	Screw	ABS1418K9-17	Item 75
	with		
16	Nut	ASNA2532-9	Item 76
	and if necessary		
16	Washer	NSA5372-916AX	
	and/or		
16	Washer	NSA5372-916EX	
	and/or		
16	Washer	NSA5372-916BX	
	and		
2	Screw	ABS1418K7-23	Item 80
	with		
2	Nut	ASNA2531-7	Item 11
	and if necessary		
2	Washer	NSA5372-716AX	
	and/or		
2	Washer	NSA5372-716EX	
	and/or		
2	Washer	NSA5372-716BX	

NOTE: Install the screw in accordance with the installation principle given in the repair instruction R57102001.

NOTE: If necessary, install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCDA Sheet 01](#).

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NOTE: Wet assembly, apply :

Polysulfide Sealant- 06ABB1	As required
Fuel Tank Fillet	

NOTE: Torque the nuts between 5.60daN and 6.70daN (493lbf.in. and 592lbf.in.).

NOTE: Outside the wing center box, apply on the screws :

Primer	04EAC2	As required
Polyurethane Paint		
- Corrosion		
Inhibiting		

Wash Primer -	04CMA2	As required
Structure		

Top Coat	04JAA3	As required
Polyurethane -		
External Structure		

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1	As required
Fuel Tank Fillet	

Adhesion Promoter- 06PAG1	As required
For Polysulfide	
Sealant	

(f) If removed, install :

1 The structure of the belly fairing

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed		Item (51)	Retained at removal
	with			
7	Bolt	EN6115T2-3	Item 60	
3	Bolt	EN6115T2-2	Item 62	
10	Nut	ASNA2536-2	Item 61	
	and/or			
1	Web 4 assy		Item (50)	Retained at removal
	attached with			

SERVICE BULLETIN

7	Bolt	EN6115T2-3	Item 60
7	Nut	ASNA2536-2	Item 61

NOTE: Apply to the interface of the structure and all the parts :

Polysulfide Sealant- 06AAA1 As required
General Purpose
Brushable

2 The cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (53)	Retained at removal
1	Duct	Item (52)	Retained at removal
	with		
1	Clamp	Item (63)	Retained at removal
8	Screw	Item (64)	Retained at removal
8	Washer	NAS1149D0332K	Item 65

(4) Subtask 571140-833-008-002 - Replace the Taperlok with the Next Nominal Diameter Taperlok on the RH Side, ADDITIONAL WORK

Work Zones and Access Panels			
	Zone	Access/Work location	
From	142	Access	Panel 148AZ
		Work location	Rib 1
To	640	Access	Panel 640AB, Panel 640BZ
		Work location	Rib 1

SERVICE BULLETIN

Manpower Resources	
Manhours	6.00
Minimum number of person	2
Subtask elapsed time	3.00
Skills	AIRFRAME NON DESTRUCTIVE TESTING

Material necessary to do the job

Supplementary kit 571140A02S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
60	EN6115T2-3	14	BOLT	
61	ASNA2536-2	17	NUT	
62	EN6115T2-2	3	BOLT	
65	NAS1149D0332K	8	WASHER	

Component COMPA04				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
11	ASNA2531-7	2	NUT	
34	ABS1418K8-19	4	SCREW	
35	ASNA2532-8	7	NUT	
37	ABS1418K8-26	1	SCREW	
43	ABS1418K8-25	1	SCREW	
44	ASNA2531-8	2	NUT	
73	ABS1418K9-15	2	SCREW	
74	ABS1418K9-16	8	SCREW	
75	ABS1418K9-17	6	SCREW	
76	ASNA2532-9	16	NUT	
78	ABS1418K8-20	2	SCREW	
80	ABS1418K7-23	2	SCREW	
82	ABS1418K8-22	1	SCREW	
	NSA5372-716AX	2	WASHER	
	NSA5372-716BX	2	WASHER	
	NSA5372-716EX	2	WASHER	
	NSA5372-816AX	9	WASHER	
	NSA5372-816BX	9	WASHER	
	NSA5372-816EX	9	WASHER	
	NSA5372-916AX	16	WASHER	
	NSA5372-916BX	16	WASHER	
	NSA5372-916EX	16	WASHER	

NOTE: The quantities given in the component table are for all the taperlocks replacement.

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NOTE: The above list of components is not an AIRBUS Kit, the required parts shall be ordered as necessary through the given channel.

NOTE: The components listed above are only necessary if Repair Task. The quantity of these items are to be ordered as required.

Consumable CMLA01				
ITEM	DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	SEE NOTES
	Wash Primer - Structure	04CMA2	As required	
	Primer Polyurethane Paint - Corrosion Inhibiting	04EAC2	As required	
	Top Coat Polyurethane - External Structure	04JAA3	As required	
	Polysulfide Sealant-General Purpose Brushable	06AAA1	As required	
	Polysulfide Sealant-Fuel Tank Fillet	06ABB1	As required	
	Adhesion Promoter-For Polysulfide Sealant	06PAG1	As required	

Kit tool 571140T02R00				
ITEM	PART N°	KEYWORD	QTY	SEE NOTES
	98D57103002000	Test	1	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Drawing	R57102001
Non Destructive Test Manual (NTM)	51-10-01 PART 6
Structural Repair Manual (SRM)	51-24-00 51-42-11 51-43-00
Fig. A-FCDA Definition of the Washer Thickness for the New Taperlok	Sheet 01
Fig. A-FDAAA Removal/Installation of the Cooling Duct	Sheet 01
Fig. A-FDBAA Removal/Installation of the Belly Fairing Structure	Sheet 01 Sheet 02 Sheet 03
Fig. A-FCGAB RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK	Sheet 01 Sheet 02 Sheet 03 Sheet 04 Sheet 05 Sheet 06 Sheet 07

In accordance with SRM 51-42-11, SRM 51-43-00 and SRM 51-24-00.

Refer to AMM Task 20-21-11-911-001.

Refer to NTM 51-10-01 PART 6

Refer to [Fig. A-FCGAB Sheet 01](#) to [Fig. A-FCGAB Sheet 07](#)

(a) Depending on the result of the inspection, remove for access :

1 Remove the cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct	Item (54)	Retain
1	Duct	Item (52)	Retain
	attached with		
1	Clamp	Item (63)	Retain
8	Screw	Item (64)	Retain
8	Washer	Item (65)	Discard

2 Remove the structure of the belly fairing.

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed	Item (171)	Retain
	attached with		
3	Rivet	Item (62)	Discard
7	Rivet	Item (60)	Discard
10	Bush	Item (61)	Discard
	and/or		
1	Web 4 assy	Item (170)	Retain
	attached with		
7	Rivet	Item (60)	Discard
7	Bush	Item (61)	Discard

(b) In accordance with the result of the inspection, remove the taperlok(s) :

Refer to [Fig. A-FCGAB Sheet 01](#) to [Fig. A-FCGAB Sheet 04](#)

4	Screw	Item (7)	Discard
1	Screw	Item (14)	Discard
1	Screw	Item (12)	Discard
2	Screw	Item (70)	Discard
8	Screw	Item (71)	Discard
6	Screw	Item (72)	Discard
2	Screw	Item (77)	Discard
2	Screw	Item (79)	Discard
1	Screw	Item (81)	Discard

NOTE: The nuts are removed during the inspection.

- (c) Do a rototest inspection of the holes for cracks refer to NTM 51-10-01 PART 6.

1 If cracks are found, contact AIRBUS before any further work.

2 If no crack is found, do the following steps.

- (d) Oversize the hole to the next nominal diameter in accordance with the drilling procedure given in the Repair Instruction Drawing R57102001 and use the drilling tool SOA-TL-R57102001B.

NOTE: To prevent any accidental damage, we recommend to train to oversize the taperlok holes on the test specimen.

98D57103002000 Test specimen 1

- (e) Depending on the removal, install :

Refer to [Fig. A-FCGAB Sheet 01](#), [Fig. A-FCGAB Sheet 02](#), [Fig. A-FCGAB Sheet 05](#), [Fig. A-FCGAB Sheet 06](#), [Fig. A-FCGAB Sheet 07](#)

4	Screw	ABS1418K8-19	Item 34
2	Screw	ABS1418K8-20	Item 78
1	Screw	ABS1418K8-22	Item 82
	with		
7	Nut	ASNA2532-8	Item 35

and if necessary

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7	Washer	NSA5372-816AX	
	and/or		
7	Washer	NSA5372-816EX	
	and/or		
7	Washer	NSA5372-816BX	
	and		
1	Screw	ABS1418K8-25	Item 43
1	Screw	ABS1418K8-26	Item 37
	with		
2	Nut	ASNA2531-8	Item 44
	and if necessary		
2	Washer	NSA5372-816AX	
	and/or		
2	Washer	NSA5372-816EX	
	and/or		
2	Washer	NSA5372-816BX	
	and		
2	Screw	ABS1418K9-15	Item 73
8	Screw	ABS1418K9-16	Item 74
6	Screw	ABS1418K9-17	Item 75
	with		
16	Nut	ASNA2532-9	Item 76
	and if necessary		
16	Washer	NSA5372-916AX	
	and/or		
16	Washer	NSA5372-916EX	
	and/or		
16	Washer	NSA5372-916BX	
	and		

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2	Screw	ABS1418K7-23	Item 80
	with		
2	Nut	ASNA2531-7	Item 11
	and if necessary		
2	Washer	NSA5372-716AX	
	and/or		
2	Washer	NSA5372-716EX	
	and/or		
2	Washer	NSA5372-716BX	

NOTE: Install the screw in accordance with the installation principle given in the repair instruction R57102001.

NOTE: If necessary, install the washer in accordance with the definition of the washer thickness given in the [Fig. A-FCDA Sheet 01](#).

NOTE: Wet assembly, apply :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

NOTE: Torque the nuts between 5.60daN and 6.70daN (493lbf.in. and 592lbf.in.).

NOTE: Outside the wing center box, apply on the screws :

Primer 04EAC2 As required
Polyurethane Paint
- Corrosion
Inhibiting

Wash Primer - 04CMA2 As required
Structure

Top Coat 04JAA3 As required
Polyurethane -
External Structure

NOTE: Inside the wing center box, apply on the nuts :

Polysulfide Sealant- 06ABB1 As required
Fuel Tank Fillet

Adhesion Promoter- 06PAG1 As required
For Polysulfide
Sealant

(f) If removed, install :

1 The structure of the belly fairing

Refer to [Fig. A-FDBAA Sheet 01](#) to [Fig. A-FDBAA Sheet 03](#)

1	Channel-Formed		Item (171)	Retained at removal
	with			
7	Bolt	EN6115T2-3	Item 60	
3	Bolt	EN6115T2-2	Item 62	
10	Nut	ASNA2536-2	Item 61	
	and/or			
1	Web 4 assy		Item (170)	Retained at removal
	attached with			
7	Bolt	EN6115T2-3	Item 60	
7	Nut	ASNA2536-2	Item 61	

NOTE: Apply to the interface of the structure and all the parts :

Polysulfide Sealant- 06AAA1 As required
General Purpose
Brushable

2 The cooling duct.

Refer to [Fig. A-FDAAA Sheet 01](#)

1	Duct		Item (54)	Retained at removal
1	Duct		Item (52)	Retained at removal
	with			
1	Clamp		Item (63)	Retained at removal
8	Screw		Item (64)	Retained at removal
8	Washer	NAS1149D0332K	Item 65	

****CONF ALL****D. TEST******CONF 001****(1) Subtask 571140-710-002-001 - Test**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 20-28-00-720-005 Task 20-28-00-869-002 Task 28-11-00-280-002 Task 28-21-00-710-006 Task 57-17-11-400-001 Task 57-27-11-400-001

NOTE: This test procedure is to be accomplished after the installation of the parts removed for the access SUBTASK 571137-410-003 001 Install/Close Items Removed/Opened for Access on the LH Side, SUBTASK 571137-410-004 001 Install/Close Items Removed/Opened for Access on the RH Side, SUBTASK 571137-410-007 001 Install the Fuel Strainer, LH Side and SUBTASK 571137-410-008 001 Install the Fuel Strainer, RH Side.

(a) Job Set-up :

- 1 Remove the safety clips and tags and close the circuit breakers as specified after the installation of the access covers 147AZ and 148AZ, refer to AMM Task 57-17-11-400-001.
- 2 Remove the safety clips and tags and close the circuit breakers as specified after the installation of the access panels 540AB and 640AB, refer to AMM Task 57-27-11-400-001.

(b) Test :

- 1 Do the test procedure as specified after the installation of the access covers 147AZ and 148AZ, refer to AMM Task 57-17-11-400-001.
- 2 Do the test procedure as specified after the installation of the access panels 540AB and 640AB, refer to AMM Task 57-27-11-400-001.
- 3 Do a fuel tanks leak checks (only leak test of the center tank or/and leak test of the wing tanks - without fuel - by the local blowing procedure), refer to AMM Task 28-11-00-280-002.

NOTE: Do this test on the taperlok bolts inspected by this Service Bulletin.

- 4 For all the access panels and covers removed during the preparation procedure, do a check of the electrical bonding (external metal surfaces/parts (hinges or fixed)) refer to AMM Task 20-28-00-720-005.

NOTE: For the maximum permitted resistance values : refer to the table AMM Task 20-28-00-869-002.

****CONF 002****(1) Subtask 571140-710-002-001 - Test**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 20-28-00-720-005
	Task 20-28-00-869-002
	Task 28-11-00-280-002
	Task 28-21-00-710-006
	Task 57-17-11-400-001
	Task 57-27-11-400-001

NOTE: This test procedure is to be accomplished after the installation of the parts removed for the access SUBTASK 571137-410-003 001 Install/Close Items Removed/Opened for Access on the LH Side, SUBTASK 571137-410-004 001 Install/Close Items Removed/Opened for Access on the RH Side, SUBTASK 571137-410-007 001 Install the Fuel Strainer, LH Side and SUBTASK 571137-410-008 001 Install the Fuel Strainer, RH Side.

(a) Job Set-up :

- 1 Remove the safety clips and tags and close the circuit breakers as specified after the installation of the access covers 147AZ and 148AZ, refer to AMM Task 57-17-11-400-001.
- 2 Remove the safety clips and tags and close the circuit breakers as specified after the installation of the access panels 540AB and 640AB, refer to AMM Task 57-27-11-400-001.

(b) Test :

- 1 Do the test procedure as specified after the installation of the access covers 147AZ and 148AZ, refer to AMM Task 57-17-11-400-001.
- 2 Do the test procedure as specified after the installation of the access panels 540AB and 640AB, refer to AMM Task 57-27-11-400-001.
- 3 Do a fuel tanks leak checks (only leak test of the center tank or/and leak test of the wing tanks - without fuel - by the local blowing procedure), refer to AMM Task 28-11-00-280-002.

NOTE: Do this test on the taperlok bolts inspected by this Service

Bulletin.

- 4 For all the access panels and covers removed during the preparation procedure, do a check of the electrical bonding (external metal surfaces/parts (hinges or fixed)) refer to AMM Task 20-28-00-720-005.

NOTE: For the maximum permitted resistance values : refer to the table AMM Task 20-28-00-869-002.

****CONF ALL**

E. CLOSE-UP

****CONF 001**

- (1) **Subtask 571140-410-003-001 - Install/Close Items Removed/Opened for Access on the LH Side**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-400-001
	Task 28-21-54-400-001
	Task 53-35-11-400-001
	Task 57-17-11-400-001
	Task 57-27-11-400-001
	Task 57-27-12-400-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-002 001 Testing.

- (a) Make sure that the work areas are clean and clear of tools and other items of equipment.
- (b) Install the covers Item (80) and (81) of the fuel pump strainer FIN 6QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-400-001.
- (c) Install the fuel pump suction valve, refer to AMM Task 28-21-42-400-001.
- (d) Close the fuel surge access door 540BZ, refer to AMM Task 57-27-12-400-001.
- (e) Install the wing access panel 540AB, refer to AMM Task 57-27-11-400-001.
- (f) Install the access cover 147AZ, refer to AMM Task 57-17-11-400-001.
- (g) If removed, install the panels 191LB and 147CB on the wing to fuselage, refer to AMM Task 53-35-11-400-001.

(2) Subtask 571140-410-004-001 - Install/Close Items Removed/Opened for Access on the RH Side

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-400-001 Task 28-21-54-400-001 Task 53-35-11-400-001 Task 57-17-11-400-001 Task 57-27-11-400-001 Task 57-27-12-400-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-002 001 Testing.

- (a) Make sure that the work areas are clean and clear of tools and other items of equipment.
- (b) Install the covers Item (80) and (81) of the fuel pump strainer FIN 8QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-400-001.
- (c) Install the fuel pump suction valve, refer to AMM Task 28-21-42-400-001.
- (d) Close the fuel surge access door 640BZ, refer to AMM Task 57-27-12-400-001.
- (e) Install the wing access panel 640AB, refer to AMM Task 57-27-11-400-001.
- (f) Install the access cover 148AZ, refer to AMM Task 57-17-11-400-001.
- (g) If removed, install the panels , 192LB and 148DB on the wing to fuselage, refer to AMM Task 53-35-11-400-001.

(3) Subtask 571140-410-007-001 - Install the Fuel Retainer, LH Side

Manpower Resources				
Skills			NON SPECIFIC	
Material necessary to do the job				
Supplementary kit 571140A01S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
69	NAS1149F0332P	2	WASHER	
References				
Aircraft Maintenance Manual (AMM)		Task 20-21-11-911-001		

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-002 001 Testing.

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

(a) If removed, install the fuel retainer for LH side:

Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)

In accordance with SRM 51-42-11.

Refer to AMM Task 20-21-11-911-001.

1	Strainer Assembly	Item (55)	Retained at removal
1	Strainer Assembly	Item (57)	Retained at removal
	with		
8	Bolt	Item (66)	Retained at removal
2	Bolt	Item (67)	Retained at removal
2	Bolt	Item (68)	Retained at removal
2	Washer	NAS1149F0332P	Item 69

(4) Subtask 571140-410-008-001 - Install the Fuel Retainer, RH Side

Manpower Resources	
Skills	NON SPECIFIC

Material necessary to do the job

Supplementary kit 571140A01S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
69	NAS1149F0332P	2	WASHER	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001

References	
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA	Sheet 01
Removal/Installation of the Fuel Strainer	Sheet 02

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-002 001 Testing.

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

(a) If removed, install the fuel retainer for RH side:

Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)

In accordance with SRM 51-42-11.

Refer to AMM Task 20-21-11-911-001.

1	Strainer Assembly	Item (56)	Retained at removal
1	Strainer Assembly	Item (58)	Retained at removal
	with		
8	Bolt	Item (66)	Retained at removal
2	Bolt	Item (67)	Retained at removal
2	Bolt	Item (68)	Retained at removal
2	Washer	NAS1149F0332P	Item 69

(5) Subtask 571140-942-002-001 - Close-up

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-25-00-650-001 Task 32-12-00-410-001

(a) Refuel the LH and RH wing tanks, refer to AMM Task 28-25-00-650-001.

(b) Refuel the center tank, refer to AMM Task 28-25-00-650-001.

(c) Close the main landing gear doors, refer to AMM Task 32-12-00-410-001.

- (d) Remove the access platform(s).
- (e) Put the aircraft back to its initial configuration.

****CONF 002****(1) Subtask 571140-410-003-001 - Install/Close Items Removed/Opened for Access on the LH Side**

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-400-001 Task 28-21-54-400-001 Task 53-35-11-400-001 Task 57-17-11-400-001 Task 57-27-11-400-001 Task 57-27-12-400-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-002 001 Testing.

- (a) Make sure that the work areas are clean and clear of tools and other items of equipment.
- (b) Install the covers Item (80) and (81) of the fuel pump strainer FIN 6QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-400-001.
- (c) Install the fuel pump suction valve, refer to AMM Task 28-21-42-400-001.
- (d) Close the fuel surge access door 540BZ, refer to AMM Task 57-27-12-400-001.
- (e) Install the wing access panel 540AB, refer to AMM Task 57-27-11-400-001.
- (f) Install the access cover 147AZ, refer to AMM Task 57-17-11-400-001.
- (g) If removed, install the panels 191LB and 147CB on the wing to fuselage, refer to AMM Task 53-35-11-400-001.

(2) Subtask 571140-410-004-001 - Install/Close Items Removed/Opened for Access on the RH Side

Manpower Resources	
Skills	NON SPECIFIC

SERVICE BULLETIN

References	
Aircraft Maintenance Manual (AMM)	Task 28-21-42-400-001 Task 28-21-54-400-001 Task 53-35-11-400-001 Task 57-17-11-400-001 Task 57-27-11-400-001 Task 57-27-12-400-001
Fig. A-FDCAA Removal/Installation of the Fuel Pump Strainer Covers	Sheet 01

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-002 001 Testing.

- (a) Make sure that the work areas are clean and clear of tools and other items of equipment.
- (b) Install the covers Item (80) and (81) of the fuel pump strainer FIN 8QM, refer to [Fig. A-FDCAA Sheet 01](#) and AMM Task 28-21-54-400-001.
- (c) Install the fuel pump suction valve, refer to AMM Task 28-21-42-400-001.
- (d) Close the fuel surge access door 640BZ, refer to AMM Task 57-27-12-400-001.
- (e) Install the wing access panel 640AB, refer to AMM Task 57-27-11-400-001.
- (f) Install the access cover 148AZ, refer to AMM Task 57-17-11-400-001.
- (g) If removed, install the panels , 192LB and 148DB on the wing to fuselage, refer to AMM Task 53-35-11-400-001.

(3) Subtask 571140-410-007-002 - Install the Fuel Retainer, LH Side

Manpower Resources	
Skills	NON SPECIFIC

Material necessary to do the job

Supplementary kit 571140A02S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
69	NAS1149F0332P	2	WASHER	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-002 001 Testing.

SERVICE BULLETIN

NOTE: This step is only valid for aircraft on which Service Bulletin No. A320-28-1149 (or Modification No. 36387J2487) is accomplished.

(a) If removed, install the fuel retainer for LH side:

Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)

In accordance with SRM 51-42-11.

Refer to AMM Task 20-21-11-911-001.

1	Strainer Assembly	Item (55)	Retained at removal
1	Strainer Assembly	Item (57)	Retained at removal
	with		
8	Bolt	Item (66)	Retained at removal
2	Bolt	Item (67)	Retained at removal
2	Bolt	Item (68)	Retained at removal
2	Washer	NAS1149F0332P Item 69	

(4) Subtask 571140-410-008-002 - Install the Fuel Retainer, RH Side

Manpower Resources	
Skills	NON SPECIFIC

Material necessary to do the job

Supplementary kit 571140A02S02				
ITEM	NEW PART N°	QTY (UM)	KEYWORD	SEE NOTES
69	NAS1149F0332P	2	WASHER	

References	
Aircraft Maintenance Manual (AMM)	Task 20-21-11-911-001
Structural Repair Manual (SRM)	51-42-11
Fig. A-FDDAA Removal/Installation of the Fuel Strainer	Sheet 01 Sheet 02

NOTE: The parts removed for the access must be installed before the SUBTASK 571137-700-002 001 Testing.

NOTE: This step is only valid for aircraft on which Service Bulletin No.

A320-28-1149 (or Modification No. 36387J2487) is accomplished.

- (a) If removed, install the fuel retainer for RH side:

Refer to [Fig. A-FDDAA Sheet 01](#) and [Fig. A-FDDAA Sheet 02](#)

In accordance with SRM 51-42-11.

Refer to AMM Task 20-21-11-911-001.

1	Strainer Assembly	Item (56)	Retained at removal
1	Strainer Assembly	Item (58)	Retained at removal
	with		
8	Bolt	Item (66)	Retained at removal
2	Bolt	Item (67)	Retained at removal
2	Bolt	Item (68)	Retained at removal
2	Washer	NAS1149F0332P	Item 69

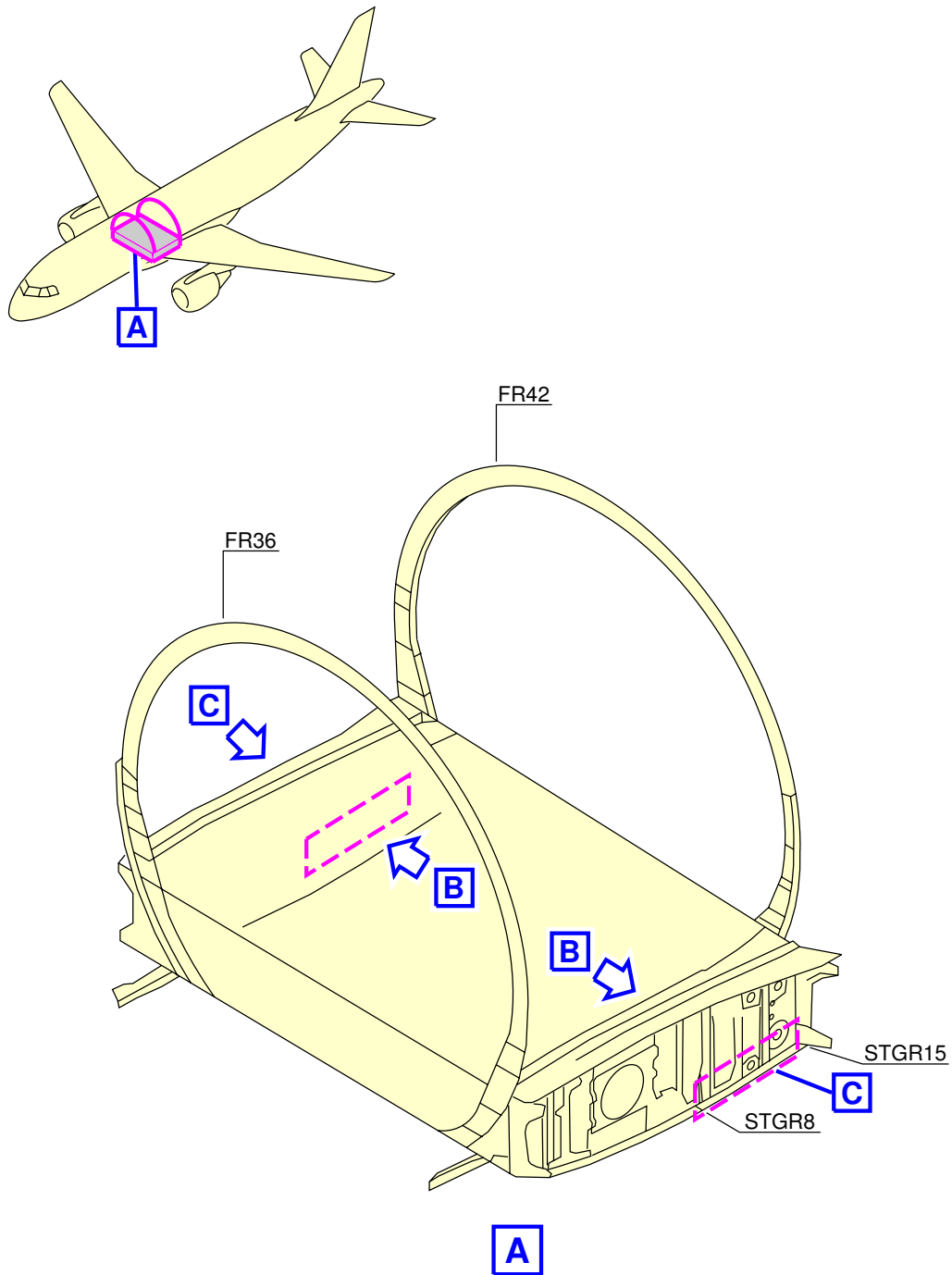
(5) Subtask 571140-942-002-001 - Close-up

Manpower Resources	
Skills	NON SPECIFIC

References	
Aircraft Maintenance Manual (AMM)	Task 28-25-00-650-001 Task 32-12-00-410-001

- (a) Refuel the LH and RH wing tanks, refer to AMM Task 28-25-00-650-001.
- (b) Refuel the center tank, refer to AMM Task 28-25-00-650-001.
- (c) Close the main landing gear doors, refer to AMM Task 32-12-00-410-001.
- (d) Remove the access platform(s).
- (e) Put the aircraft back to its initial configuration.

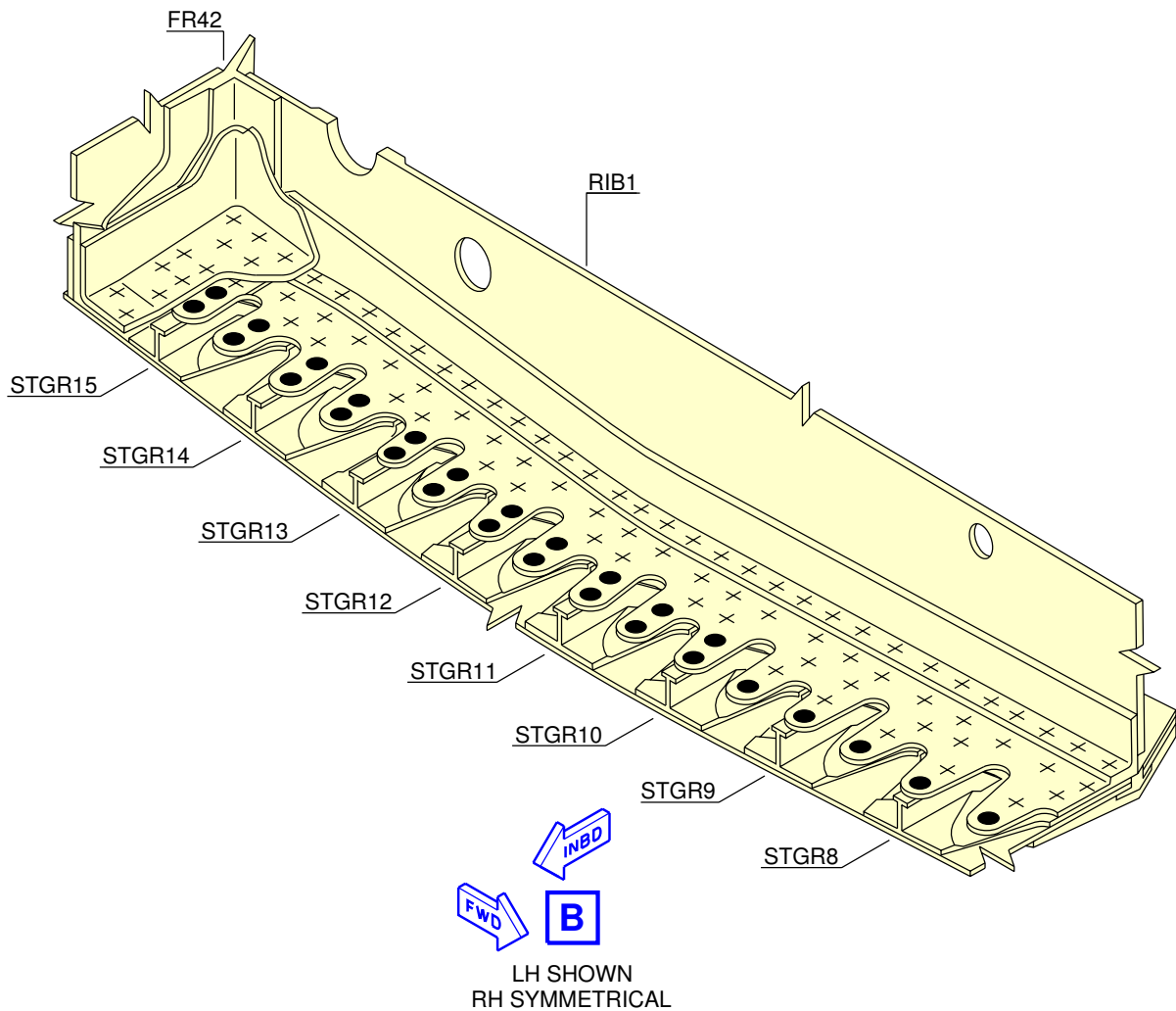
****CONF ALL**



N_SB_571140_5_BAAA_01_00

Figure A-FBAAA - Sheet 01
Inspection of the Fasteners

****CONF ALL**



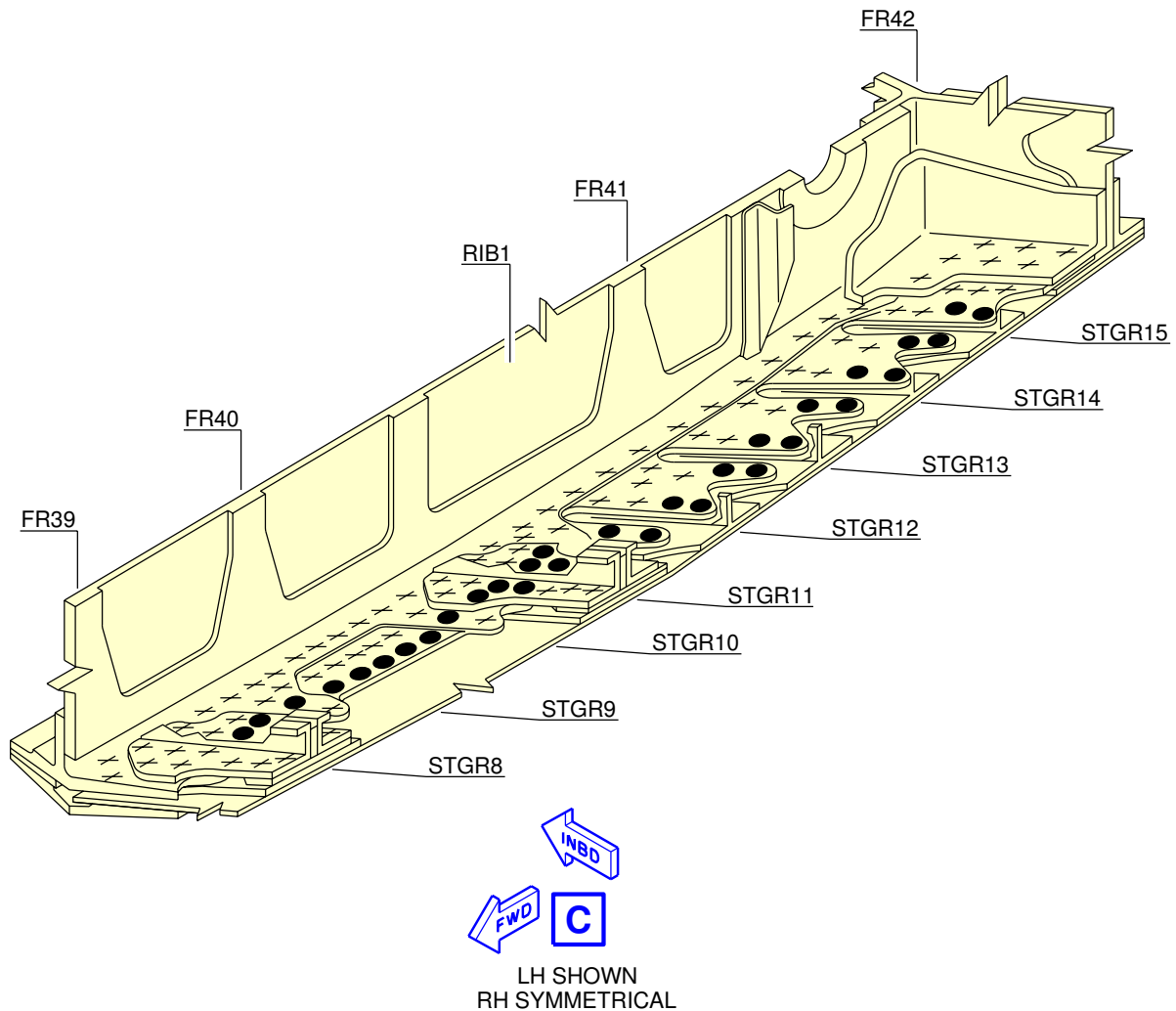
NOTE:

- + FASTENER NOT AFFECTED.
- FASTENER TO BE REMOVED.



N_SB_571140_5_BAAA_02_01

Figure A-FBAAA - Sheet 02
Inspection of the Fasteners

****CONF ALL**



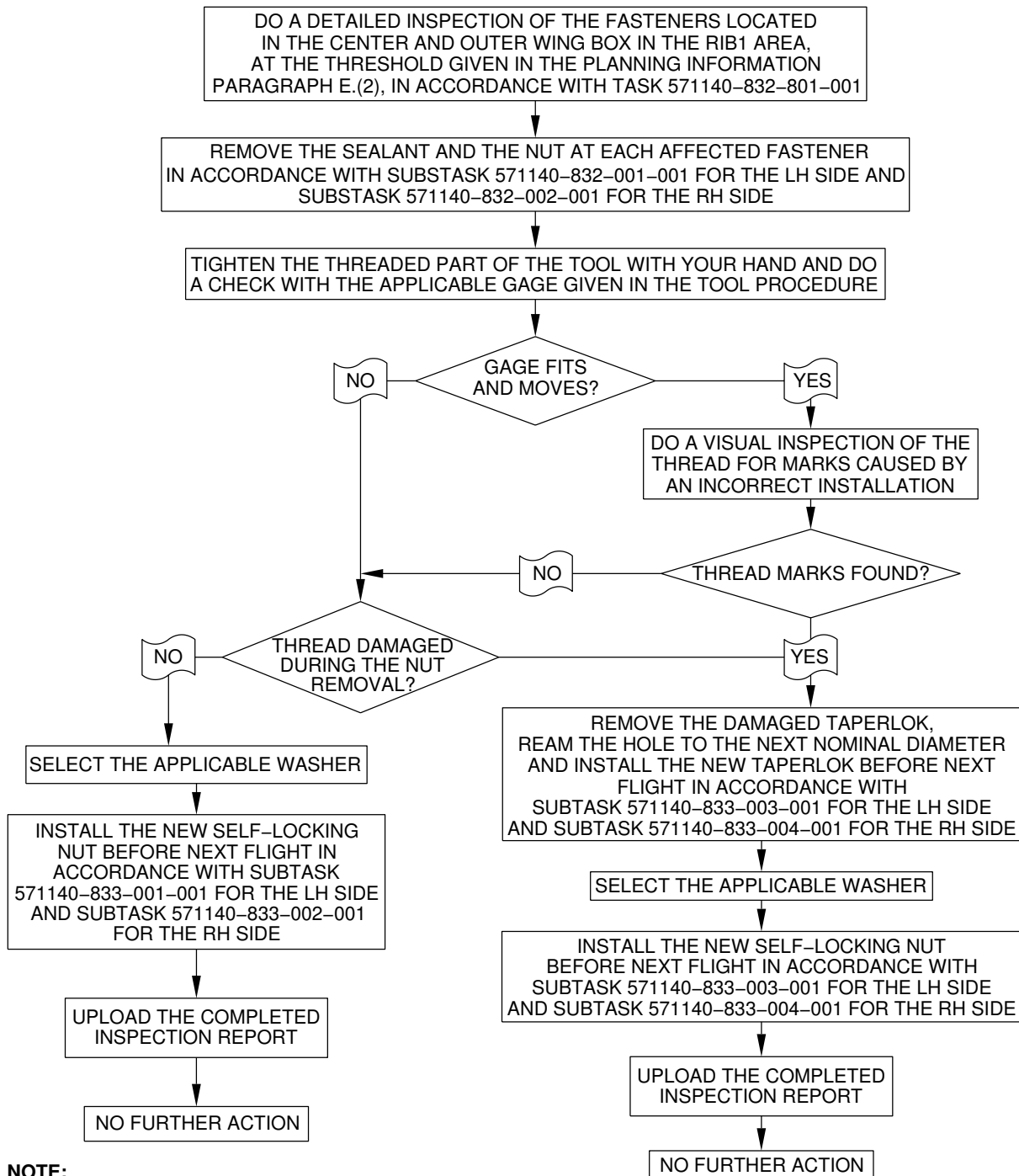
NOTE:

-  FASTENER NOT AFFECTED.
-  FASTENER TO BE REMOVED.

N_SB_571140_5_BAAA_03_01

Figure A-FBAAA - Sheet 03
Inspection of the Fasteners

****CONF 001**



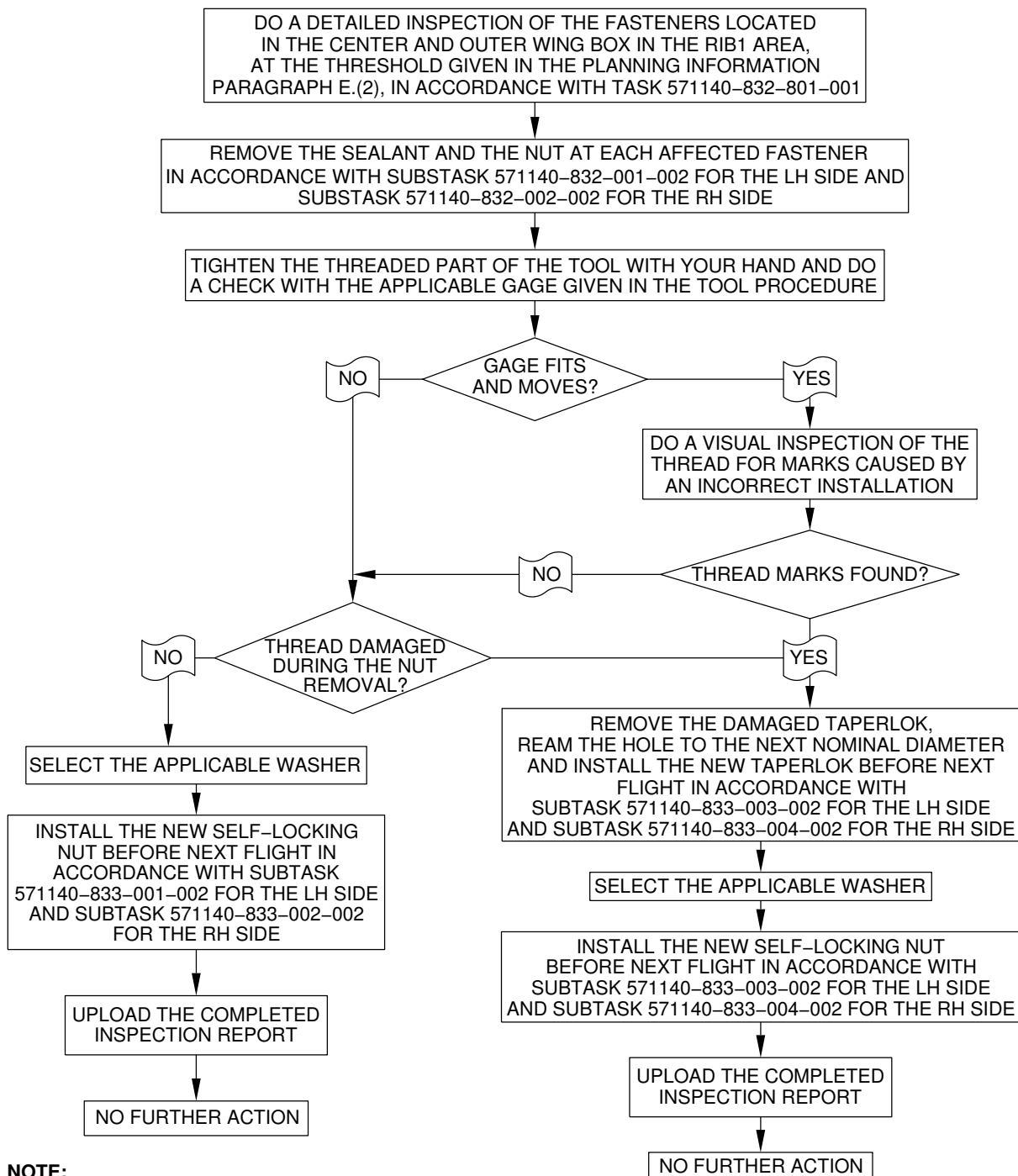
NOTE:

THE PURPOSE OF FLOWCHARTS IS TO SUPPLEMENT THE INFORMATION GIVEN IN THE PROCEDURE AND COMPLIANCE PARAGRAPHS AND NOT TO SERVE AS THE PRIMARY SOURCE FOR TASKS OR COMPLIANCE TIMES GIVEN IN THIS SERVICE BULLETIN.

N_SB_571140_5_FAAA_01_04

Figure A-FFAAA - Sheet 01
Flow Chart

****CONF 002**



NOTE:

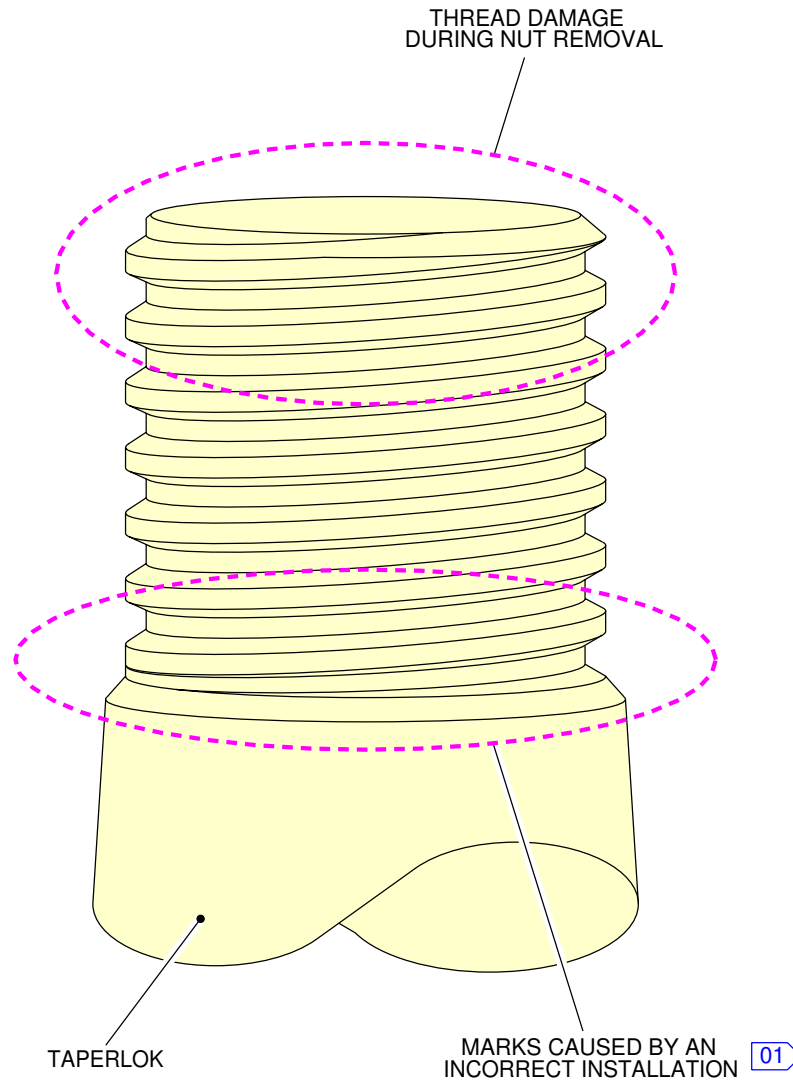
THE PURPOSE OF FLOWCHARTS IS TO SUPPLEMENT THE INFORMATION GIVEN IN THE PROCEDURE AND COMPLIANCE PARAGRAPHS AND NOT TO SERVE AS THE PRIMARY SOURCE FOR TASKS OR COMPLIANCE TIMES GIVEN IN THIS SERVICE BULLETIN.

N_SB_571140_5_FAAB_01_02

Figure A-FFAAB - Sheet 01
Flow Chart

****CONF ALL**

THREAD DAMAGE

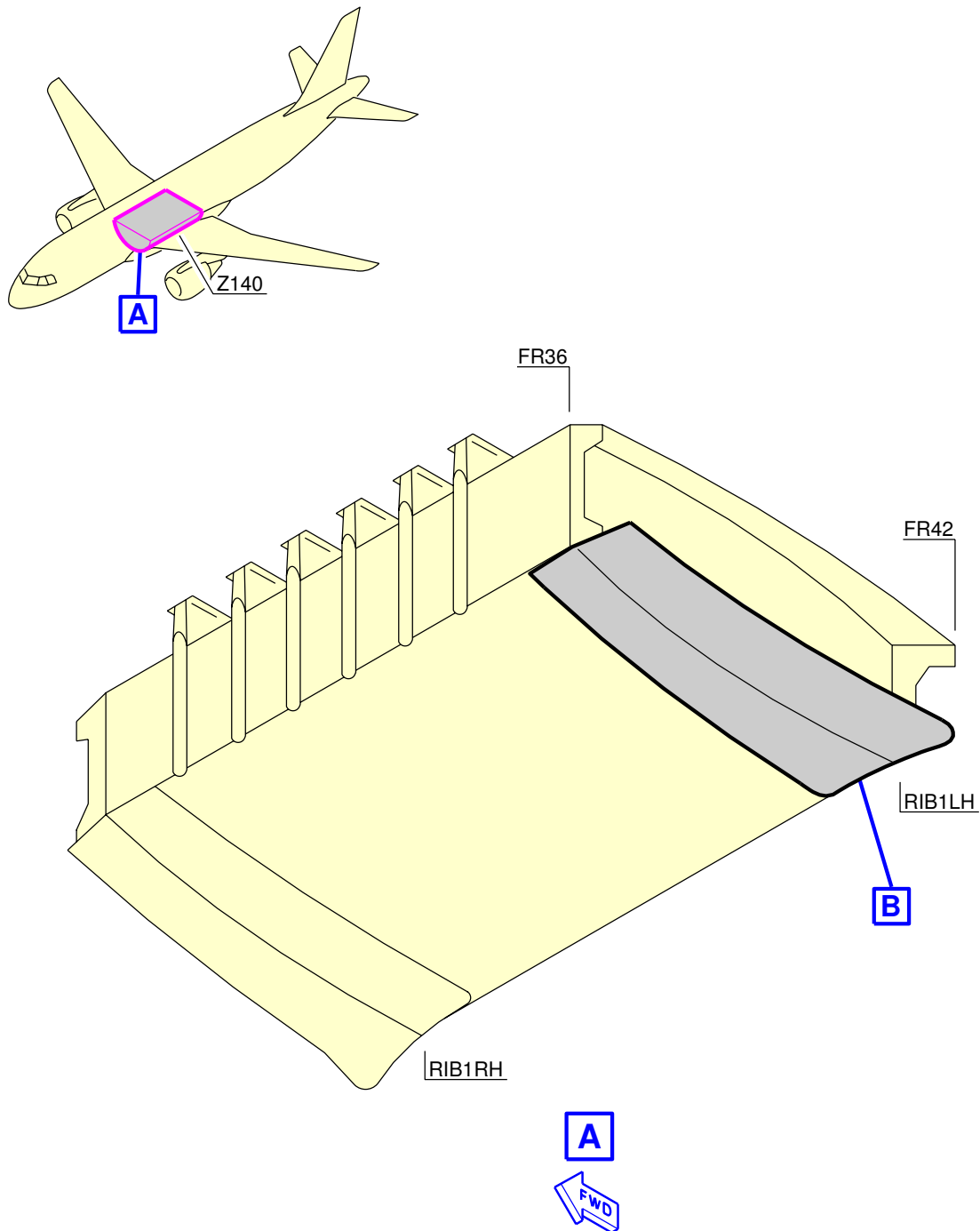


NOTE:

01 FOR MORE INFORMATION OF THE CRITERION MARKS REFER TO THE BOOKLET 06-0009
N_SB_571140_5_BCAA_01_00

Figure A-FBCAA - Sheet 01
Thread Damage

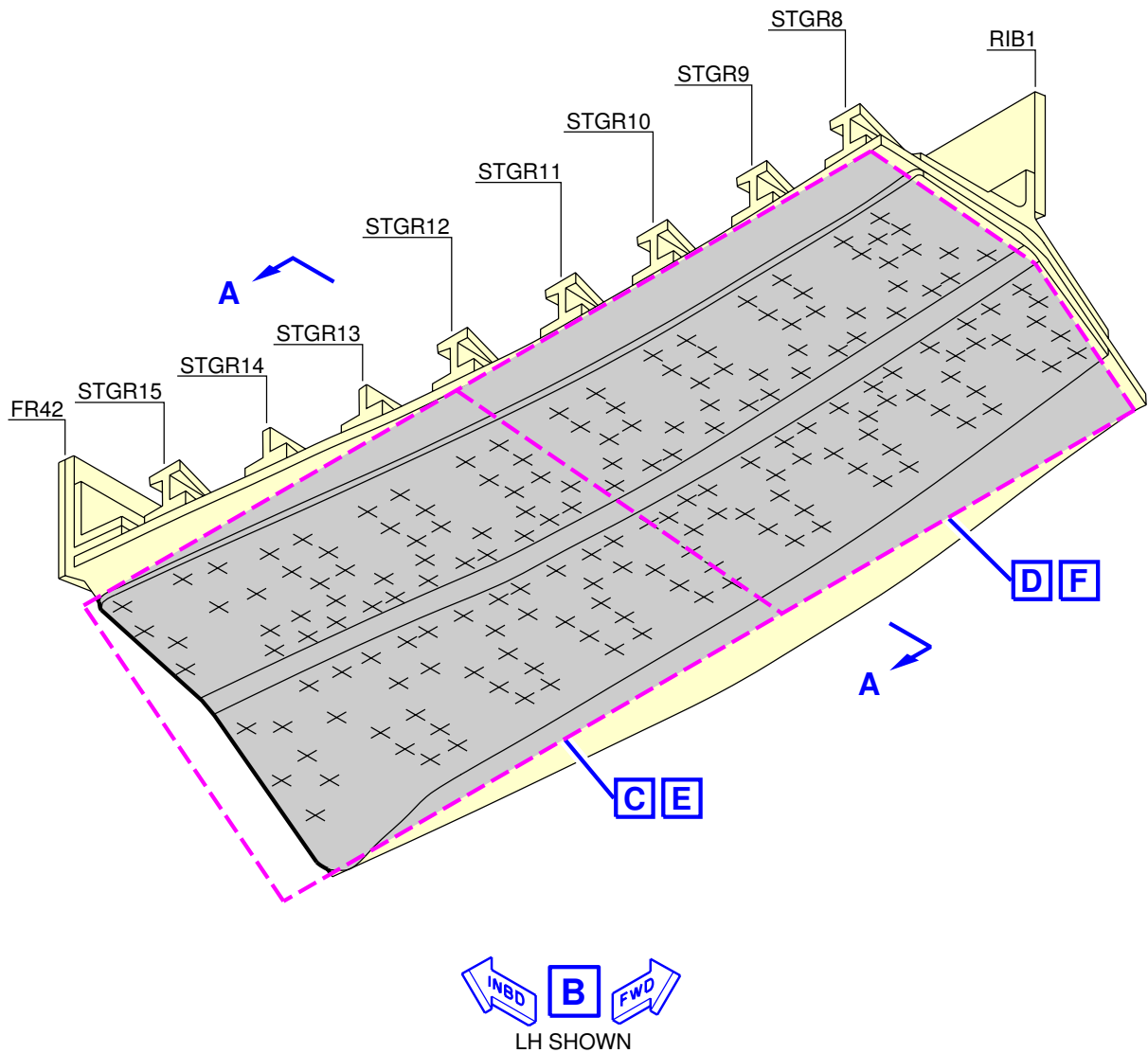
****CONF ALL**



N_SB_571140_5_CAAA_01_02

Figure A-FCAAA - Sheet 01
LH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



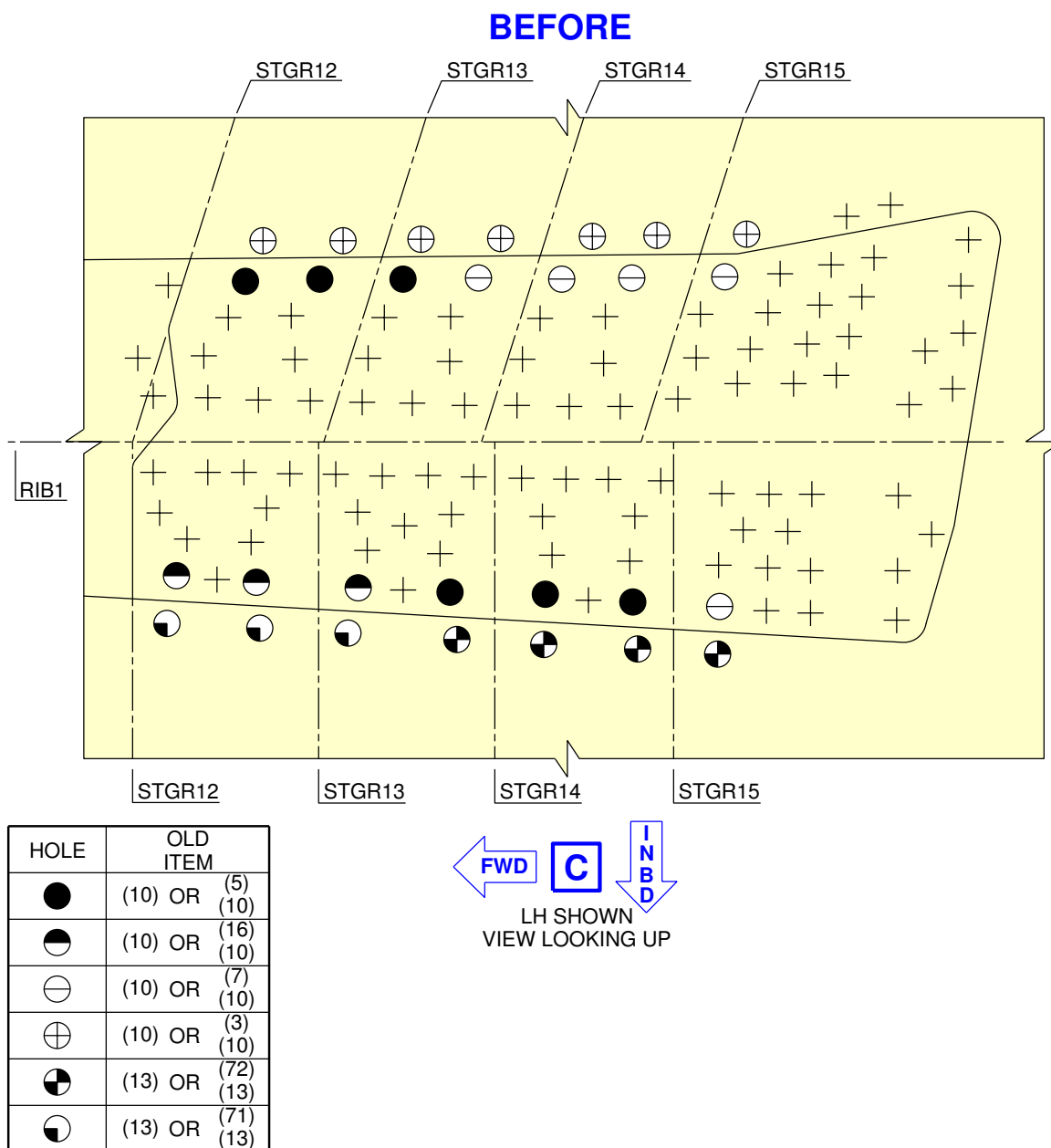
NOTE:

+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CAAA_02_02

Figure A-FCAAA - Sheet 02
LH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



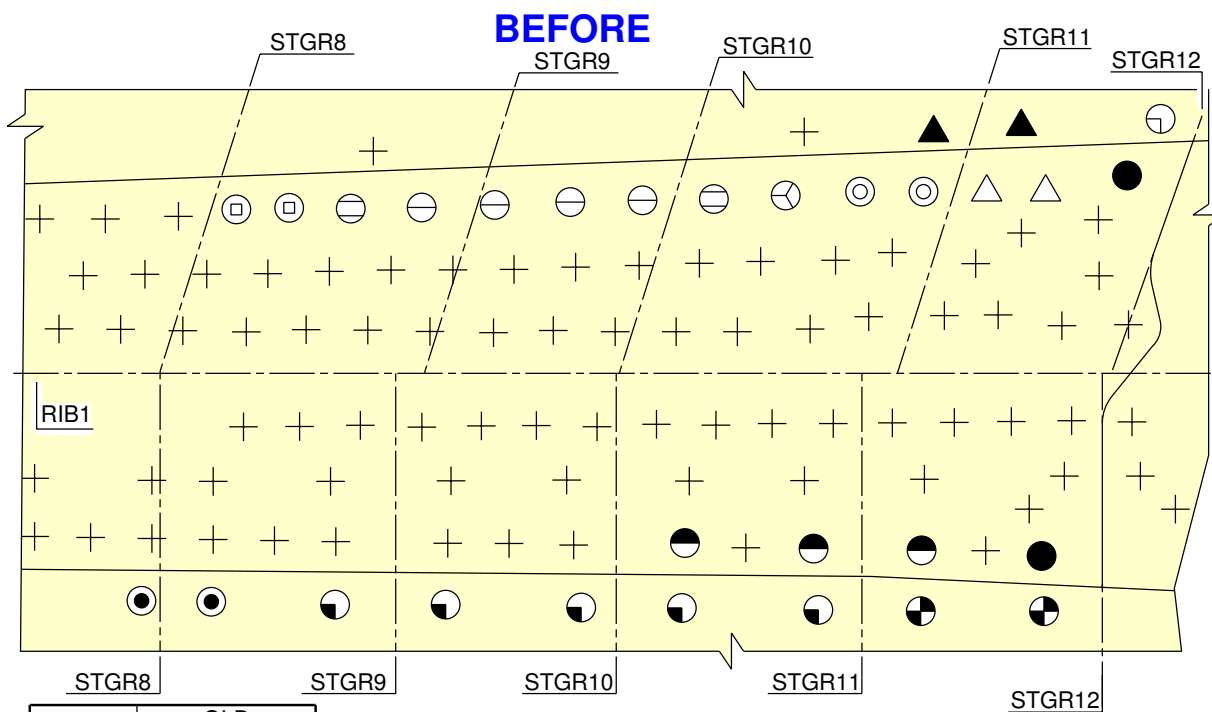
NOTE:

✚ FASTENERS NOT AFFECTED.

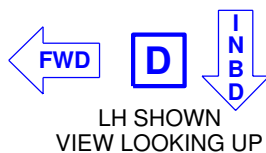
N_SB_571140_5_CAAA_03_02

Figure A-FCAAA - Sheet 03
LH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



HOLE	OLD ITEM
●	(10) OR (5) (10)
◐	(10) OR (16) (10)
⊕	(10) OR (1) (10)
△	(11) OR (14) (11)
▲	(11) OR (5) (11)
⊙	(11) OR (12) (11)
◑	(13) OR (72) (13)
◒	(13) OR (71) (13)
⦿	(13) OR (70) (13)
⊖	(10) OR (77) (10)
⊗	(10) OR (7) (10)
⊘	(10) OR (81) (10)
⊙	(15) OR (79) (15)



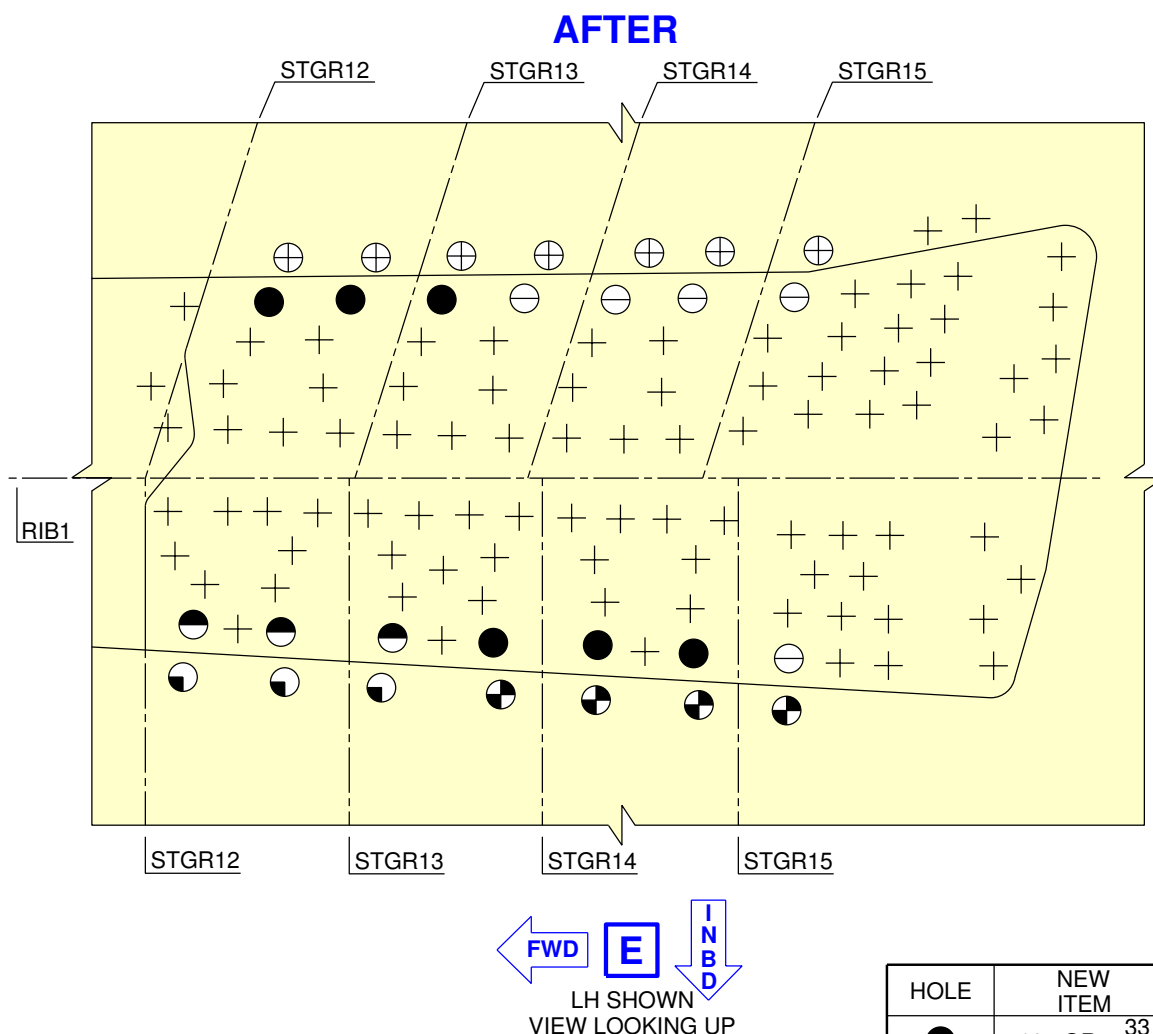
NOTE:

⊕ FASTENERS NOT AFFECTED.

N_SB_571140_5_CAAA_04_01

Figure A-FCAAA - Sheet 04
LH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**









NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

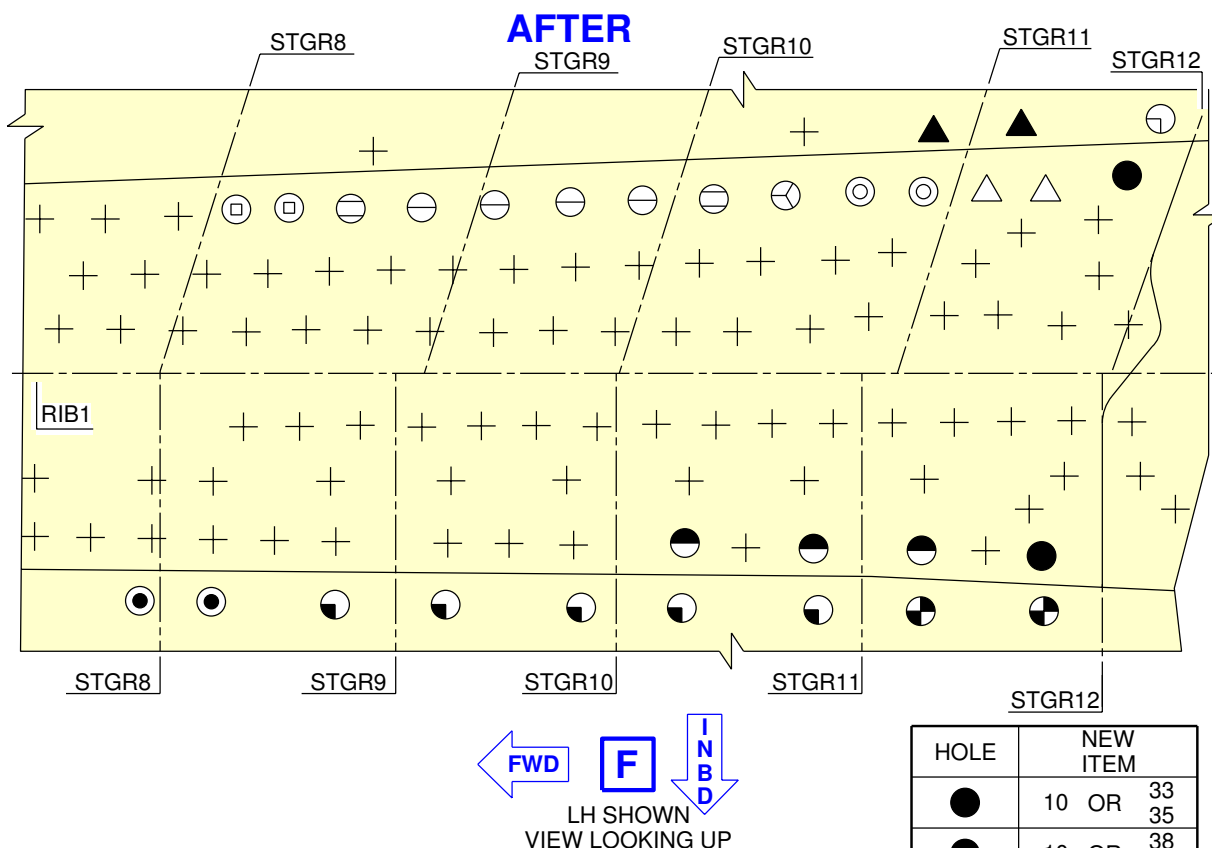
IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

HOLE	NEW ITEM
	10 OR 33 35
	10 OR 38 35
	10 OR 34 35
	10 OR 32 35
	13 OR 75 76
	13 OR 74 76

N_SB_571140_5_CAAA_05_01

Figure A-FCAAA - Sheet 05
LH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



HOLE	NEW ITEM
	10 OR 33 35
	10 OR 38 35
	10 OR 31 35
	11 OR 37 44
	11 OR 36 44
	11 OR 43 44
	13 OR 75 76
	13 OR 74 76
	13 OR 73 76
	10 OR 78 35
	10 OR 34 35
	10 OR 82 35
	15 OR 80 11

NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

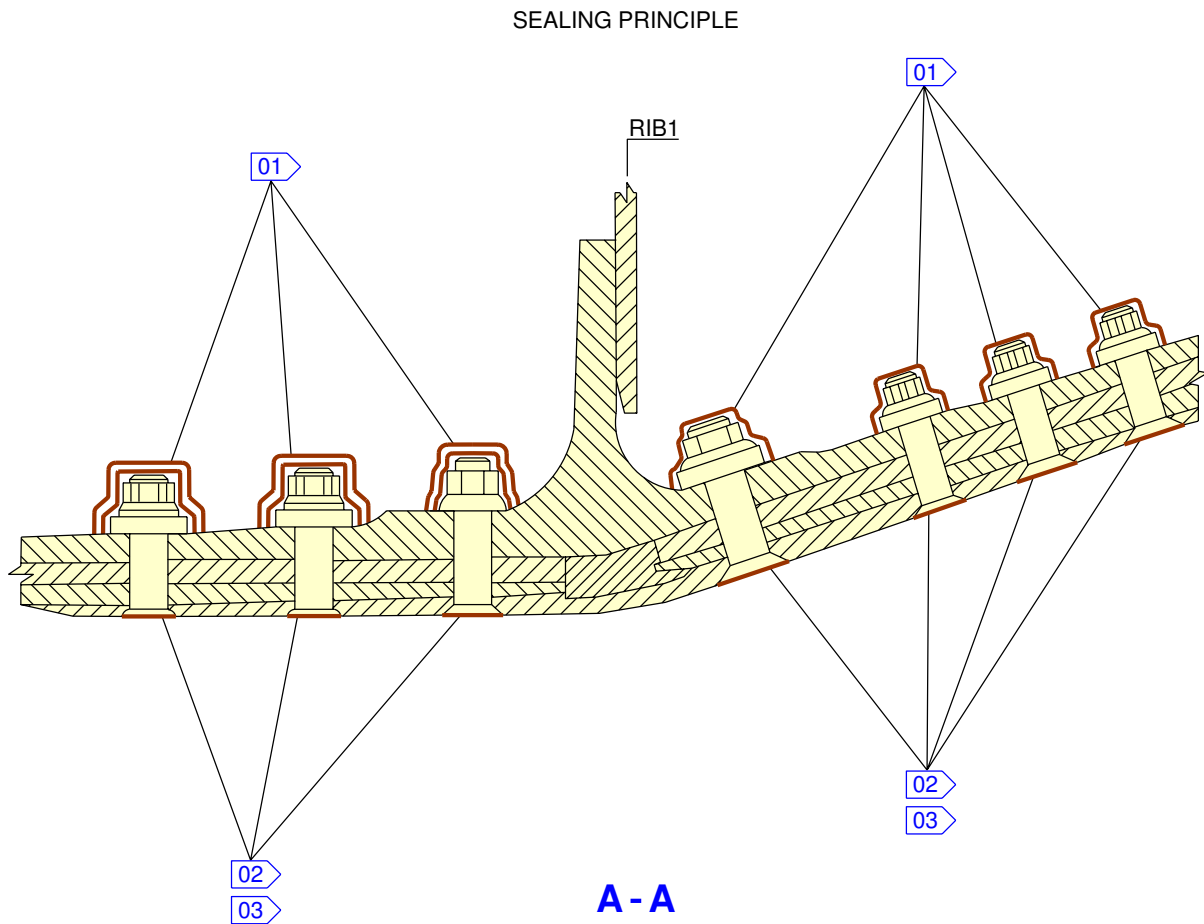
IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

N_SB_571140_5_CAAA_06_01

Figure A-FCAAA - Sheet 06
LH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



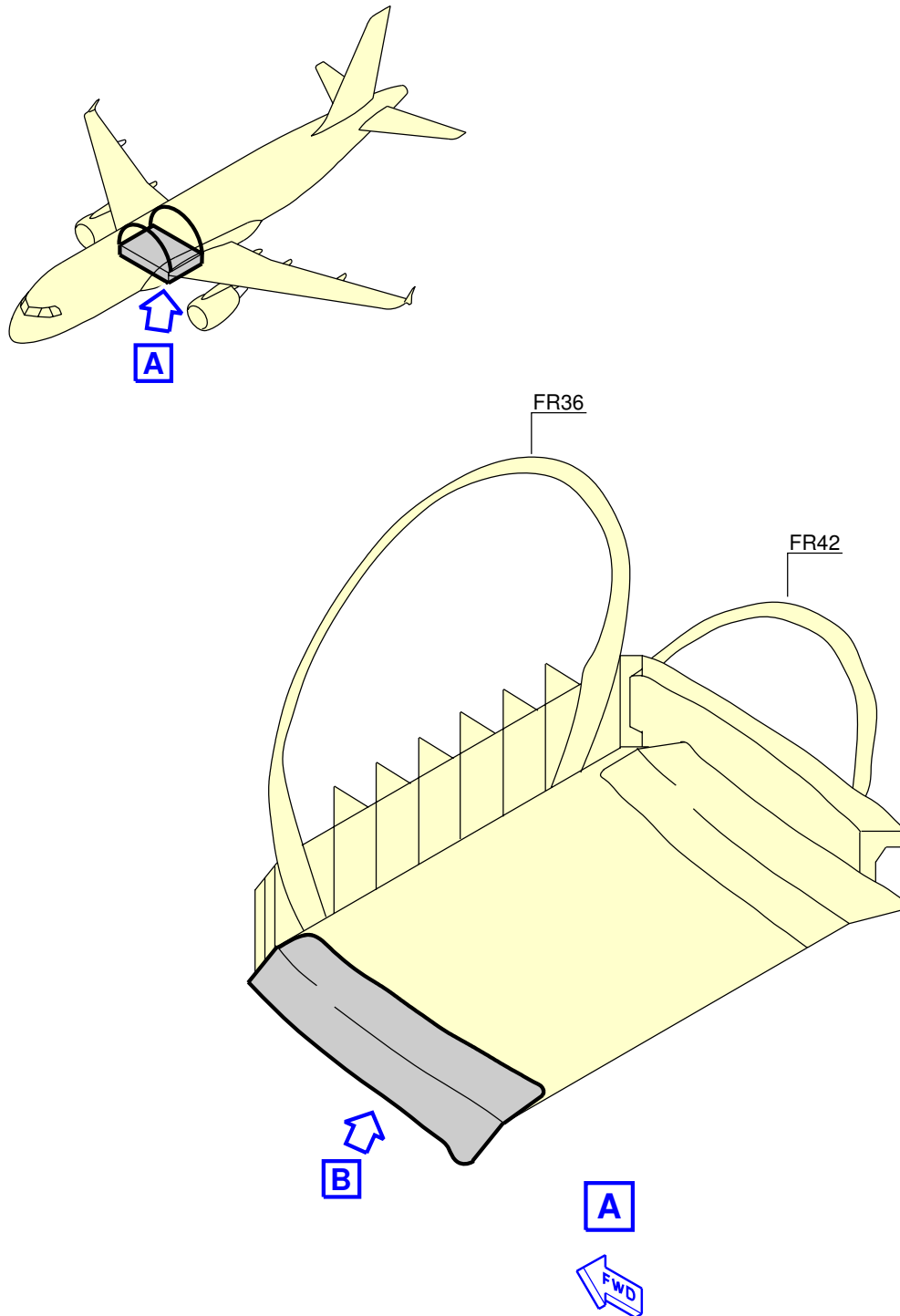
NOTE:

- 01** APPLY MATERIAL No 06ABB1 AND MATERIAL No 06PAG1 ON THE NUTS.
- 02** APPLY MATERIAL No 04EAC2, MATERIAL No 04CMA2 AND MATERIAL No 04JAA3 ON THE BOLTS.
- 03** WET ASSEMBLY, APPLY MATERIAL No 06ABB1.

N_SB_571140_5_CAAA_07_01

Figure A-FCAAA - Sheet 07
LH Side, Replacement of the Lower Panel Fasteners

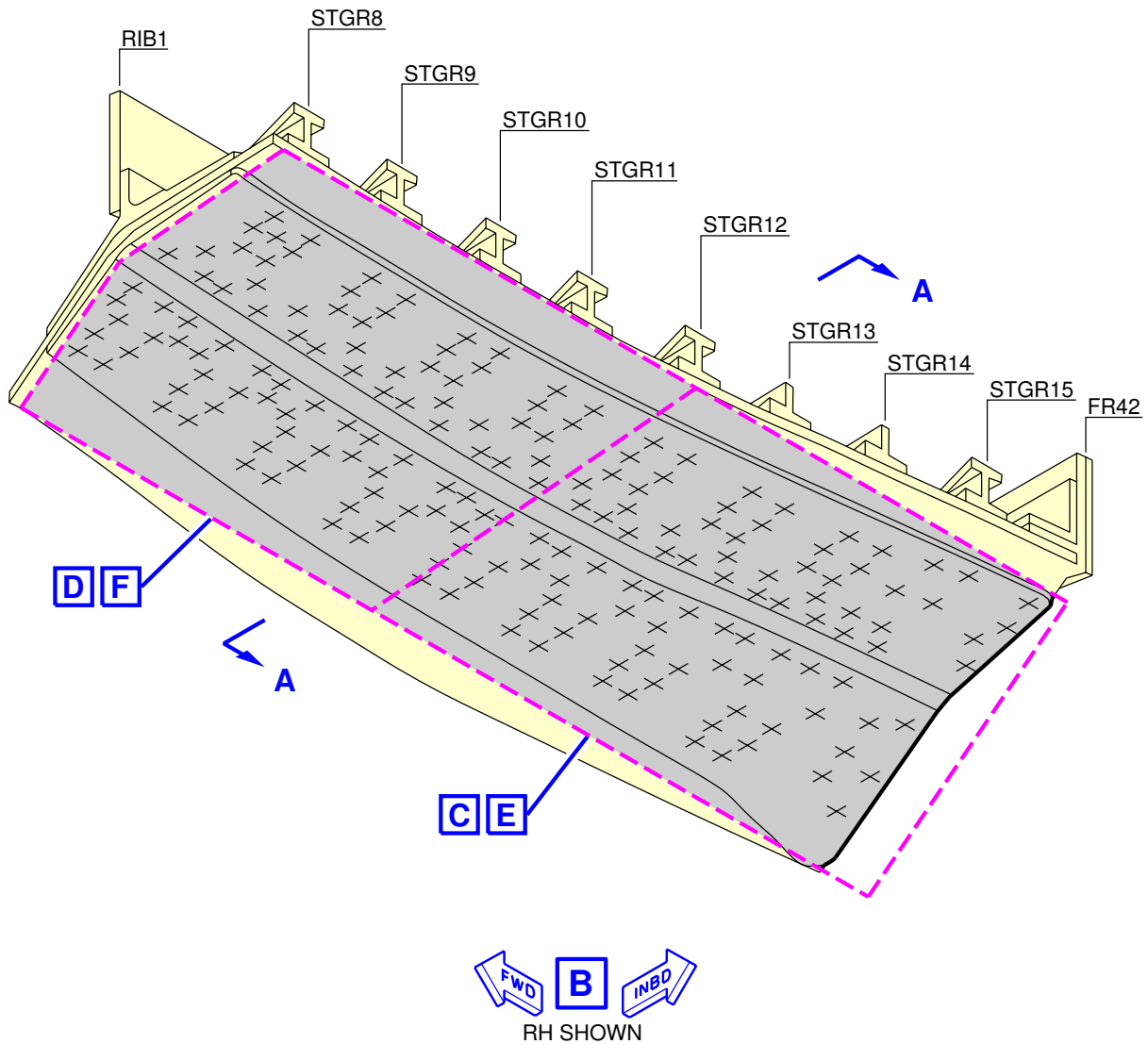
****CONF ALL**



N_SB_571140_5_CBAA_01_00

Figure A-FCBAA - Sheet 01
RH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



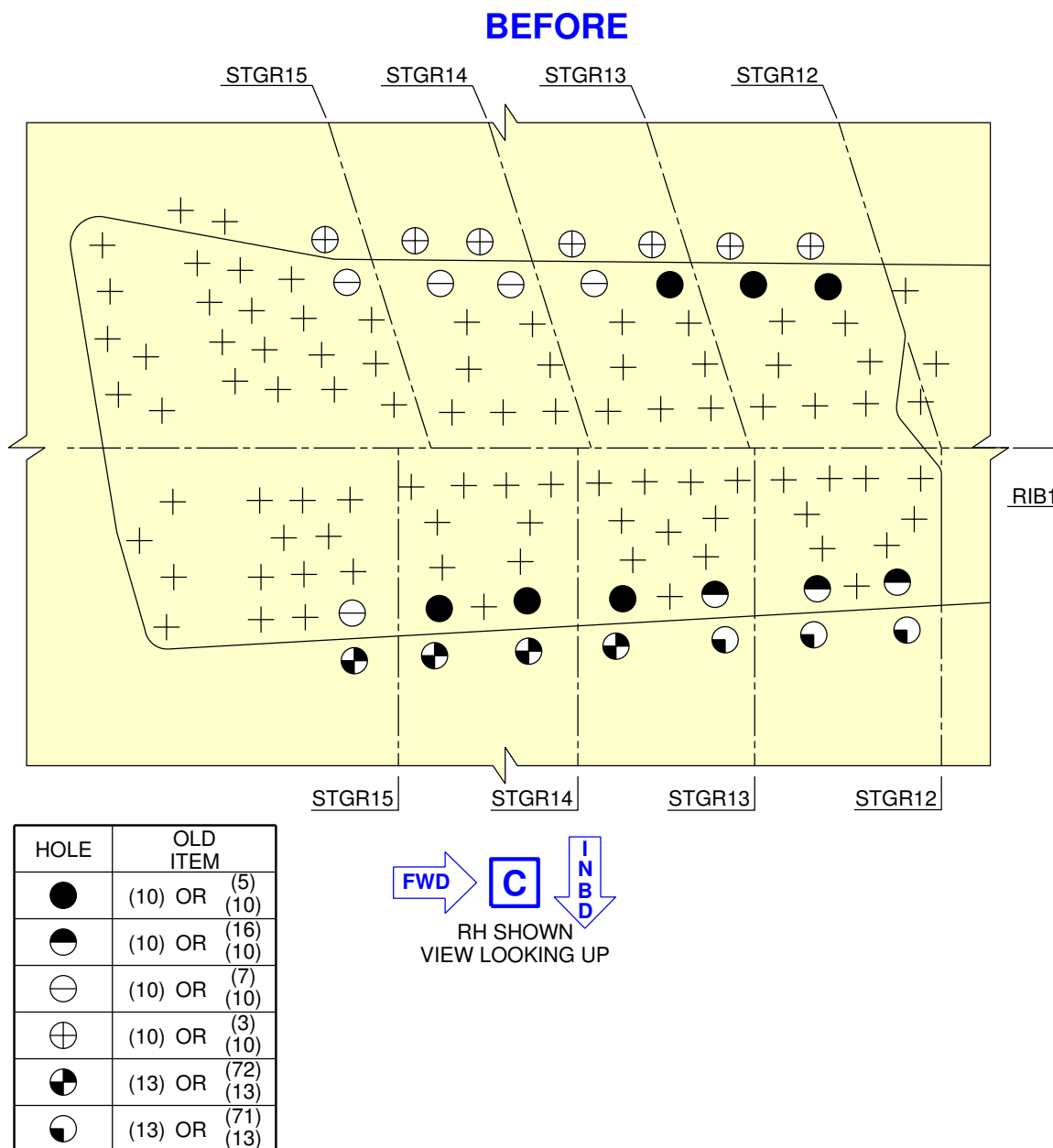
NOTE:

+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CBAA_02_01

Figure A-FCBAA - Sheet 02
RH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



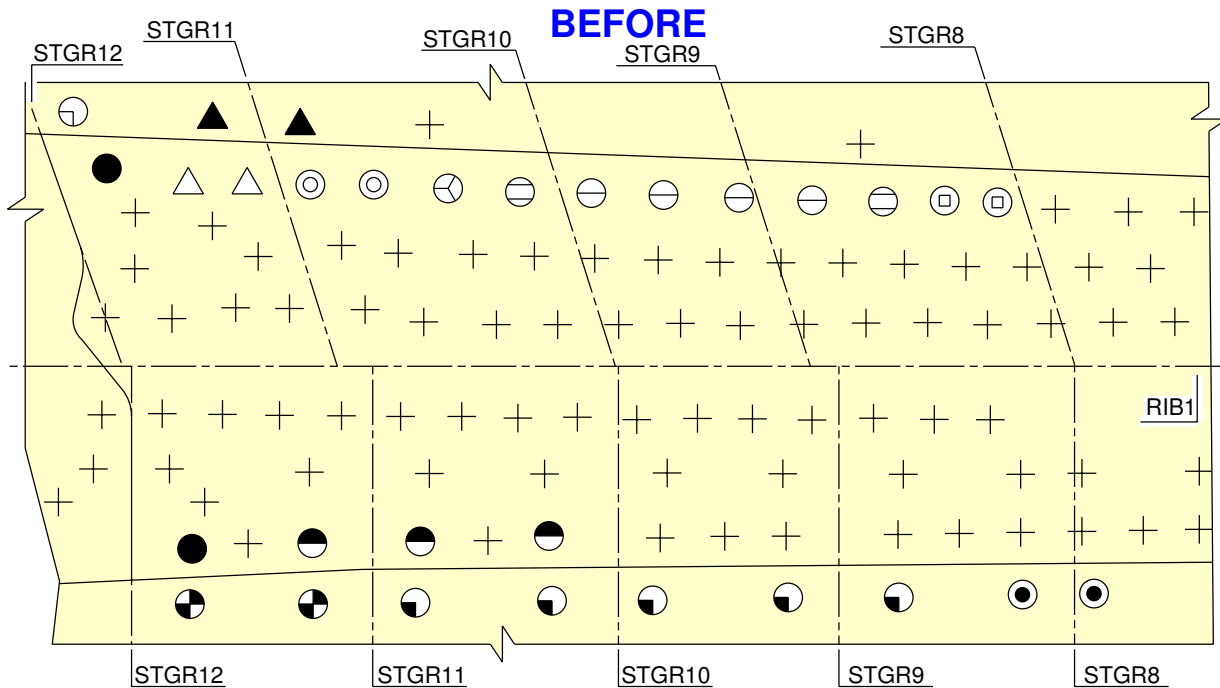
NOTE:














⊕ FASTENERS NOT AFFECTED.

N_SB_571140_5_CBAA_03_01

Figure A-FCBAA - Sheet 03
RH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



HOLE	OLD ITEM
	(10) OR (5) (10)
	(10) OR (16) (10)
	(10) OR (1) (10)
	(11) OR (14) (11)
	(11) OR (5) (11)
	(11) OR (12) (11)
	(13) OR (72) (13)
	(13) OR (71) (13)
	(13) OR (70) (13)
	(10) OR (77) (10)
	(10) OR (7) (10)
	(10) OR (81) (10)
	(15) OR (79) (15)



NOTE:

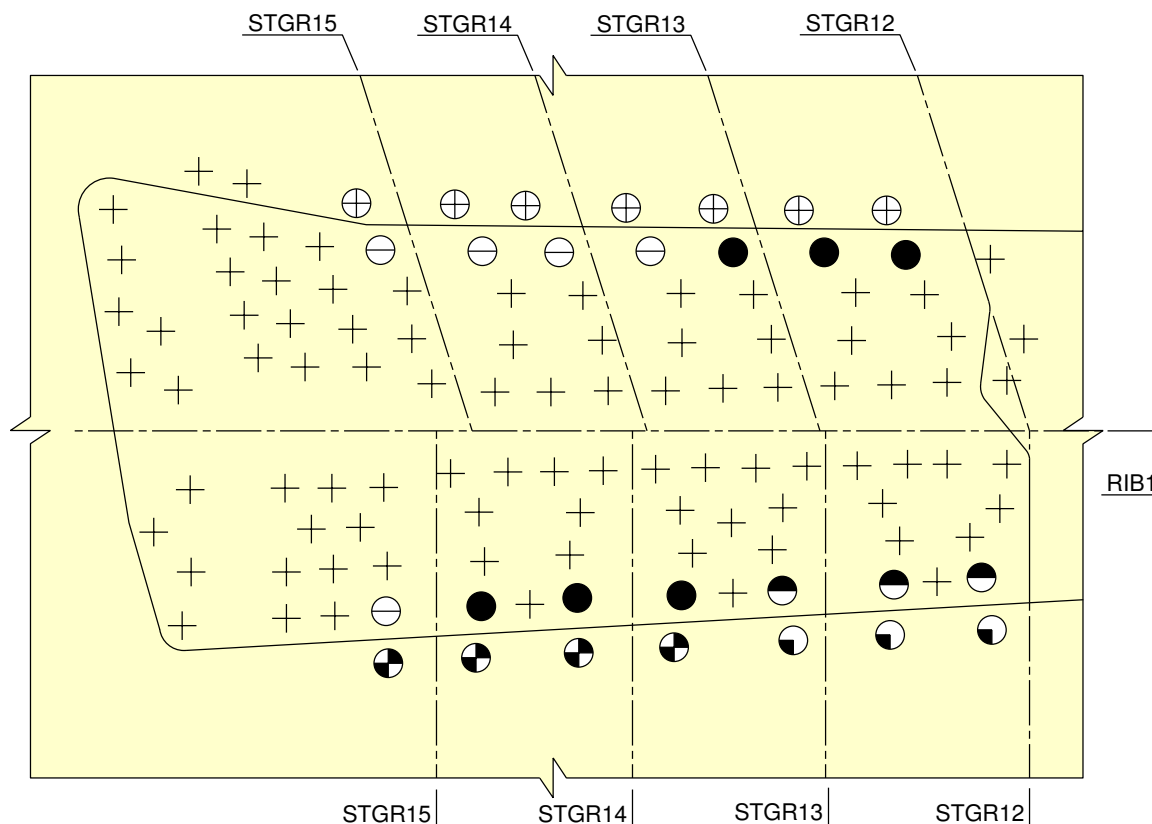
✚ FASTENERS NOT AFFECTED.







N_SB_571140_5_CBAA_04_01

Figure A-FCBAA - Sheet 04
RH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**

AFTER



HOLE	NEW ITEM		
	10	OR	33 35
	10	OR	38 35
	10	OR	34 35
	10	OR	32 35
	13	OR	75 76
	13	OR	74 76

NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCFAA OR FIG. A-FCDA.

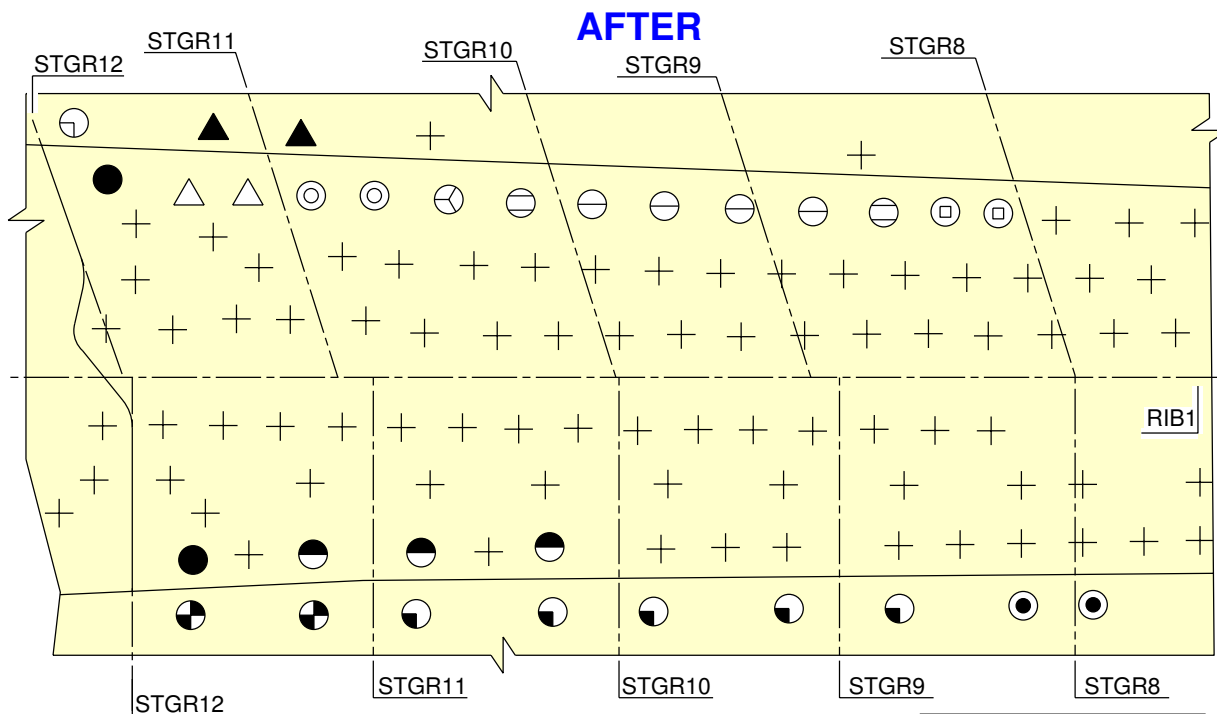
IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).














IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

N_SB_571140_5_CBAA_05_01

Figure A-FCBAA - Sheet 05
RH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



HOLE	NEW ITEM	
	10 OR	33 35
	10 OR	38 35
	10 OR	31 35
	11 OR	37 44
	11 OR	36 44
	11 OR	43 44
	13 OR	75 76
	13 OR	74 76
	13 OR	73 76
	10 OR	78 35
	10 OR	34 35
	10 OR	82 35
	15 OR	80 11

NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

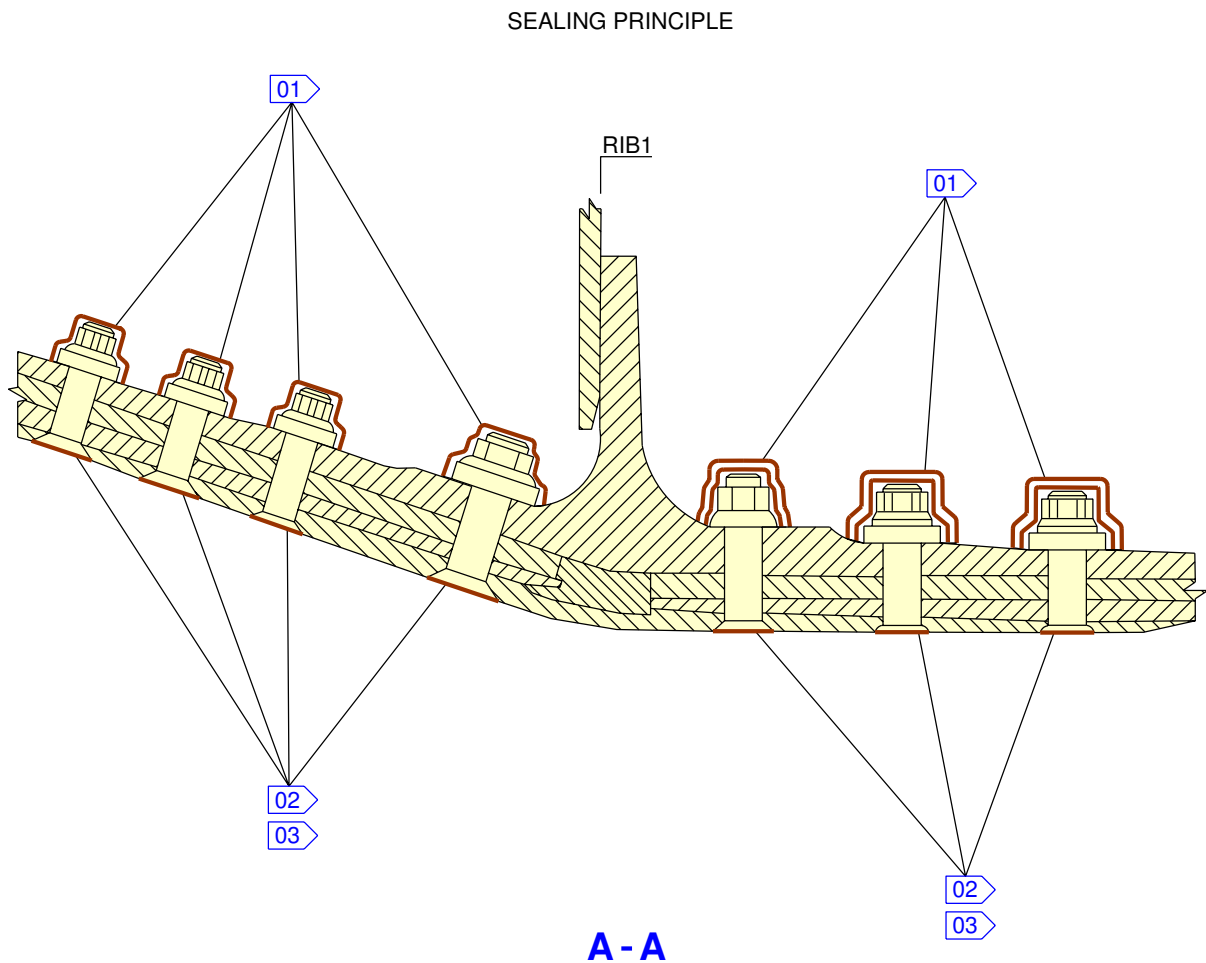
IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

N_SB_571140_5_CBAA_06_01

Figure A-FCBAA - Sheet 06
RH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**



NOTE:

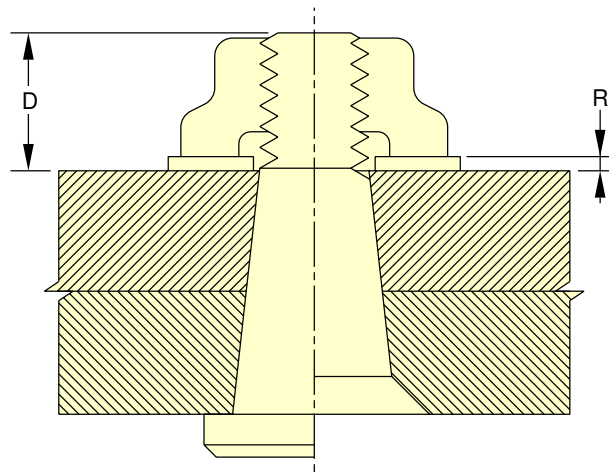
- 01** APPLY MATERIAL No 06AAB1 AND MATERIAL No 06PAG1 ON THE NUTS.
- 02** APPLY MATERIAL No 04EAC2, MATERIAL No 04CAA2 AND MATERIAL No 04JAA3 ON THE BOLTS.
- 03** WET ASSEMBLY, APPLY MATERIAL No 06ABB1.

N_SB_571140_5_CBAA_07_01

Figure A-FCBAA - Sheet 07
RH Side, Replacement of the Lower Panel Fasteners

****CONF ALL**

DEFINITION OF THE WASHER THICKNESS FOR THE ORIGINAL TAPERLOK



NOTE:

D: TAPERLOK LENGTH PROTRUDING BEYOND THE PLATE.
R: THICKNESS OF WASHER REQUIRED.

N_SB_571140_5_CCAA_01_01

Figure A-FCCAA - Sheet 01
Definition of the Washer Thickness for the Original Taperlok

****CONF ALL**

DIAMETER CODE No	D MIN	D		R		WASHERS THICKNESSES CODES	EXAMPLE	REMARKS
		mm	in	mm	in			
6	10.06 mm (0.397 in)	≥ 12.6	0.496	0.80	0.031	A	IF "D" ≥ 13.80 mm (0.543 in) INSTALL THE WASHERS NSA5372–616AX + NSA5372–616EX	IF "D" < 10.06 mm (0.397 in) INSTALL ABS1418 NEXT LENGHT CODE
		≥ 13	0.512	1.20	0.047	E		
		≥ 13.4	0.528	1.60	0.063	B		
		≥ 13.8	0.543	2	0.079	A+E		
		≥ 14.2	0.560	2.40	0.094	C		
		≥ 14.6	0.575	2.80	0.110	E+B		
		≥ 15	0.590	3.20	0.126	D		
		≥ 15.4	0.606	3.60	0.142	E+C		
		≥ 15.8	0.622	4	0.157	B+C		
		≥ 16.2	0.638	4.40	0.173	E+D		
		≥ 16.6	0.653	4.80	0.189	B+D		
		≥ 17	0.669	5.20	0.205	E+B+C		
		≥ 17.4	0.685	5.60	0.220	C+D		
		≥ 17.8	0.701	6	0.236	E+B+D		
		≥ 18.2	0.716	6.40	0.252	D+D		
≥ 18.6	0.732	6.80	0.268	E+C+D				

N_SB_571140_5_CCAA_02_01

Figure A-FCCAA - Sheet 02
Definition of the Washer Thickness for the Original Taperlok

****CONF ALL**

VALID FOR ASNA2531 AND ASNA2532 NUT

DIAMETER CODE No	D MIN	D		R		WASHERS THICKNESSES CODES	EXAMPLES	REMARKS
		mm	in	mm	in			
7	11.25 mm (0.443 in)	≥ 13.5	0.531	0.80	0.031	A	IF "D" ≥ 15.50 mm (0.610 in) INSTALL THE WASHERS NSA5372–716EX + NSA5372–716BX	IF "D" < 11.25 mm (0.443 in) INSTALL ABS1418 NEXT LENGHT CODE
		≥ 13.9	0.547	1.20	0.047	E		
		≥ 14.3	0.563	1.60	0.063	B		
		≥ 14.8	0.583	2	0.079	A+E		
		≥ 15.1	0.594	2.40	0.094	C		
		≥ 15.5	0.610	2.80	0.110	E+B		
		≥ 15.9	0.626	3.20	0.126	D		
		≥ 16.3	0.642	3.60	0.142	E+C		
		≥ 16.8	0.661	4	0.157	B+C		IF "D" < 13.46 mm (0.530 in) NO WASHER TO INSTALL
		≥ 17.1	0.673	4.40	0.173	E+D		
		≥ 17.5	0.689	4.80	0.189	B+D		
		≥ 17.9	0.705	5.20	0.205	E+B+C		
		≥ 18.3	0.720	5.60	0.220	C+D		
		≥ 18.7	0.736	6	0.236	E+B+D		
		≥ 19.1	0.752	6.40	0.252	D+D		
		≥ 19.5	0.768	6.80	0.268	E+C+D		
		≥ 19.9	0.783	7.20	0.283	C+C+C		
8	12.07 mm (0.476 in)	≥ 14.3	0.563	0.80	0.031	A	IF "D" ≥ 16.30 mm (0.642 in) INSTALL THE WASHERS NSA5372–816EX + NSA5372–816BX	IF "D" < 12.07 mm (0.476 in) INSTALL ABS1418 NEXT LENGHT CODE
		≥ 14.8	0.583	1.20	0.047	E		
		≥ 15.1	0.594	1.60	0.063	B		
		≥ 15.5	0.610	2	0.079	A+E		
		≥ 15.9	0.626	2.40	0.094	C		
		≥ 16.3	0.642	2.80	0.110	E+B		
		≥ 16.8	0.661	3.20	0.126	D		
		≥ 17.1	0.673	3.60	0.142	E+C		IF "D" < 14.27 mm (0.562 in) NO WASHER TO INSTALL
		≥ 17.5	0.689	4	0.157	B+C		
		≥ 17.9	0.705	4.40	0.173	E+D		
		≥ 18.3	0.720	4.80	0.189	B+D		
		≥ 18.7	0.736	5.20	0.205	E+B+C		
		≥ 19.1	0.752	5.60	0.220	C+D		
		≥ 19.5	0.768	6	0.236	E+B+D		
		≥ 19.9	0.783	6.40	0.252	D+D		
		≥ 20.3	0.799	6.80	0.268	E+C+D		

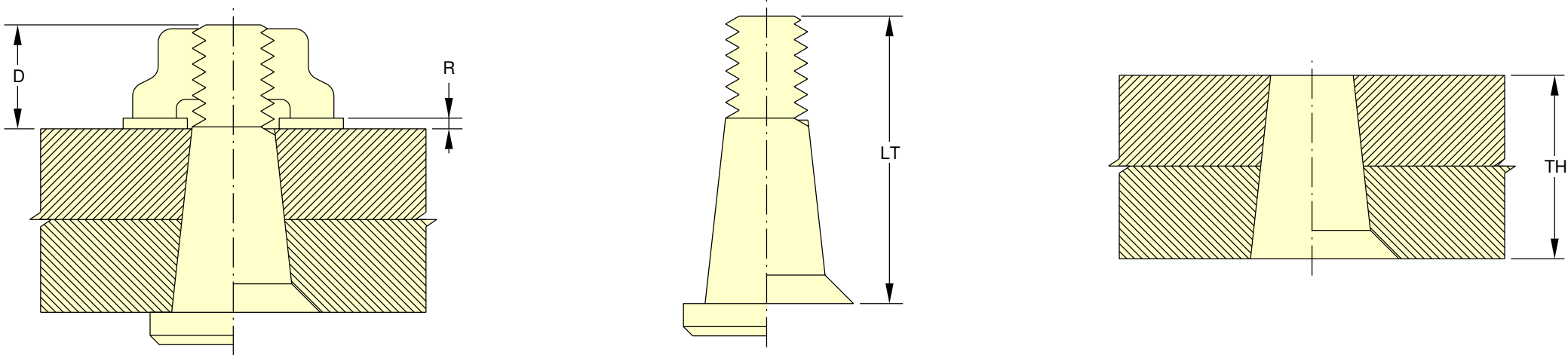
NOTE:
IF NECESSARY, THE INSTALLATION OF THREE WASHERS IS APPROVED IN THIS CONDITION.

N_SB_571140_5_CCAA_03_01

Figure A-FCCAA - Sheet 03
Definition of the Washer Thickness for the Original Taperlok

**CONF ALL

DEFINITION OF THE WASHER THICKNESS
FOR THE NEW TAPERLOK



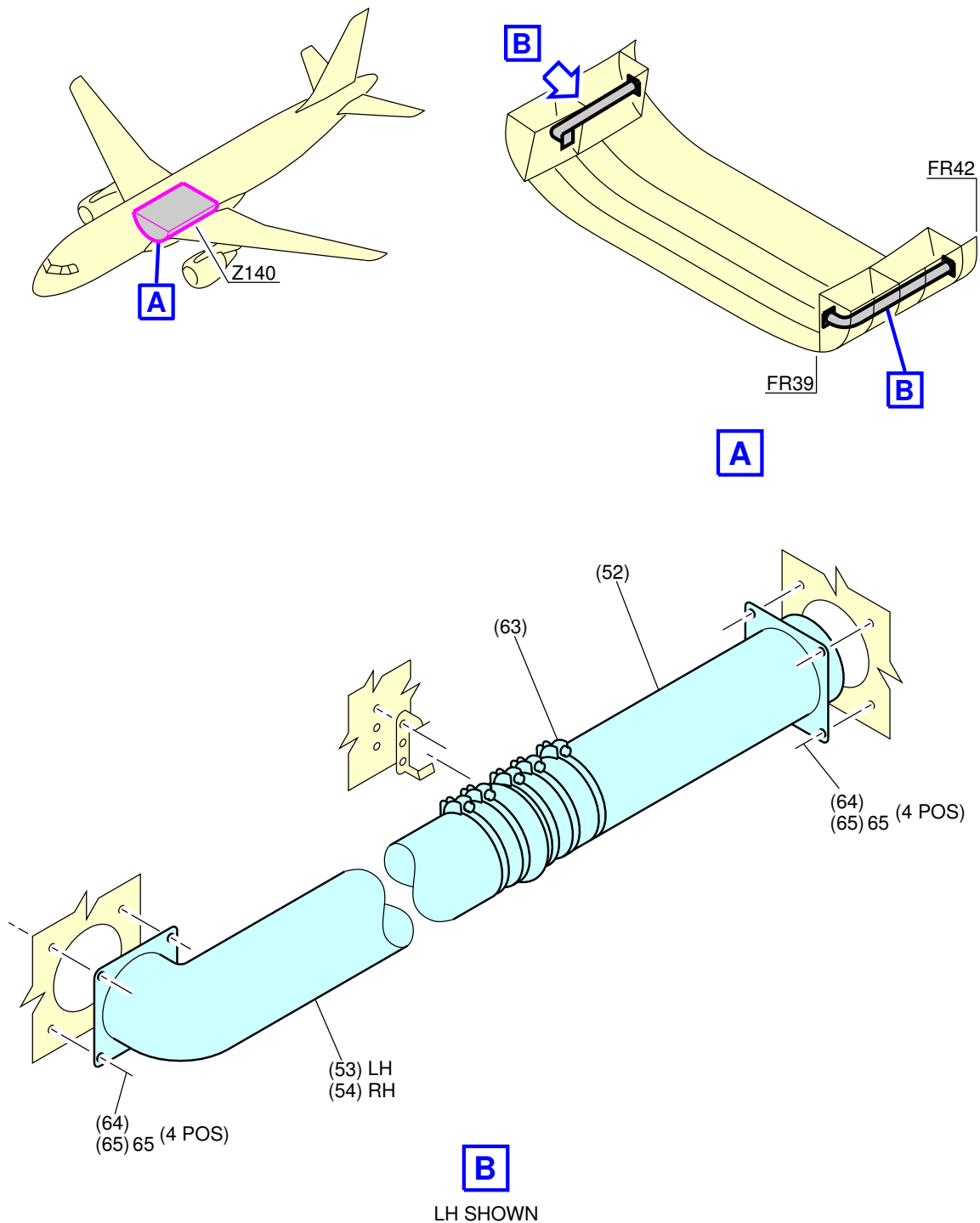
DIAMETER CODE No	D MIN	D		R		WASHERS THICKNESSES CODES	EXAMPLES	REMARKS
		mm	in	mm	in			
7	12.50 mm (0.493 in)	≥ 14.20	0.559	0.8	0.031	A	IF "D" ≥ 14.20 mm (0.559 in) INSTALL THE WASHER NSA5372-716AX	BETWEEN 12.50 mm (0.493 in) AND 14.20 mm (0.559 in) NO WASHER TO INSTALL
		≥ 14.60	0.574	1.2	0.047	E		
		≥ 15	0.590	1.6	0.063	B		
8	13.52 mm (0.533 in)	≥ 15.30	0.602	0.8	0.031	A	IF "D" ≥ 15.70 mm (0.618 in) INSTALL THE WASHER NSA5372-816EX	BETWEEN 13.52 mm (0.533 in) AND 15.30 mm (0.602 in) NO WASHER TO INSTALL
		≥ 15.70	0.618	1.2	0.047	E		
		≥ 16.10	0.633	1.6	0.063	B		
9	15.42 mm (0.608 in)	≥ 17.30	0.681	0.8	0.031	A	IF "D" ≥ 18.10 mm (0.712 in) INSTALL THE WASHER NSA5372-916BX	BETWEEN 15.42 mm (0.608 in) AND 17.29 mm (0.680 in) NO WASHER TO INSTALL
		≥ 17.70	0.696	1.2	0.047	E		
		≥ 18.10	0.712	1.6	0.063	B		

NOTE:
D: TAPERLOK LENGTH PROTRUDING BEYOND THE PLATE.
R: THICKNESS OF WASHER REQUIRED.
LT: LENGTH OF THE REPAIR TAPERLOK.
TH: TOTAL THICKNESS.
BEFORE INSTALLATION OF THE NEW TAPERLOK, YOU MUST CALCULATE THE "D" VALUE: D = LT – TH.
TABLE ONLY VALID FOR THE OPTIONAL ABS1418 TAPERLOK.
IF THE LENGTH OF THE ABS1418 TAPERLOK IS CORRECTLY MEASURED,
THE INSTALLATION OF THE WASHER IS NOT REQUIRED.
IF THE LENGTH OF THE ABS1418 IS LESS THAN "D MIN" INSTALL THE NEXT LENGTH CODE OF FASTENER.

N_SB_571140_5_CDAA_01_01

Figure A-FCDA - Sheet 01
Definition of the Washer Thickness for the New Taperlok

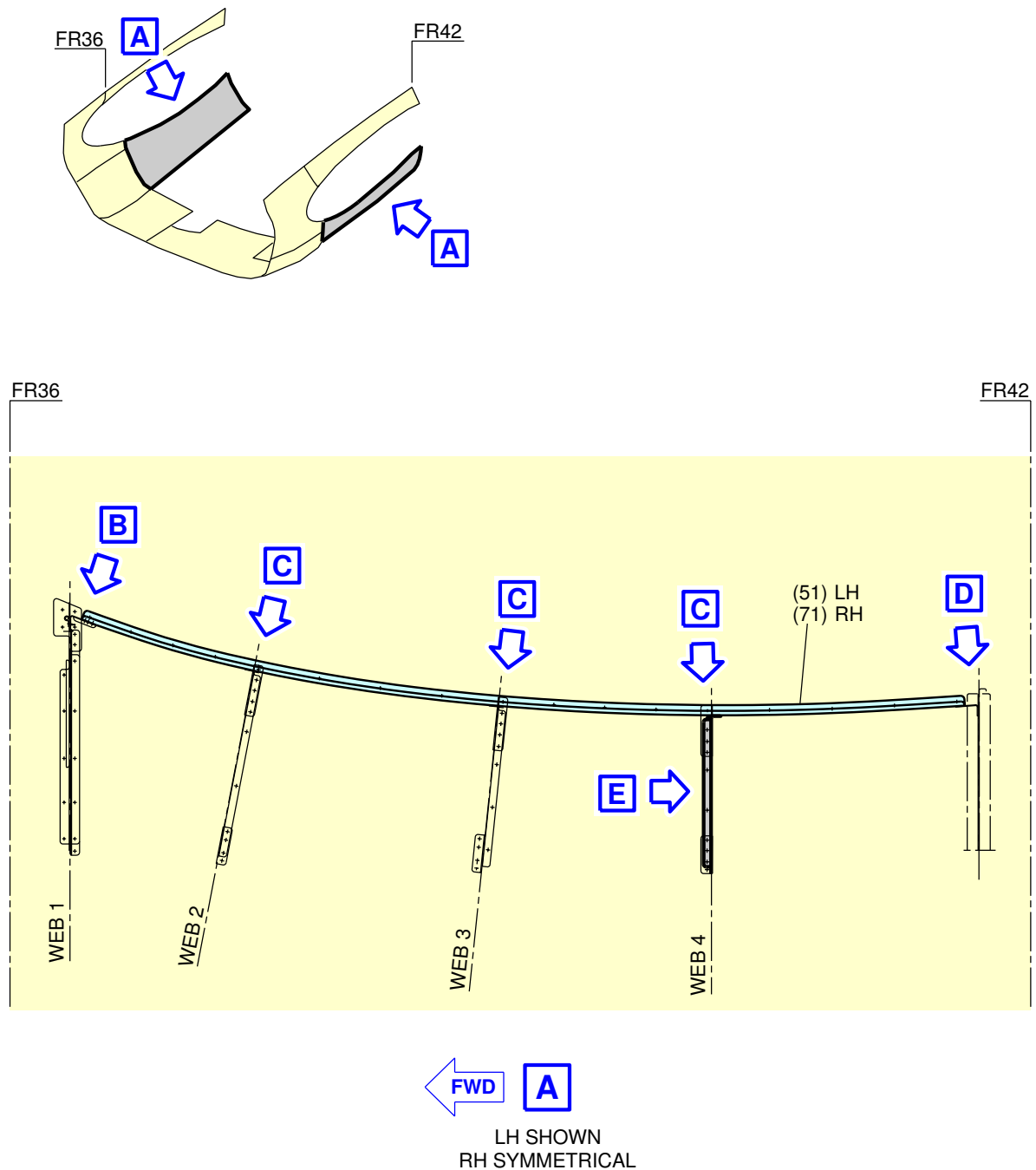
****CONF ALL**



N_SB_571140_5_DAAA_01_01

Figure A-FDAAA - Sheet 01
Removal/Installation of the Cooling Duct

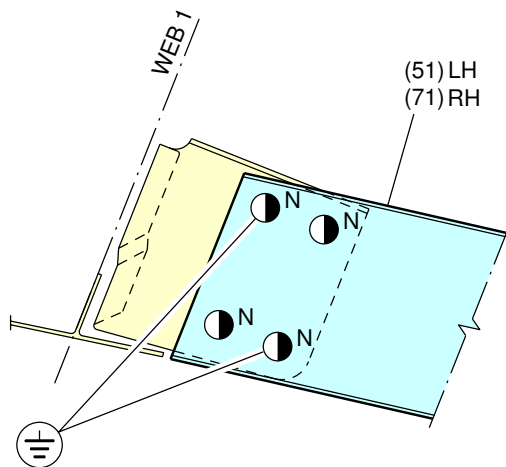
****CONF ALL**



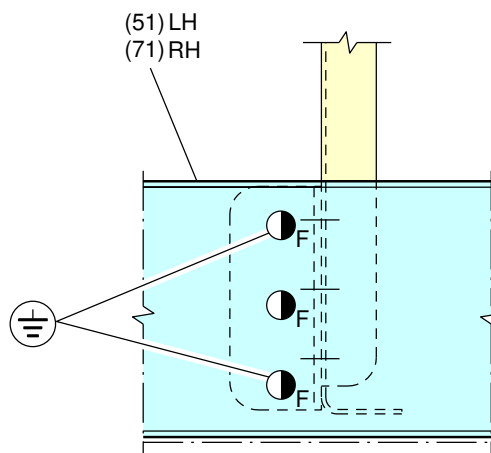
N_SB_571140_5_DBAA_01_01

Figure A-FDBAA - Sheet 01
Removal/Installation of the Belly Fairing Structure

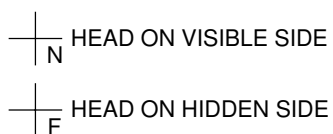
****CONF ALL**





B

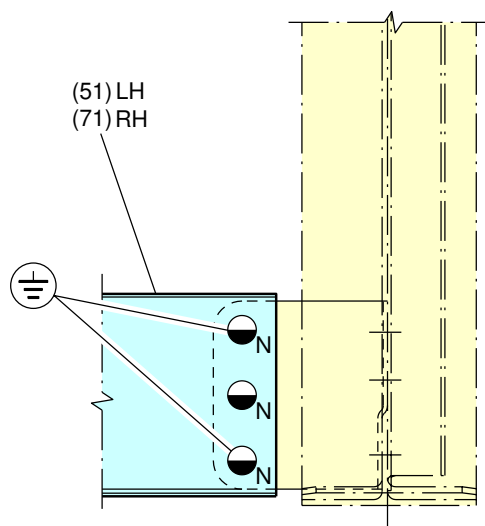


C



HOLE	OLD ITEM	NEW ITEM
	(60) (61)	60 61
	(62) (61)	62 61

NOTE :
APPLY MAT. No. 06AAA1 TO THE
INTERFACES OF THE PARTS

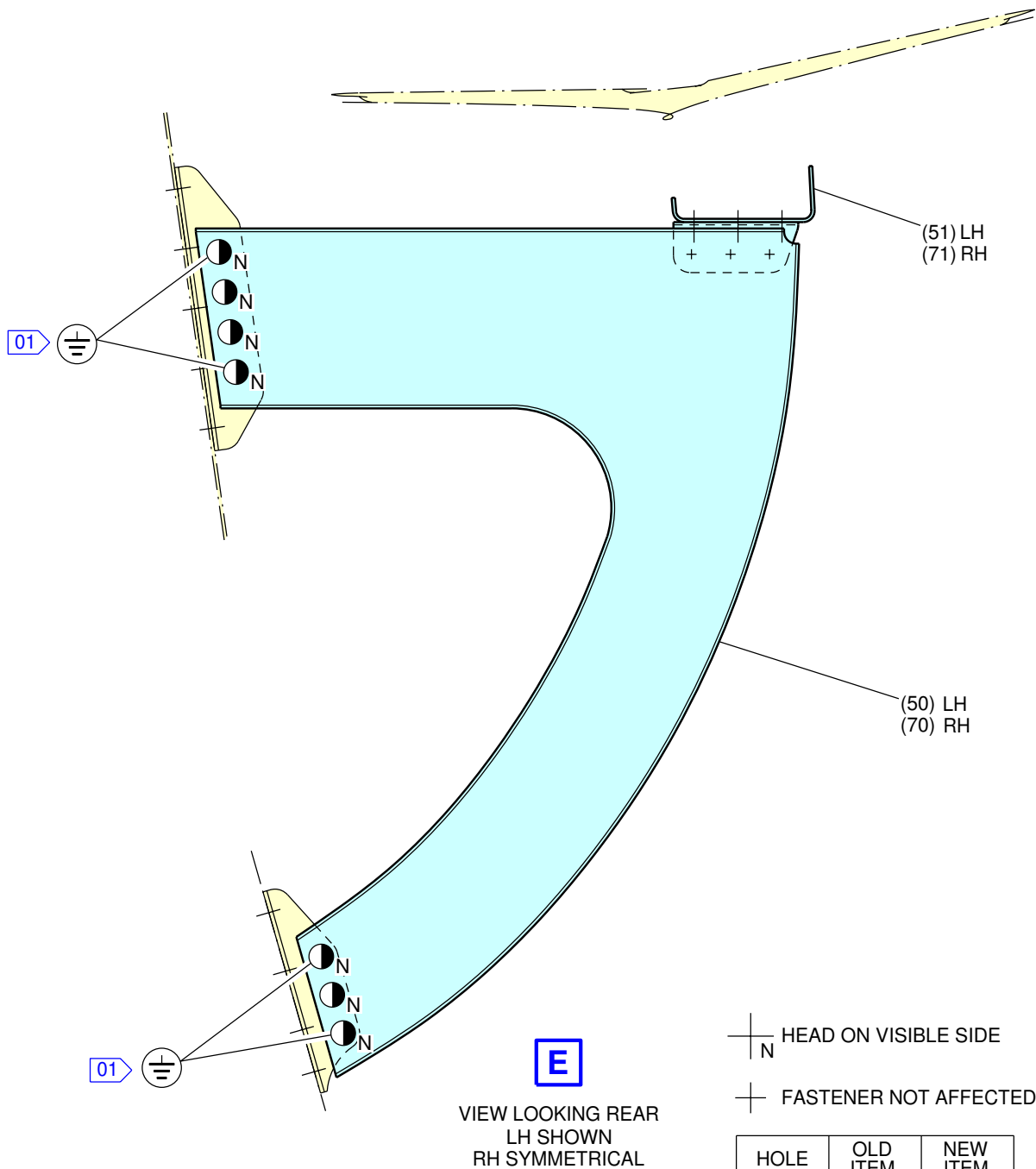


D

N_SB_571140_5_DBAA_02_00

Figure A-FDBAA - Sheet 02
Removal/Installation of the Belly Fairing Structure

****CONF ALL**



NOTE:

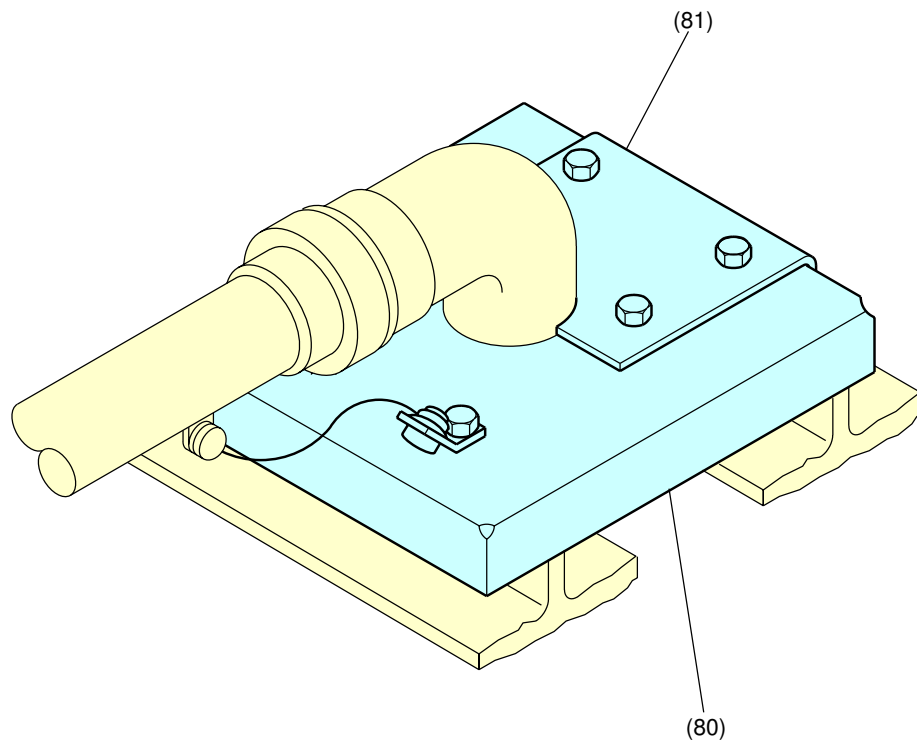
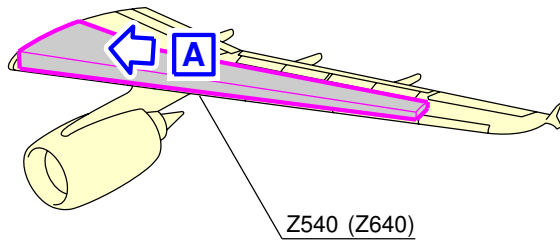
APPLY MAT. No. 06AAA1 TO THE INTERFACES OF THE STRUCTURE AND ALL THE PARTS

(01) DEPENDING ON AIRCRAFT CONFIGURATION

N_SB_571140_5_DBAA_03_00

Figure A-FDBAA - Sheet 03
Removal/Installation of the Belly Fairing Structure

****CONF ALL**

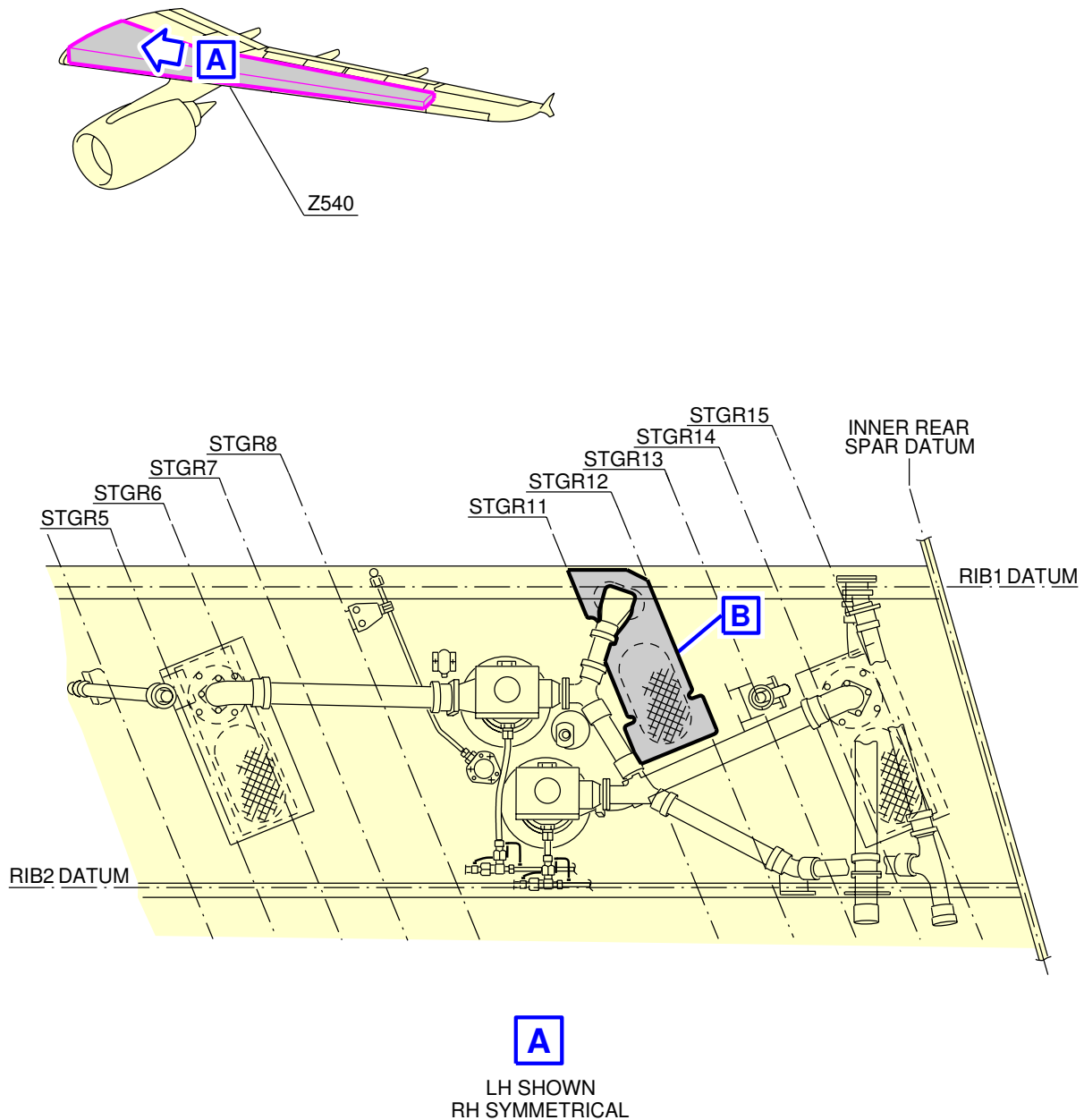


LH SHOWN
RH SYMMETRICAL

N_SB_571140_5_DCAA_01_01

Figure A-FDCAA - Sheet 01
Removal/Installation of the Fuel Pump Strainer Covers

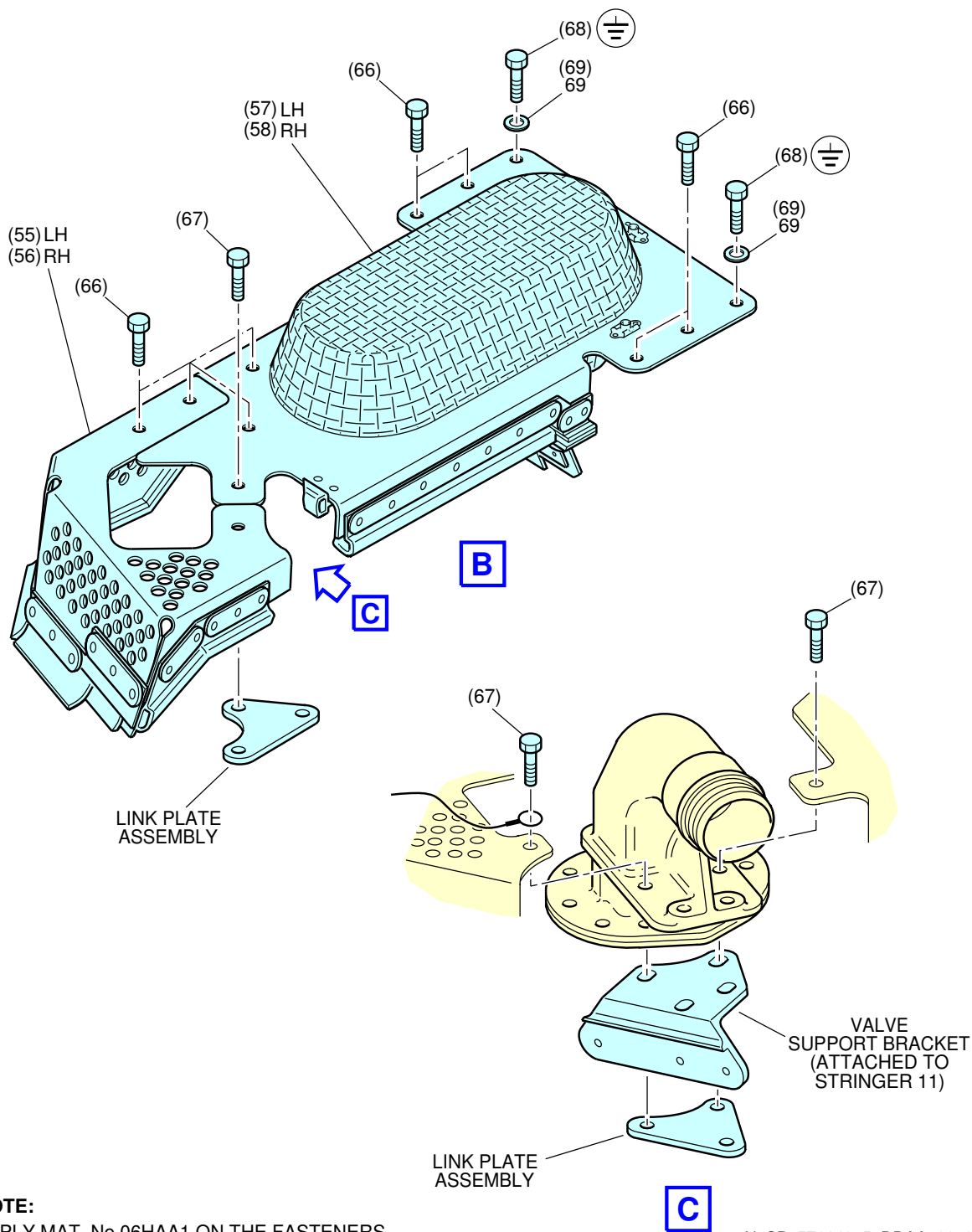
****CONF ALL**



N_SB_571140_5_DDAA_01_01

Figure A-FDDAA - Sheet 01
Removal/Installation of the Fuel Strainer

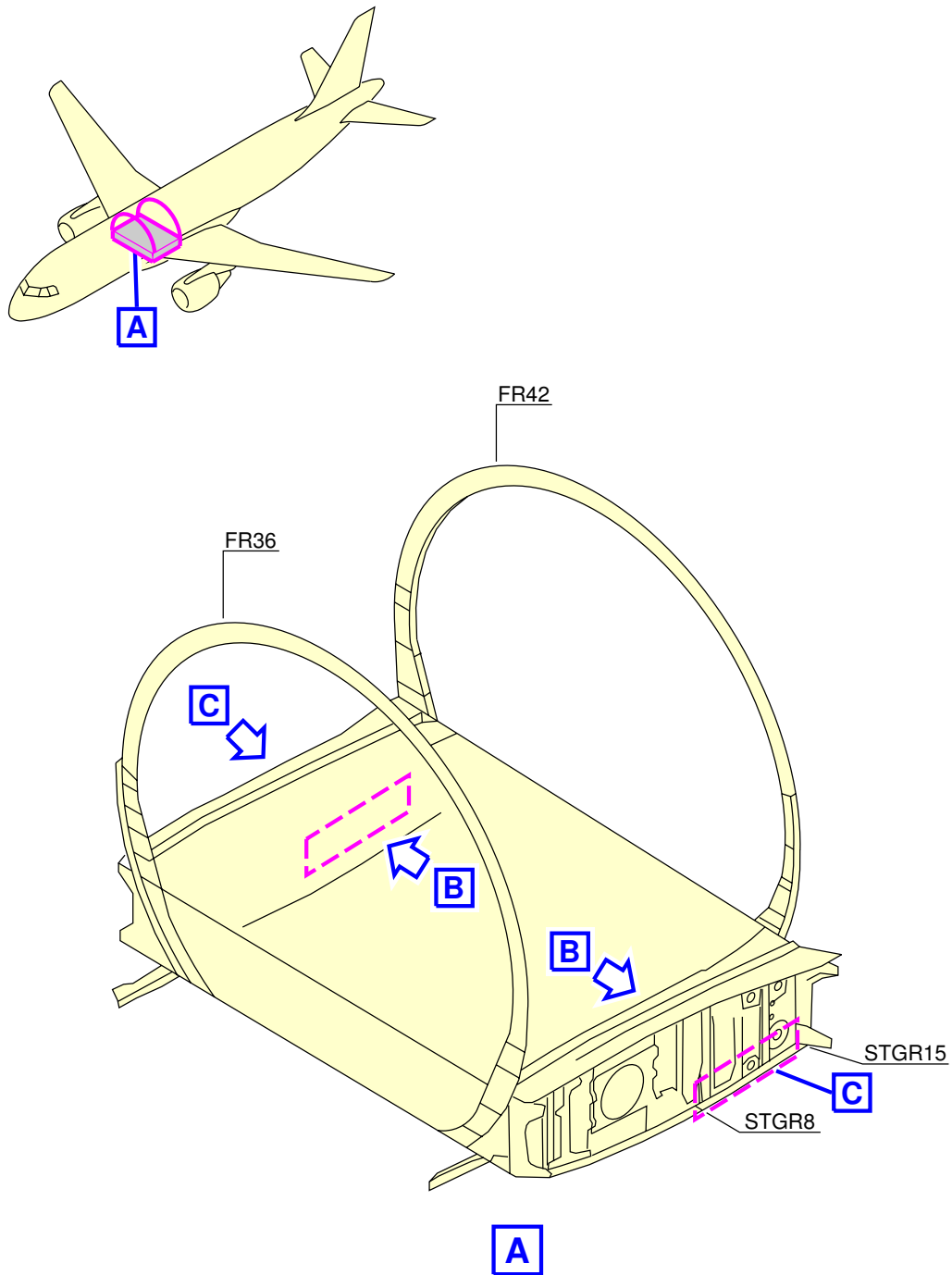
****CONF ALL**



N_SB_571140_5_DDAA_02_01

Figure A-FDDAA - Sheet 02
Removal/Installation of the Fuel Strainer

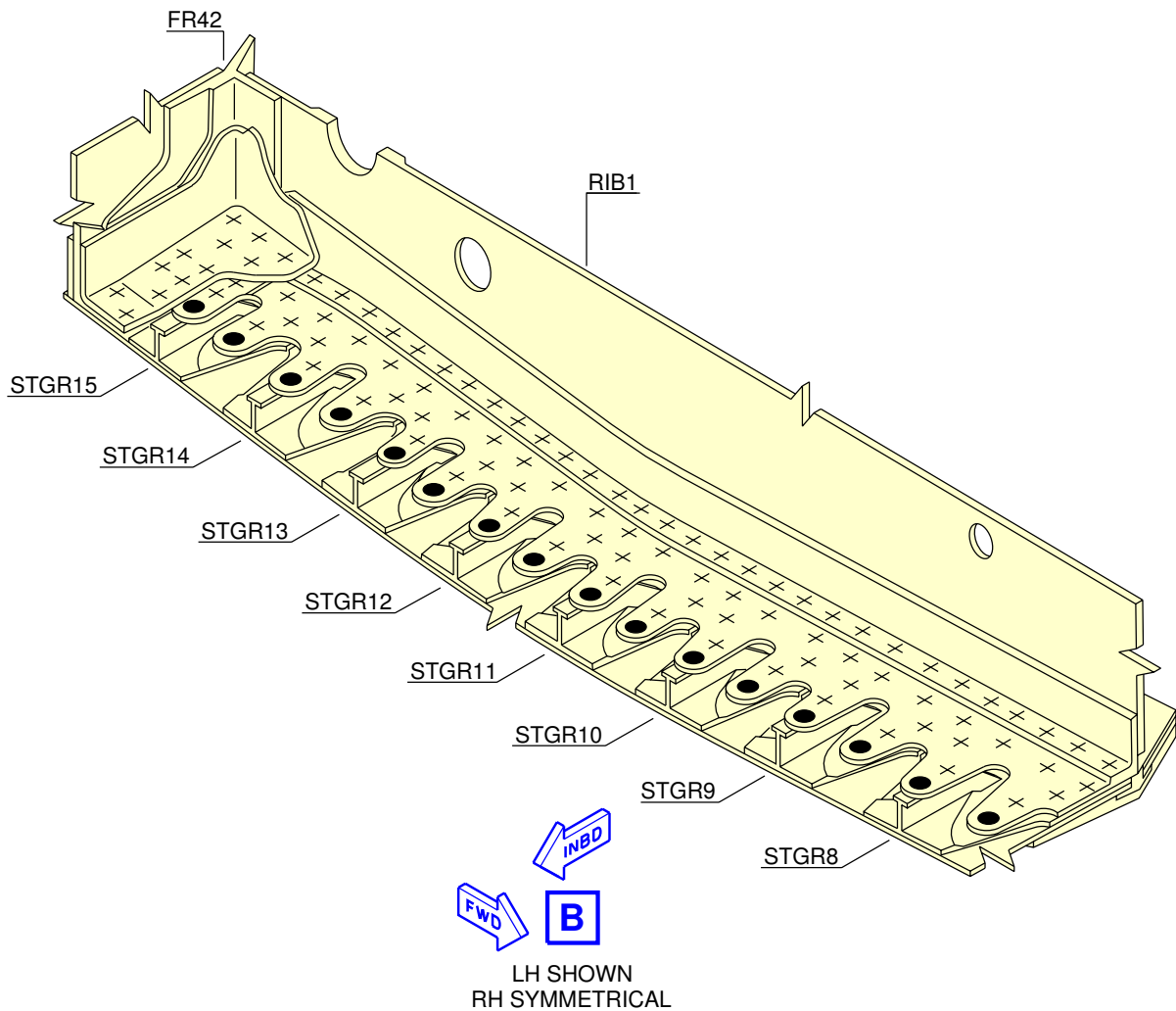
****CONF 001**



N_SB_571140_5_CEEA_01_00

Figure A-FCEAA - Sheet 01
Inspection of the Fasteners for the ADDITIONAL WORK

****CONF 001**



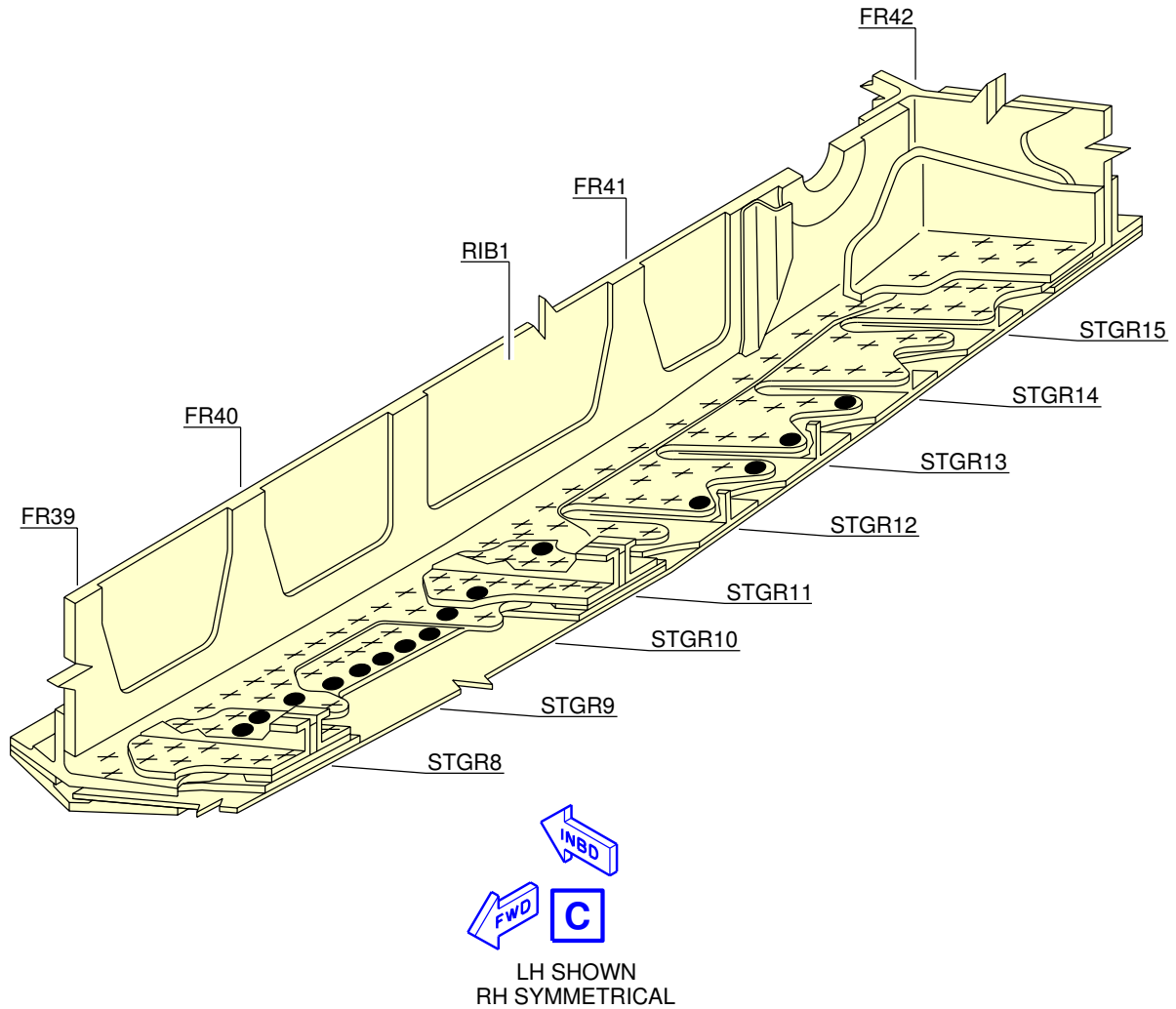
NOTE:

- + FASTENER NOT AFFECTED.
- FASTENER TO BE REMOVED.

N_SB_571140_5_CEEA_02_01

Figure A-FCEAA - Sheet 02
Inspection of the Fasteners for the ADDITIONAL WORK

****CONF 001**



NOTE:

- + FASTENER NOT AFFECTED.
- FASTENER TO BE REMOVED.

N_SB_571140_5_CEEA_03_01

Figure A-FCEAA - Sheet 03
Inspection of the Fasteners for the ADDITIONAL WORK

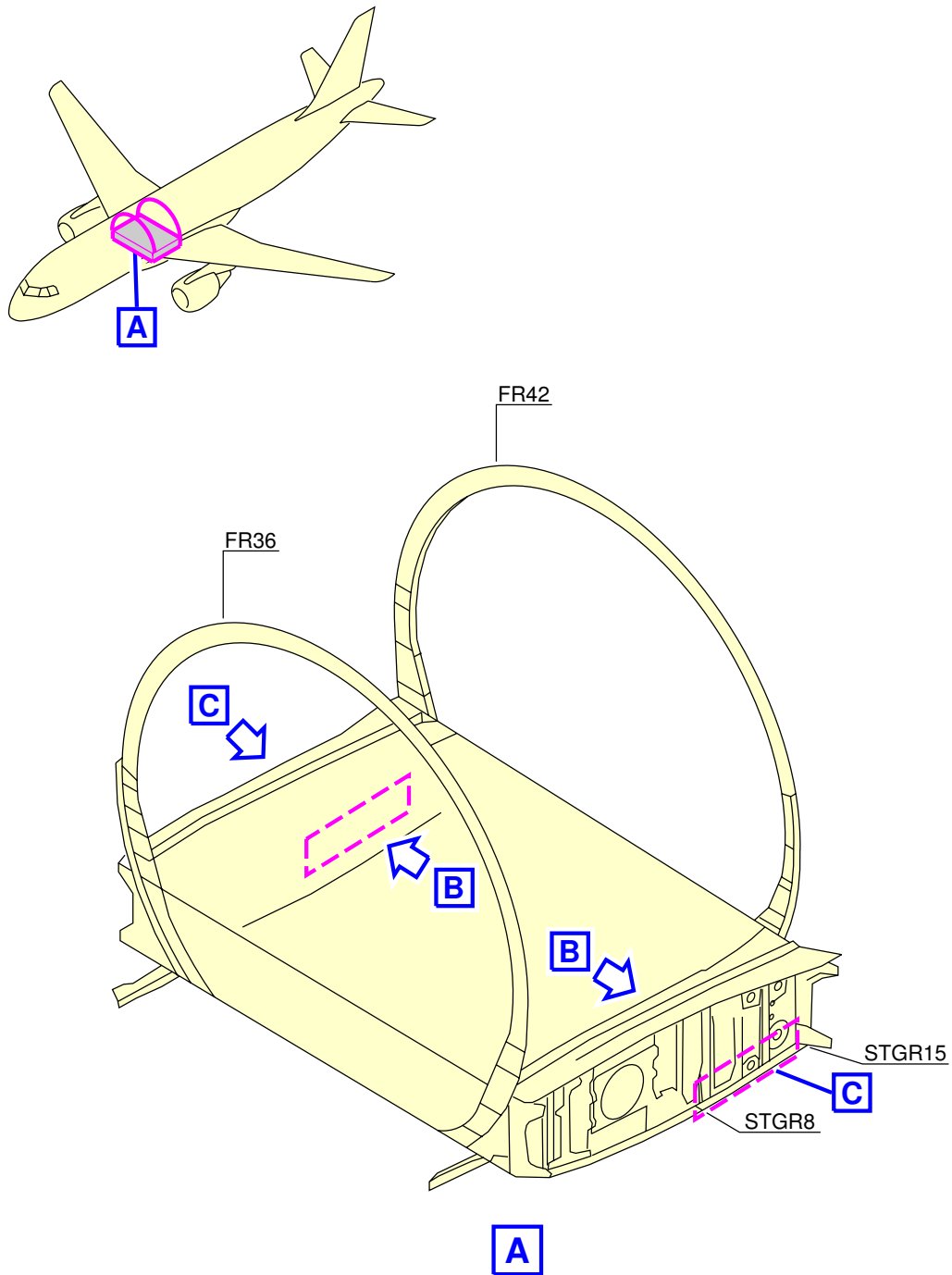
5 DATE: Nov 21/06

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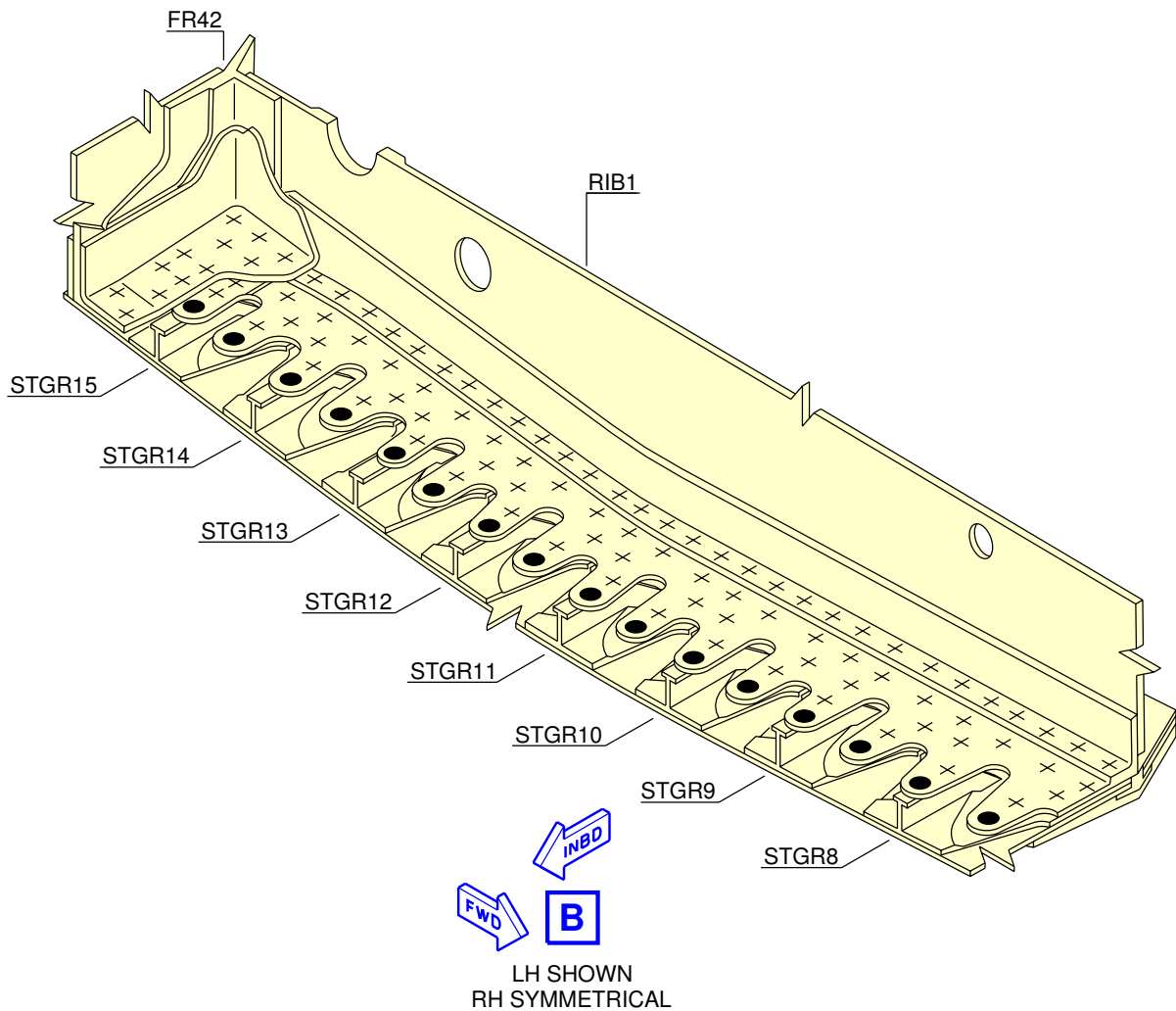
****CONF 002**



N_SB_571140_5_CEAB_01_00

Figure A-FCEAB - Sheet 01
Inspection of the Fasteners for the ADDITIONAL WORK

****CONF 002**



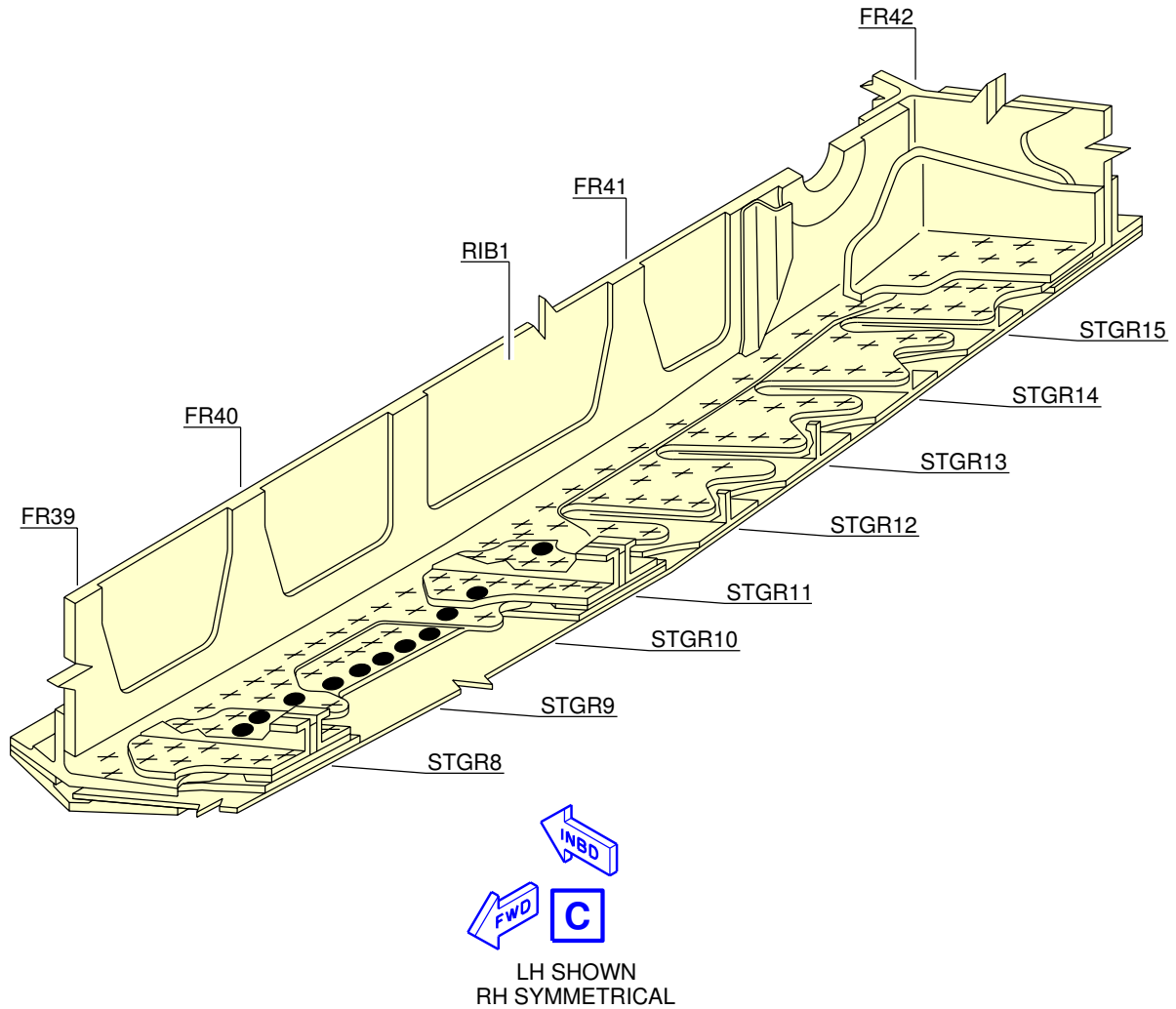
NOTE:

- + FASTENER NOT AFFECTED.
- FASTENER TO BE REMOVED.



N_SB_571140_5_CEAB_02_01

Figure A-FCEAB - Sheet 02
Inspection of the Fasteners for the ADDITIONAL WORK

****CONF 002**



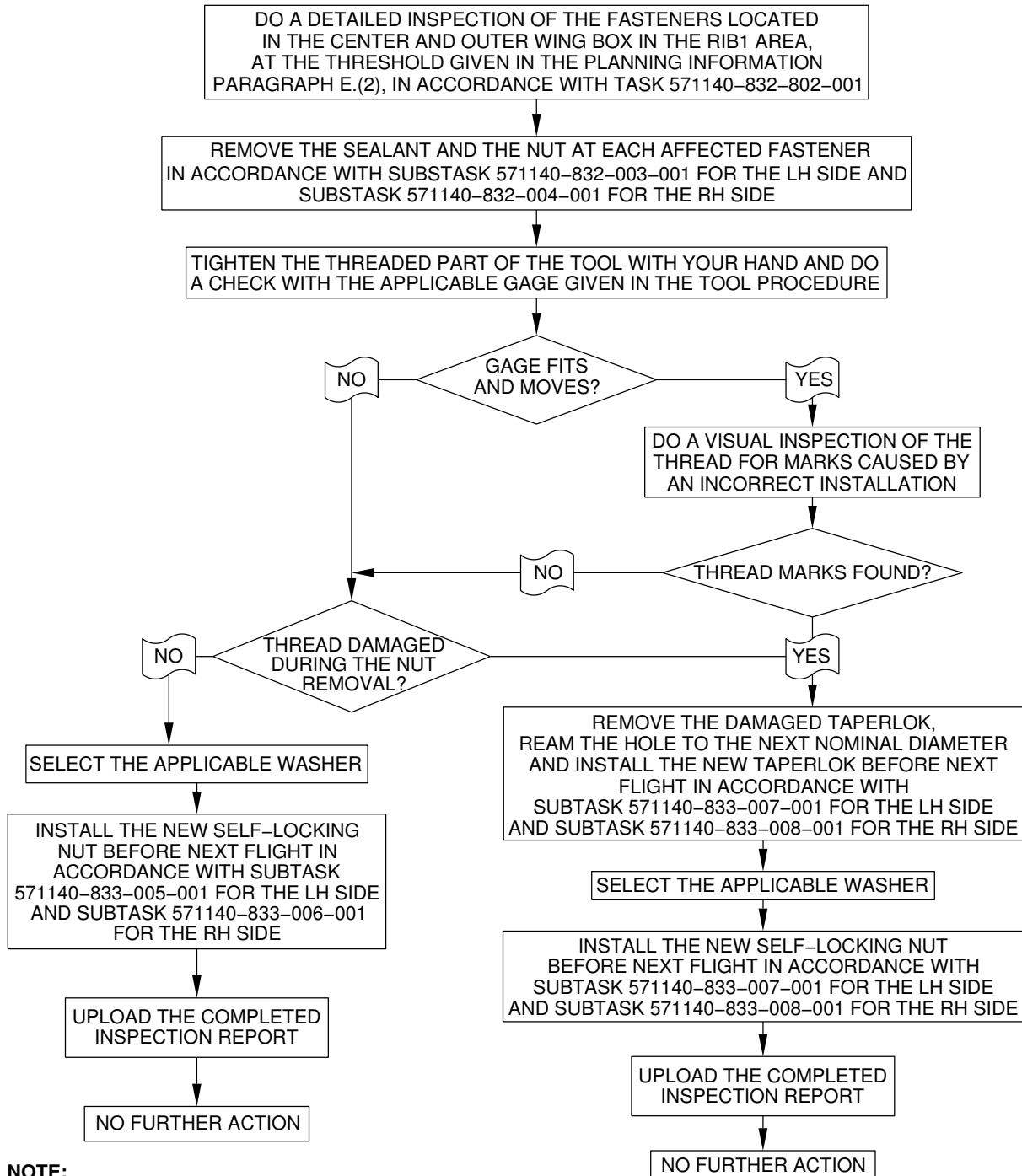
NOTE:

-  FASTENER NOT AFFECTED.
-  FASTENER TO BE REMOVED.

N_SB_571140_5_CEAB_03_01

Figure A-FCEAB - Sheet 03
Inspection of the Fasteners for the ADDITIONAL WORK

****CONF 001**



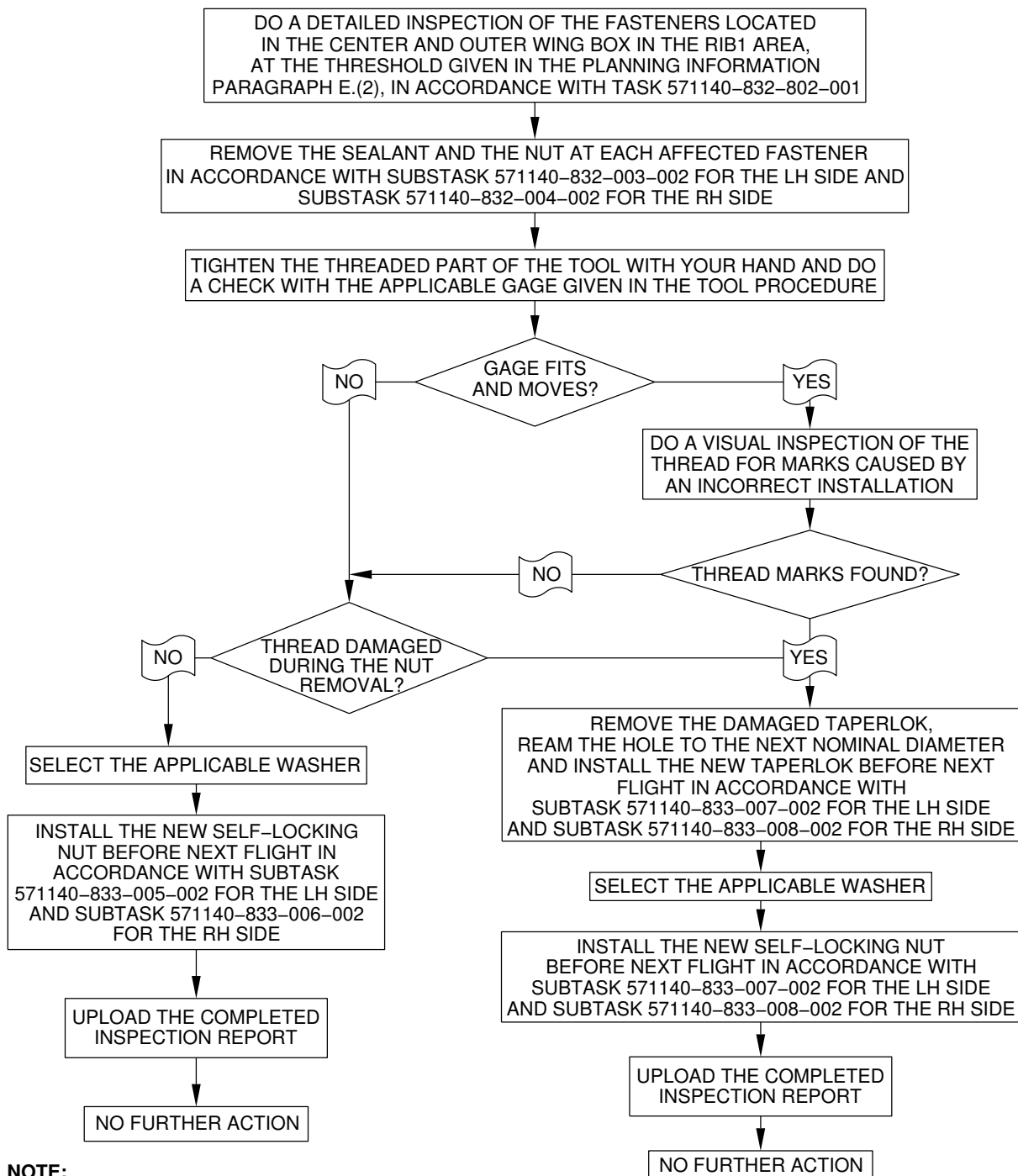
NOTE:

THE PURPOSE OF FLOWCHARTS IS TO SUPPLEMENT THE INFORMATION GIVEN IN THE PROCEDURE AND COMPLIANCE PARAGRAPHS AND NOT TO SERVE AS THE PRIMARY SOURCE FOR TASKS OR COMPLIANCE TIMES GIVEN IN THIS SERVICE BULLETIN.

N_SB_571140_5_FBAA_01_04

Figure A-FFBAA - Sheet 01
Flow Chart for the ADDITIONAL WORK

****CONF 002**



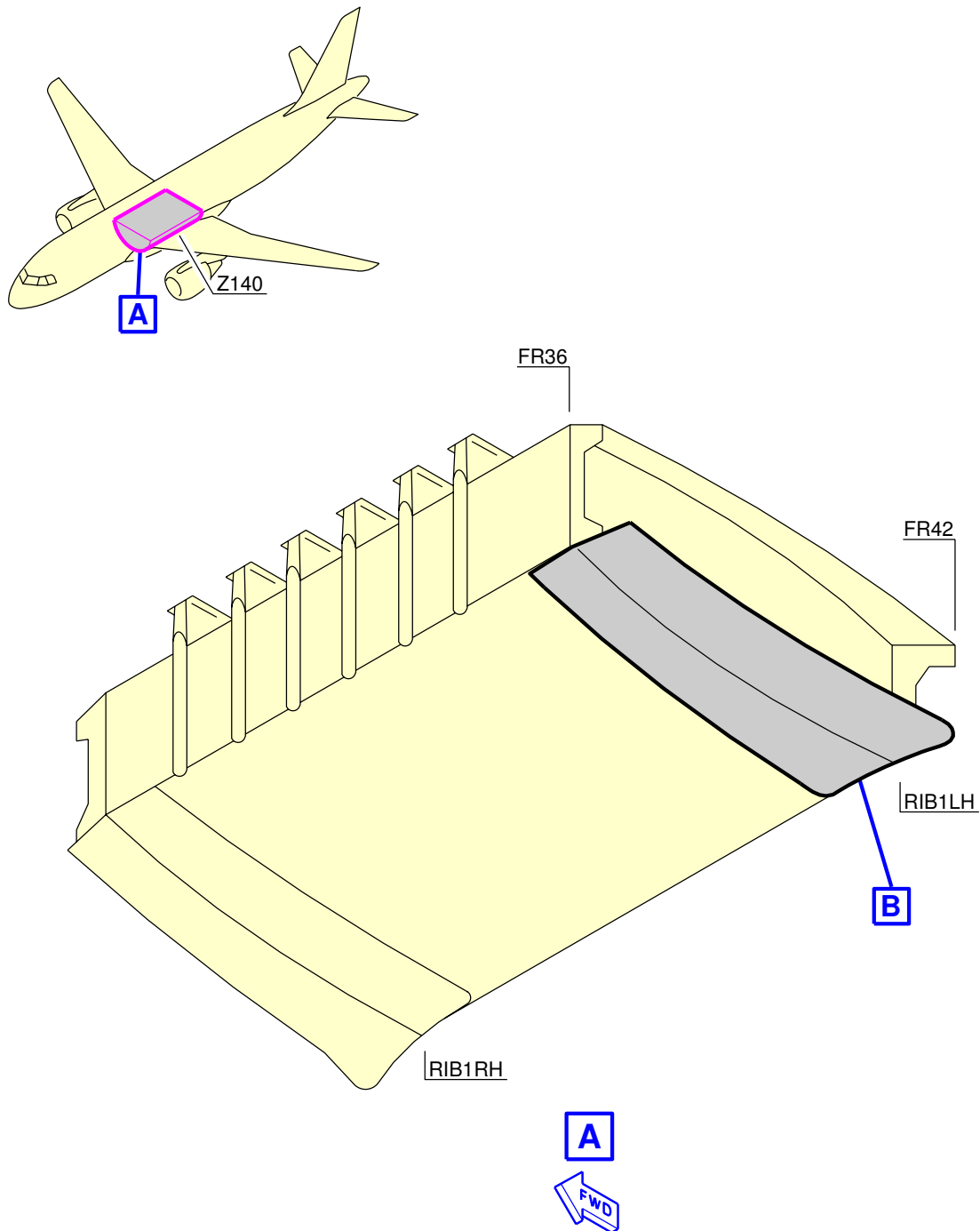
NOTE:

THE PURPOSE OF FLOWCHARTS IS TO SUPPLEMENT THE INFORMATION GIVEN IN THE PROCEDURE AND COMPLIANCE PARAGRAPHS AND NOT TO SERVE AS THE PRIMARY SOURCE FOR TASKS OR COMPLIANCE TIMES GIVEN IN THIS SERVICE BULLETIN.

N_SB_571140_5_FBAB_01_04

Figure A-FFBAB - Sheet 01
Flow Chart for the ADDITIONAL WORK

****CONF 001**



N_SB_571140_5_CFAA_01_01

Figure A-FCFAA - Sheet 01
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

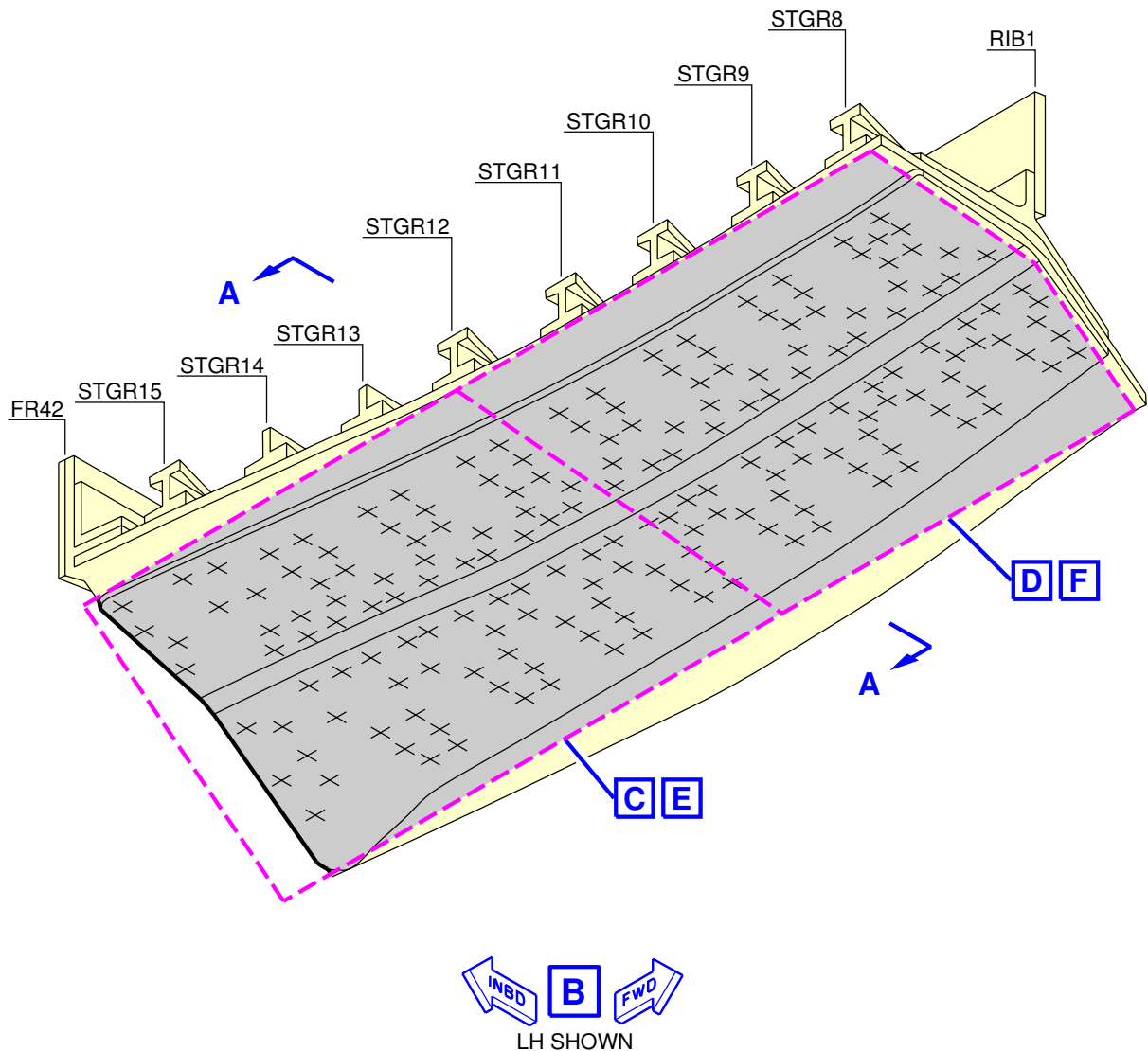
5 DATE: Nov 21/06

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****CONF 001**



NOTE:

+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CFAA_02_01

Figure A-FCFAA - Sheet 02
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

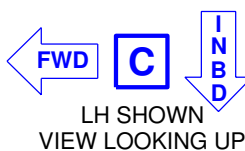
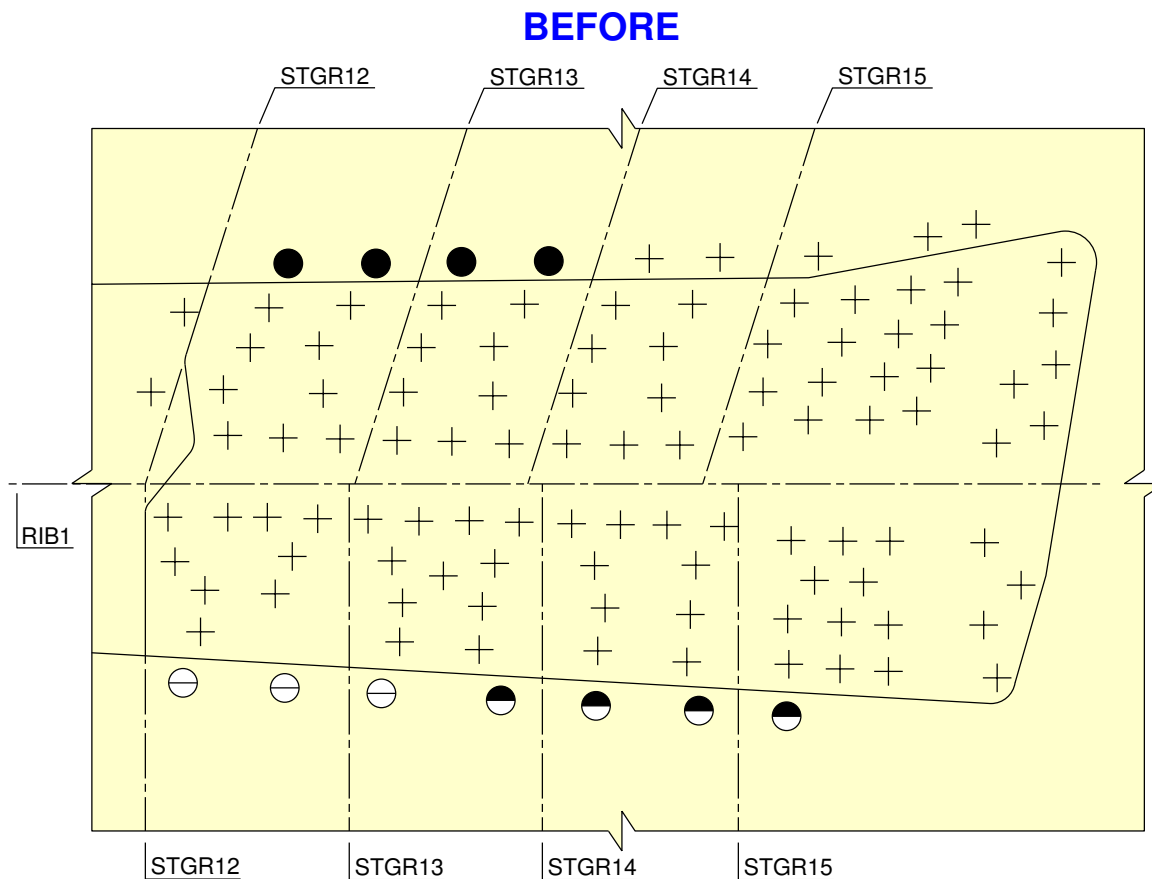
5 DATE: Nov 21/06

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****CONF 001**



HOLE	OLD ITEM
	(10) OR ⁽³⁾ ₍₁₀₎
	(13) OR ⁽⁷²⁾ ₍₁₃₎
	(13) OR ⁽⁷¹⁾ ₍₁₃₎

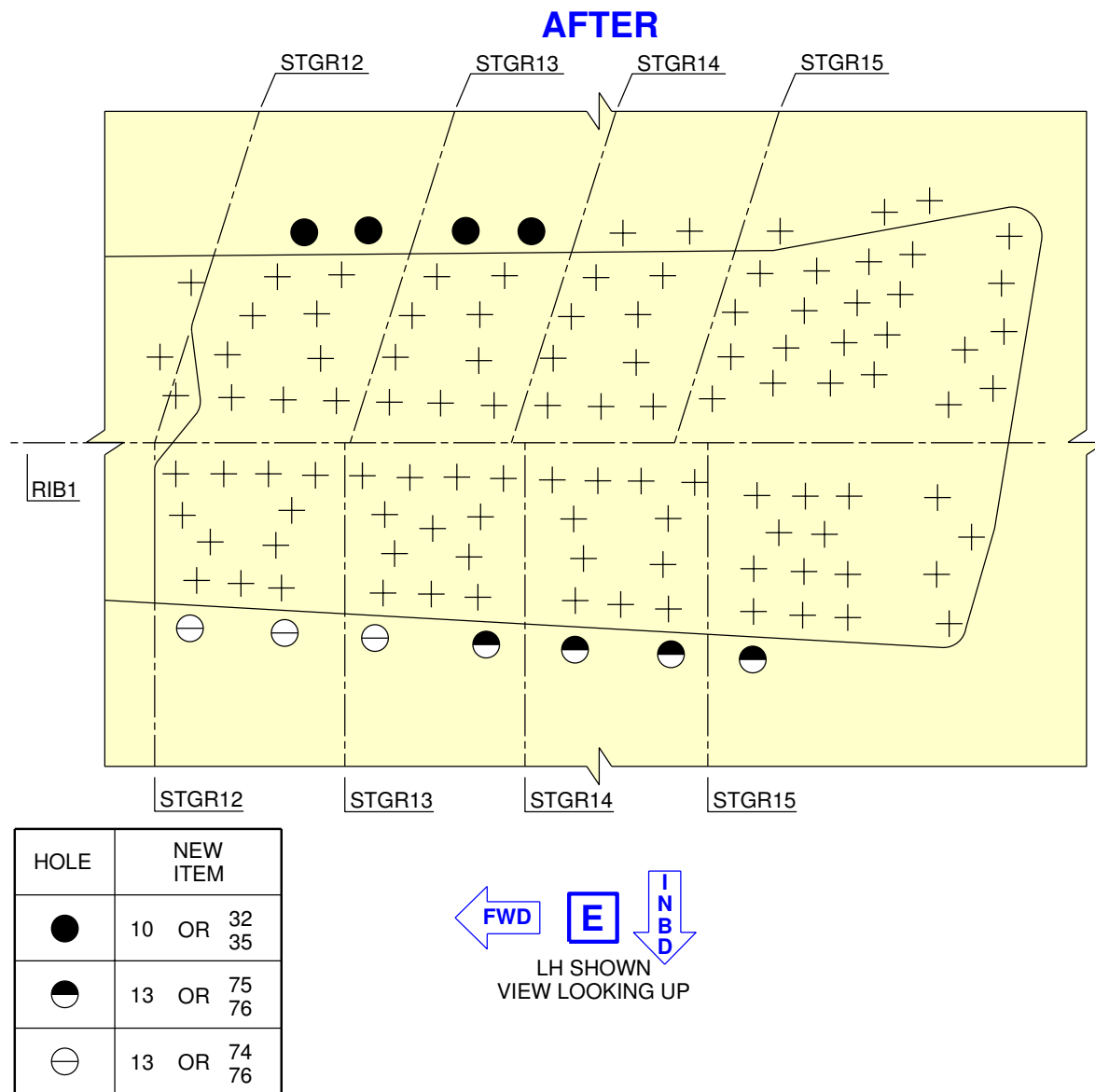
NOTE:

+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CFAA_03_01

Figure A-FCFAA - Sheet 03
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 001**



NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

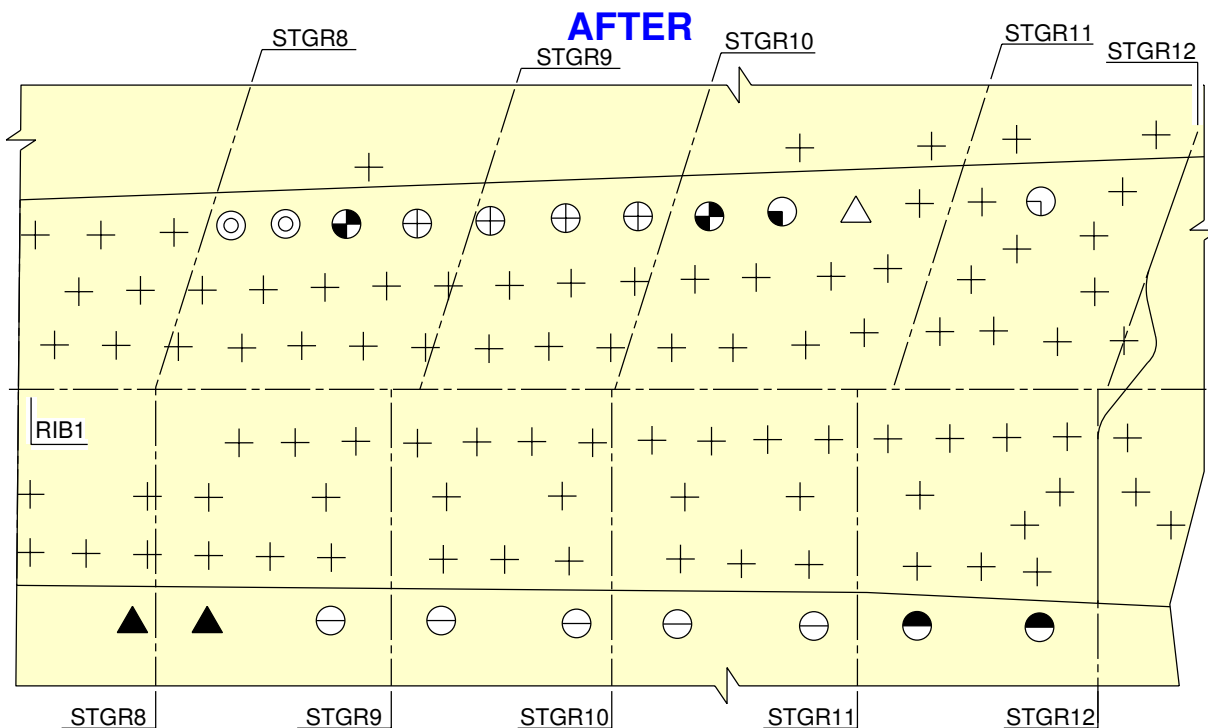
IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

N_SB_571140_5_CFAA_05_01

Figure A-FCFAA - Sheet 05
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 001**



NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

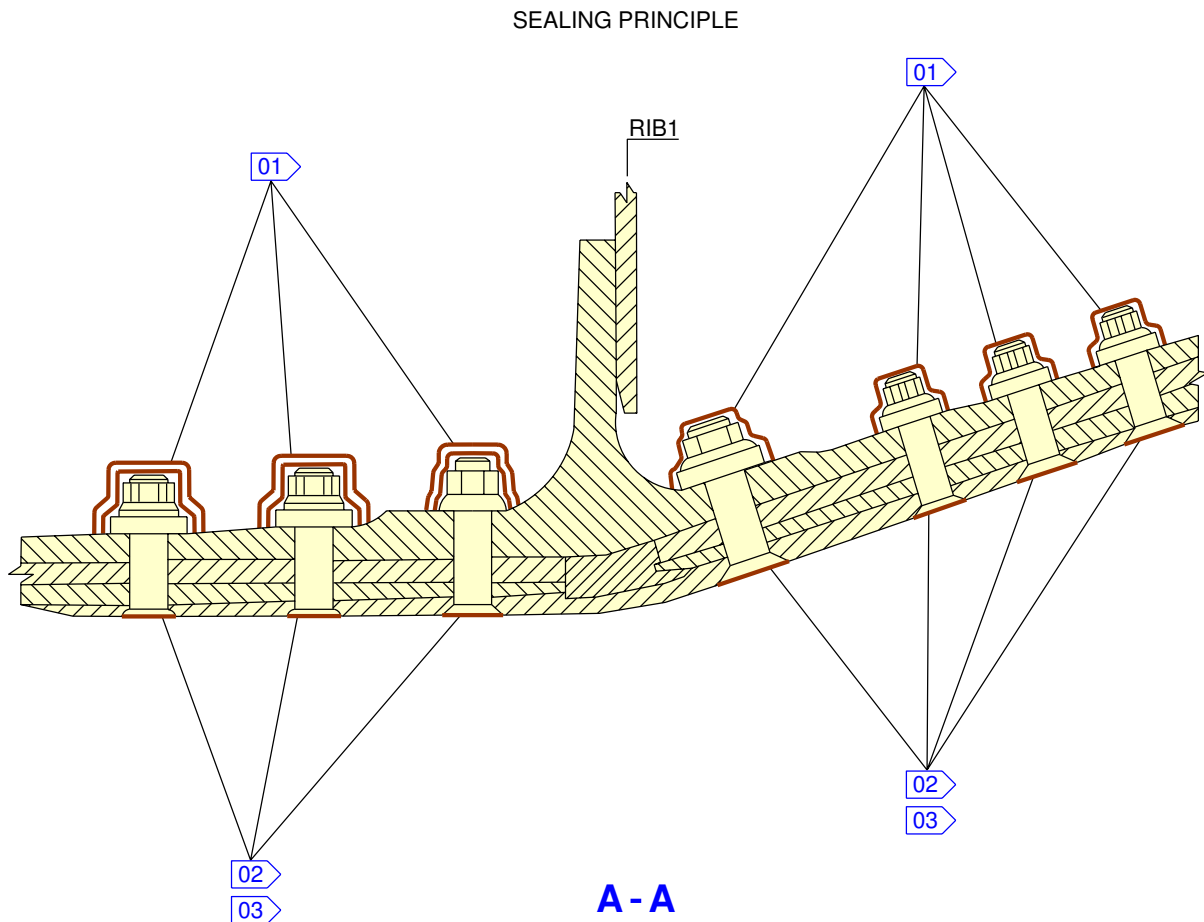
IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

HOLE	NEW ITEM
⊕	10 OR 34 35
⊕•	10 OR 78 35
⊕△	10 OR 82 35
⊕□	11 OR 37 44
▲	11 OR 43 44
⊕○	13 OR 73 76
⊕○•	13 OR 75 76
⊕○△	13 OR 74 76
⊕⊕	15 OR 80 11

N_SB_571140_5_CFAA_06_01

Figure A-FCFAA - Sheet 06
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 001**



NOTE:

- 01** APPLY MATERIAL No 06ABB1 AND MATERIAL No 06PAG1 ON THE NUTS.
- 02** APPLY MATERIAL No 04EAC2, MATERIAL No 04CMA2 AND MATERIAL No 04JAA3 ON THE BOLTS.
- 03** WET ASSEMBLY, APPLY MATERIAL No 06ABB1.

N_SB_571140_5_CFAA_07_01

Figure A-FCFAA - Sheet 07
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

A yellow airplane is shown from a side profile. A pink rectangular area is highlighted on the fuselage, with a blue box containing the letter 'A' pointing to it. The text 'Z140' is written next to the pink area.

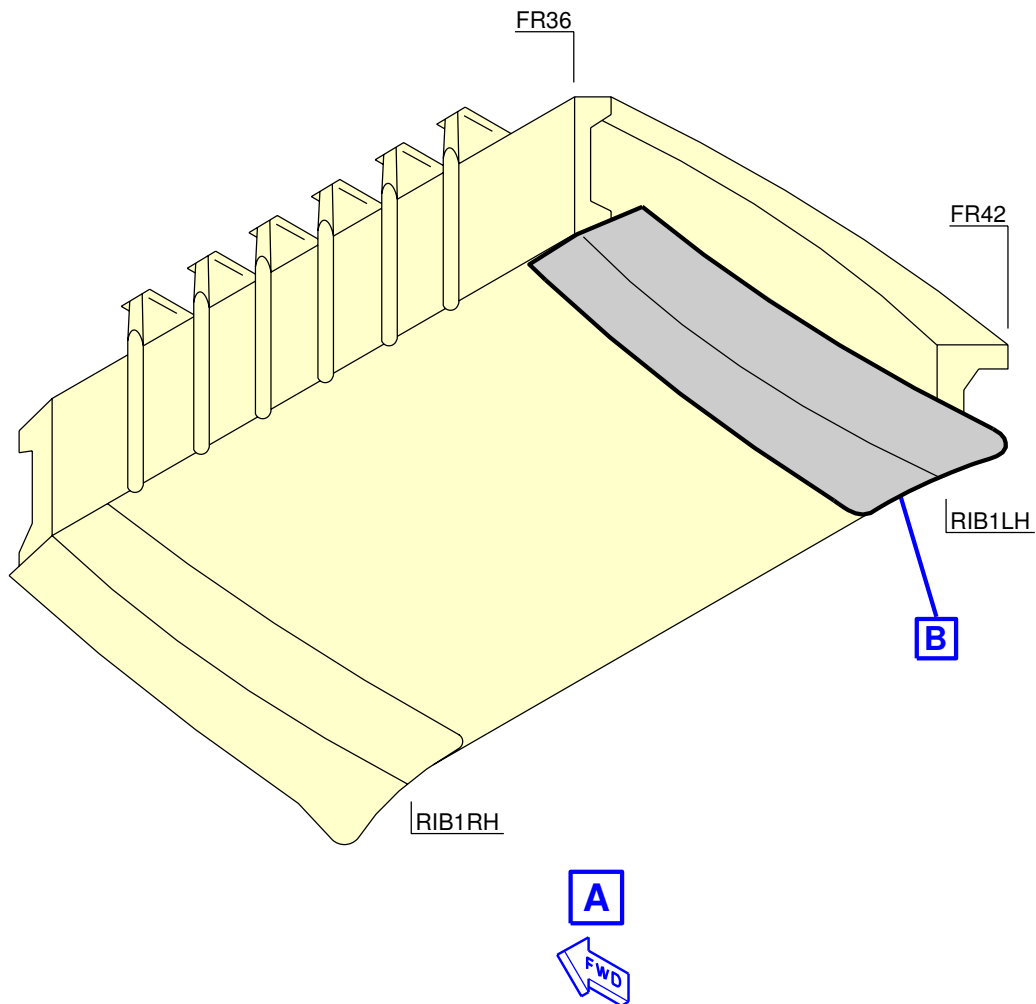
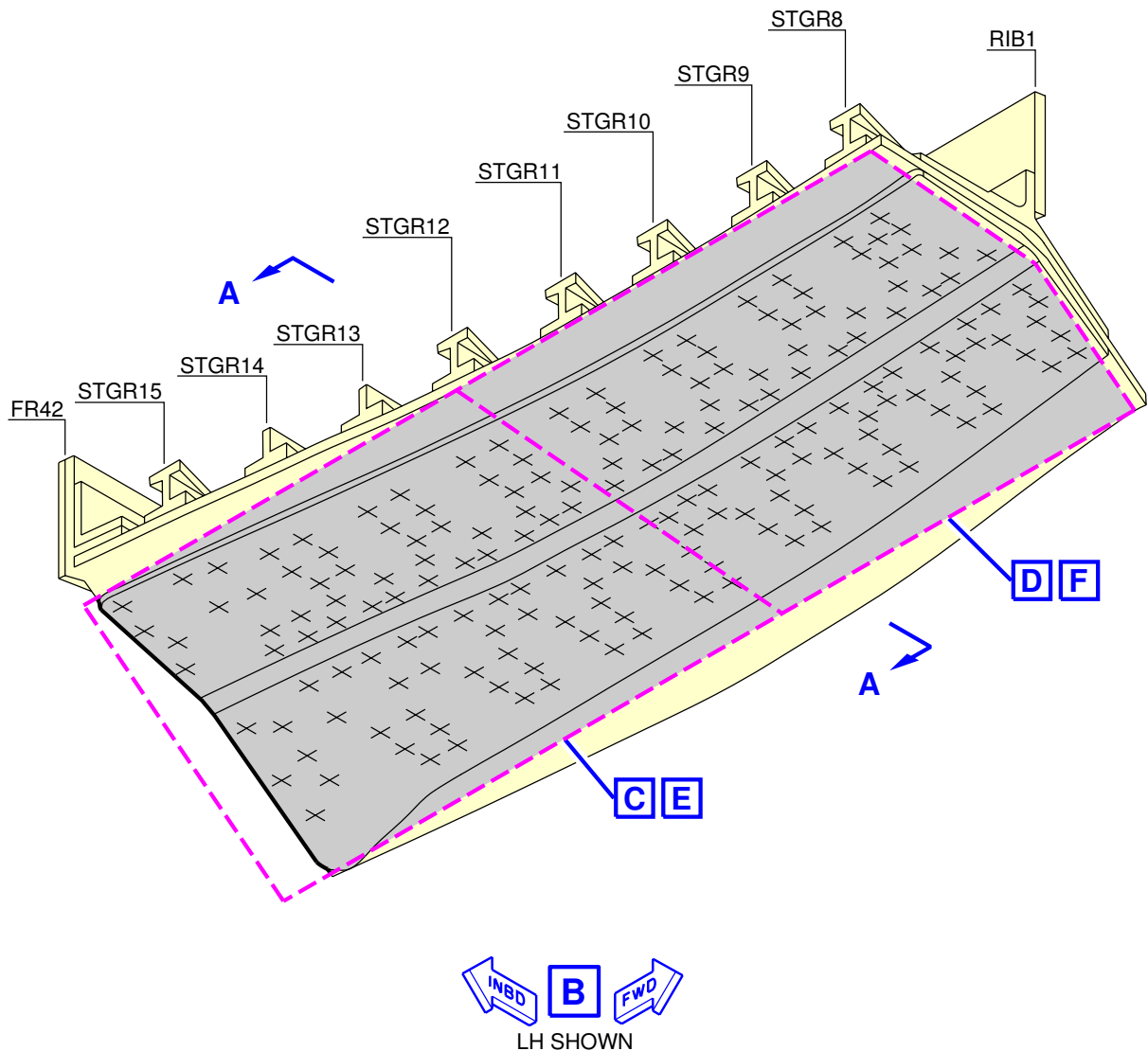


Figure A-FCFAB - Sheet 01
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 002**



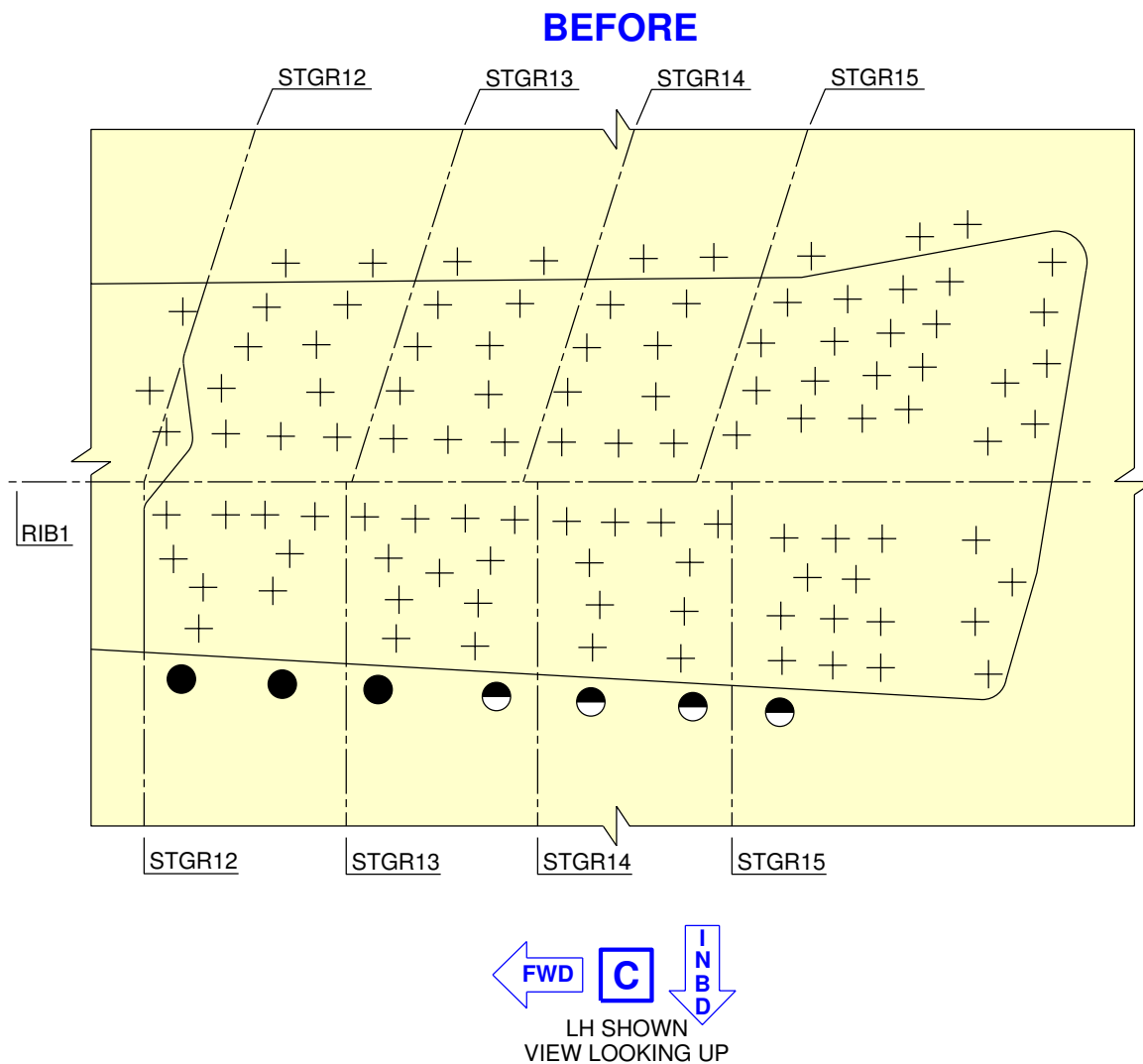
NOTE:



+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CFAB_02_01

Figure A-FCFAB - Sheet 02
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 002**



HOLE	OLD ITEM
	(13) OR (72) (13)
	(13) OR (71) (13)

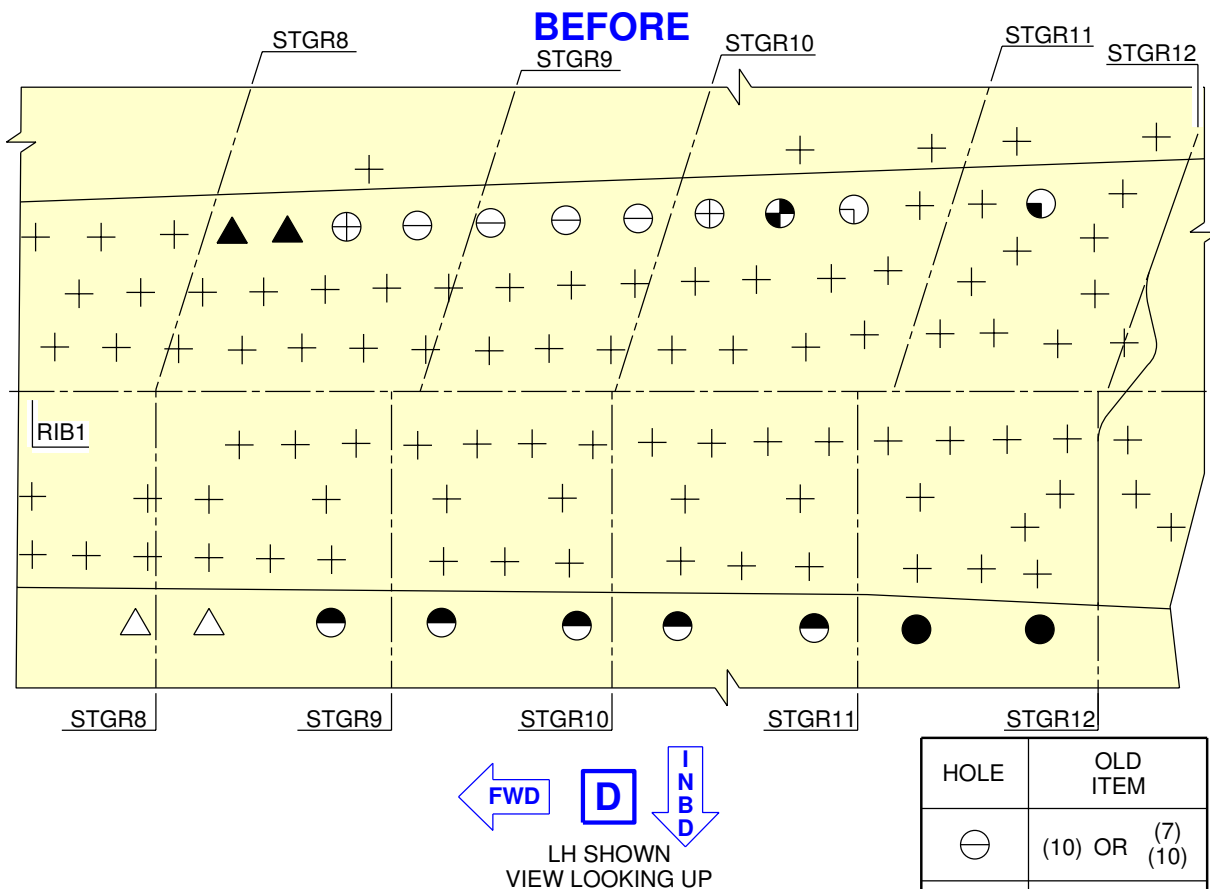
NOTE:










+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CFAB_03_01

Figure A-FCFAB - Sheet 03
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 002**



HOLE	OLD ITEM
	(10) OR (7) (10)
	(10) OR (77) (10)
	(10) OR (81) (10)
	(11) OR (14) (11)
	(11) OR (12) (11)
	(13) OR (70) (13)
	(13) OR (72) (13)
	(13) OR (71) (13)
	(15) OR (79) (15)

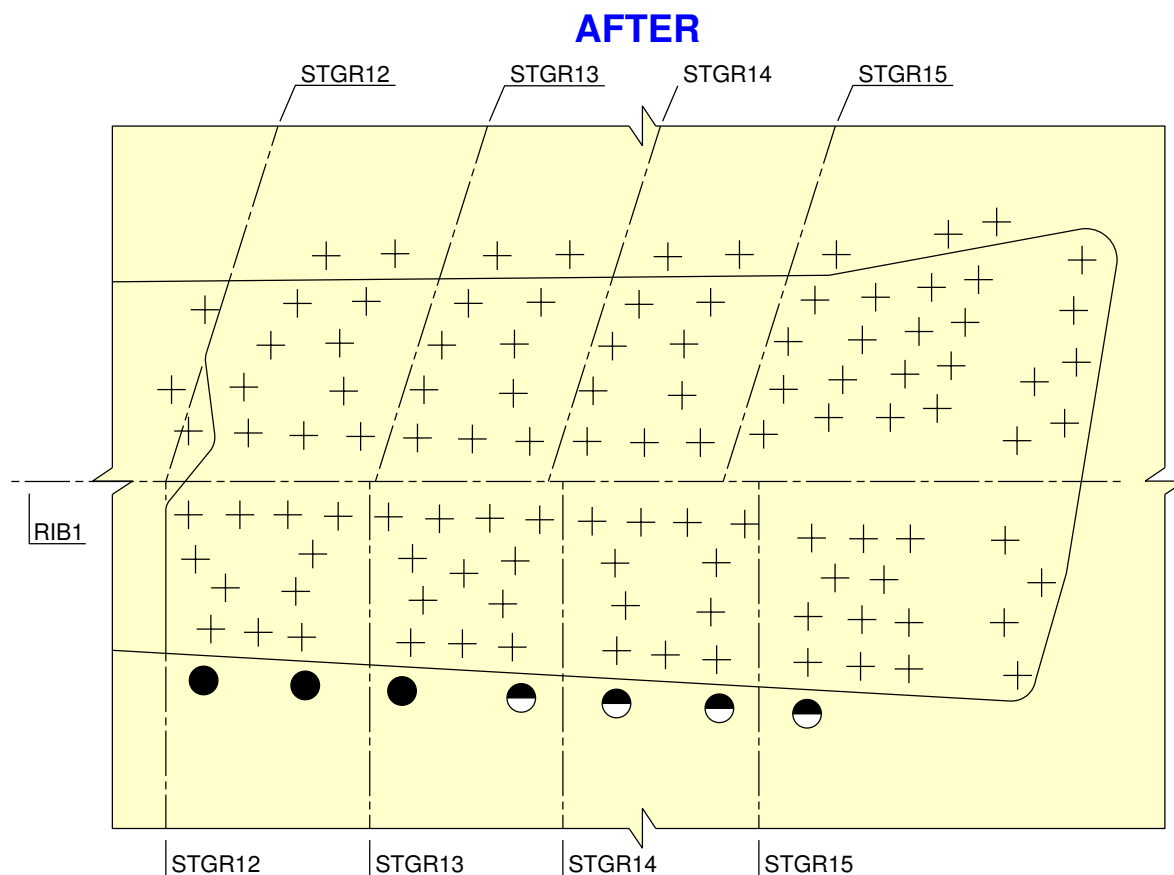
NOTE:



+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CFAB_04_01

Figure A-FCFAB - Sheet 04
 LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 002**



HOLE	NEW ITEM
	13 OR 75 76
	13 OR 74 76



NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

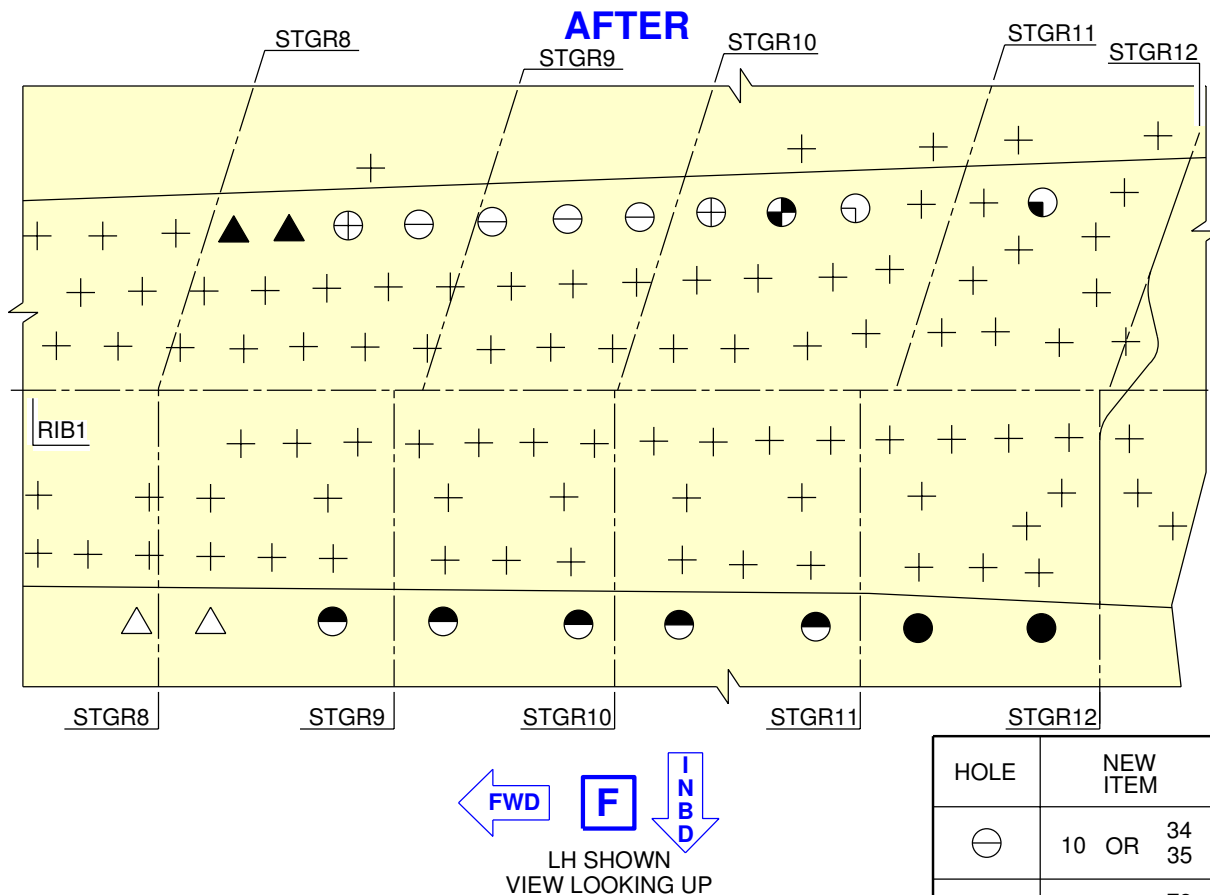
IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

N_SB_571140_5_CFAB_05_01

Figure A-FCFAB - Sheet 05
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 002**












NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

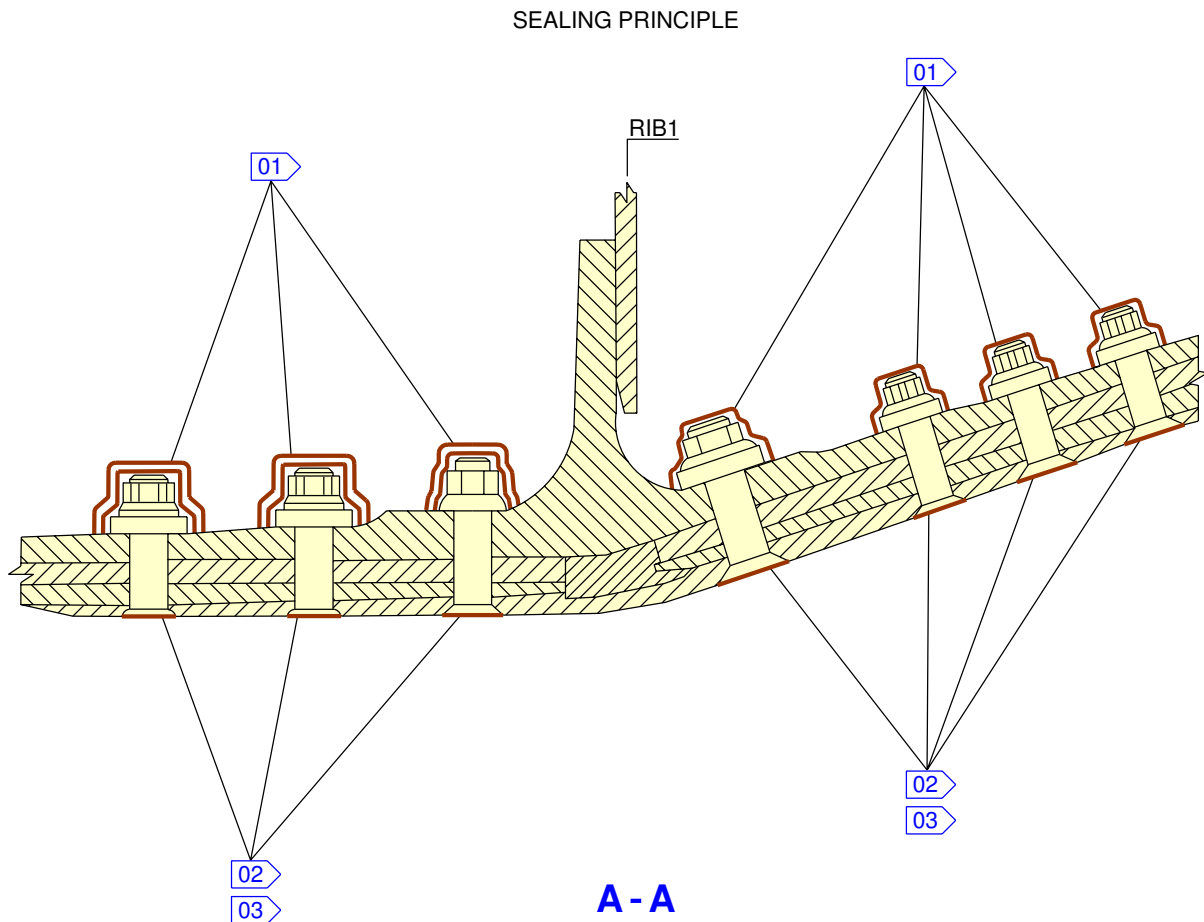
IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

HOLE	NEW ITEM	
	10 OR	34 35
	10 OR	78 35
	10 OR	82 35
	11 OR	37 44
	11 OR	43 44
	13 OR	73 76
	13 OR	75 76
	13 OR	74 76
	15 OR	80 11

N_SB_571140_5_CFAB_06_01

Figure A-FCFAB - Sheet 06
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 002**



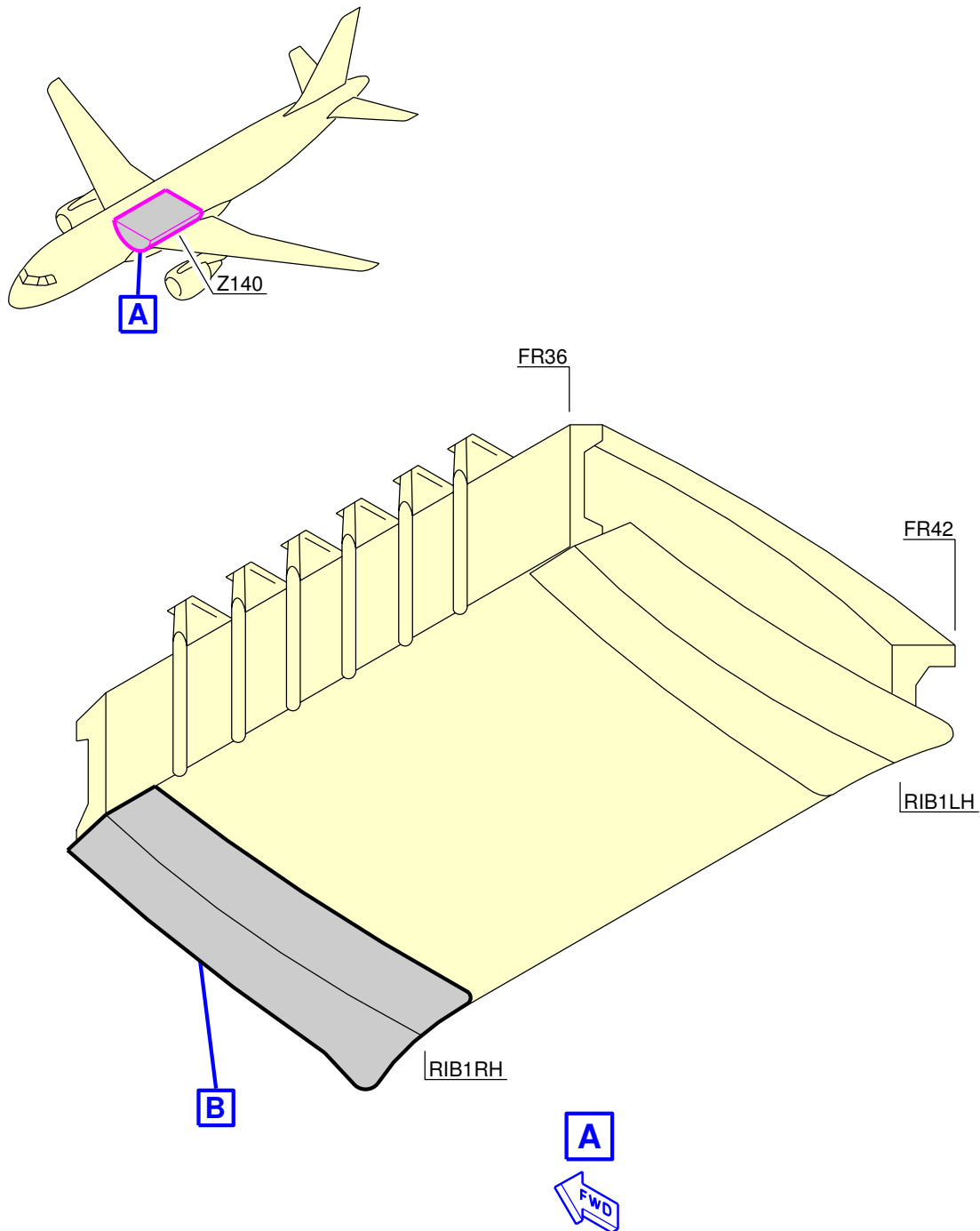
NOTE:

- 01** APPLY MATERIAL No 06ABB1 AND MATERIAL No 06PAG1 ON THE NUTS.
- 02** APPLY MATERIAL No 04EAC2, MATERIAL No 04CMA2 AND MATERIAL No 04JAA3 ON THE BOLTS.
- 03** WET ASSEMBLY, APPLY MATERIAL No 06ABB1.

N_SB_571140_5_CFAB_07_01

Figure A-FCFAB - Sheet 07
LH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 001**



N_SB_571140_5_CGAA_01_01

Figure A-FCGAA - Sheet 01
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

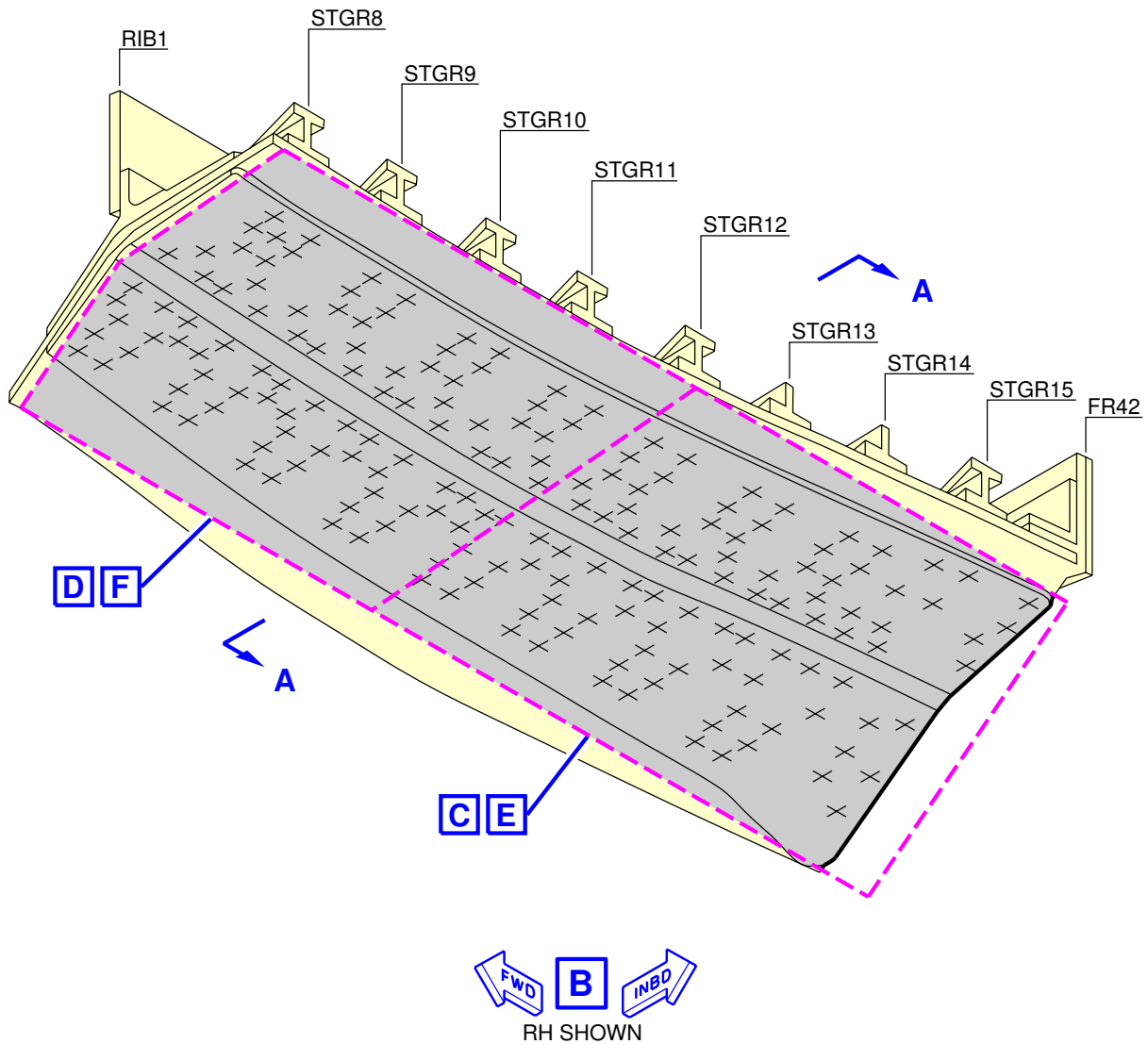
5 DATE: Nov 21/06

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NOTE:

+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CGAA_02_01

Figure A-FCGAA - Sheet 02
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

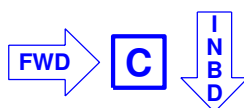
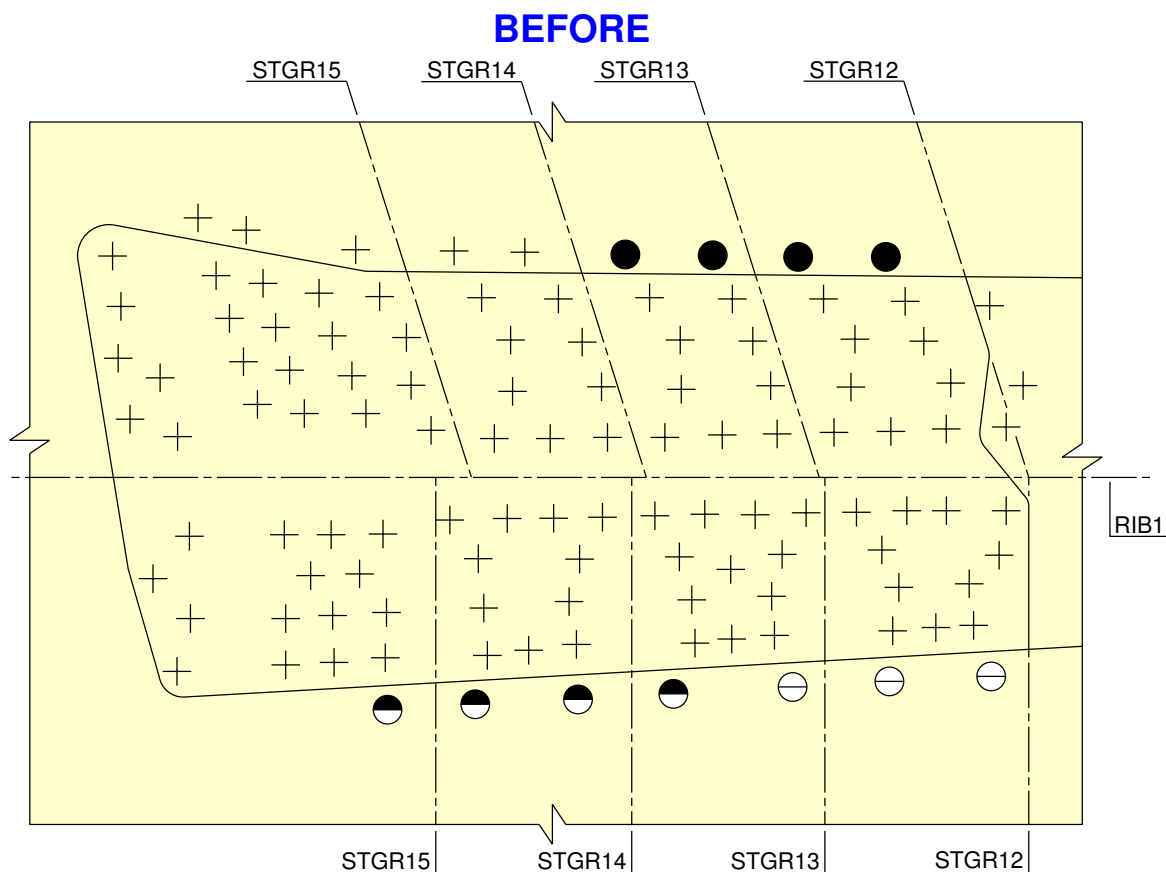
5 DATE: Nov 21/06

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****CONF 001**



RH SHOWN
VIEW LOOKING UP

HOLE	OLD ITEM
	(10) OR ⁽³⁾ ₍₁₀₎
	(13) OR ⁽⁷²⁾ ₍₁₃₎
	(13) OR ⁽⁷¹⁾ ₍₁₃₎

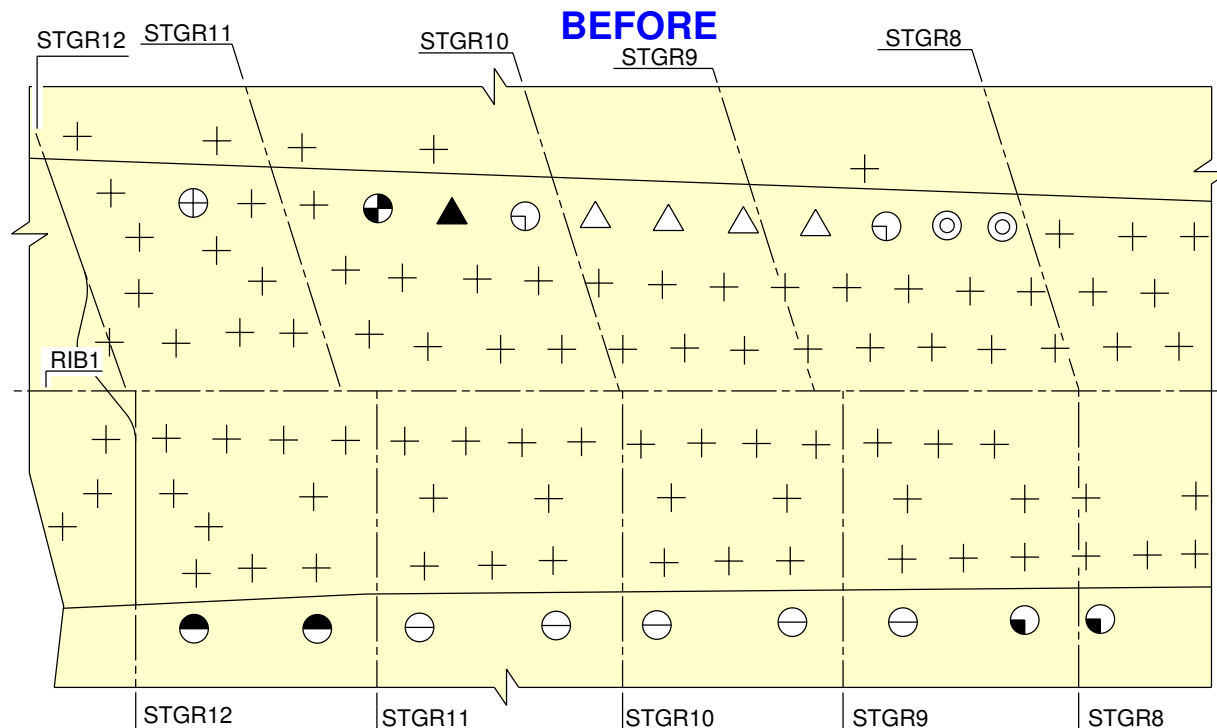
NOTE:

+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CGAA_03_01

Figure A-FCGAA - Sheet 03
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 001**



HOLE	OLD ITEM
⊕	(11) OR (14) (11)
⊙	(11) OR (12) (11)
⊖	(13) OR (72) (13)
⊗	(13) OR (71) (13)
⊘	(13) OR (70) (13)
⊙	(10) OR (77) (10)
Δ	(10) OR (7) (10)
▲	(10) OR (81) (10)
⊙	(13) OR (79) (15)

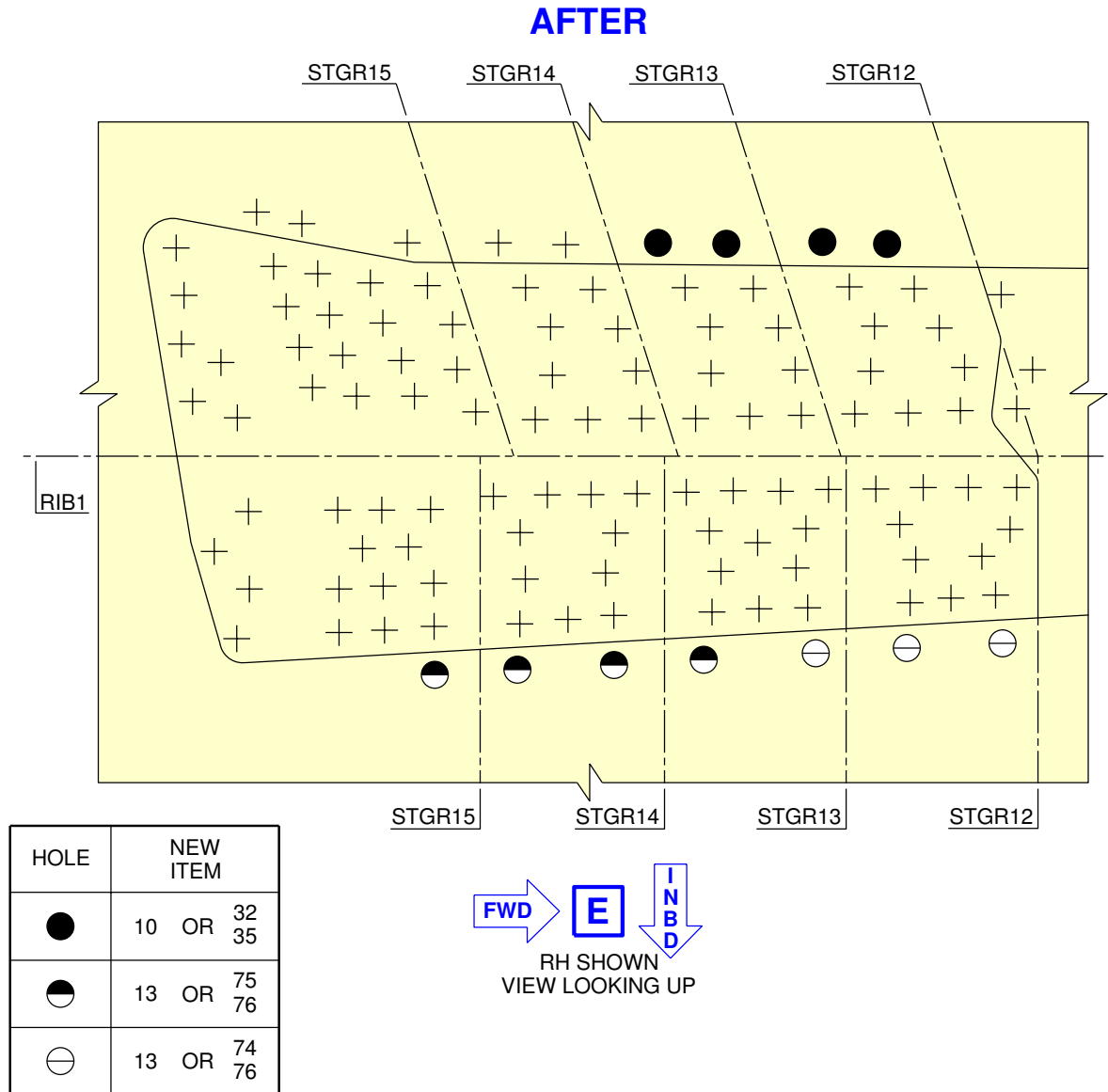
NOTE:

+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CGAA_04_01

Figure A-FCGAA - Sheet 04
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 001**



NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

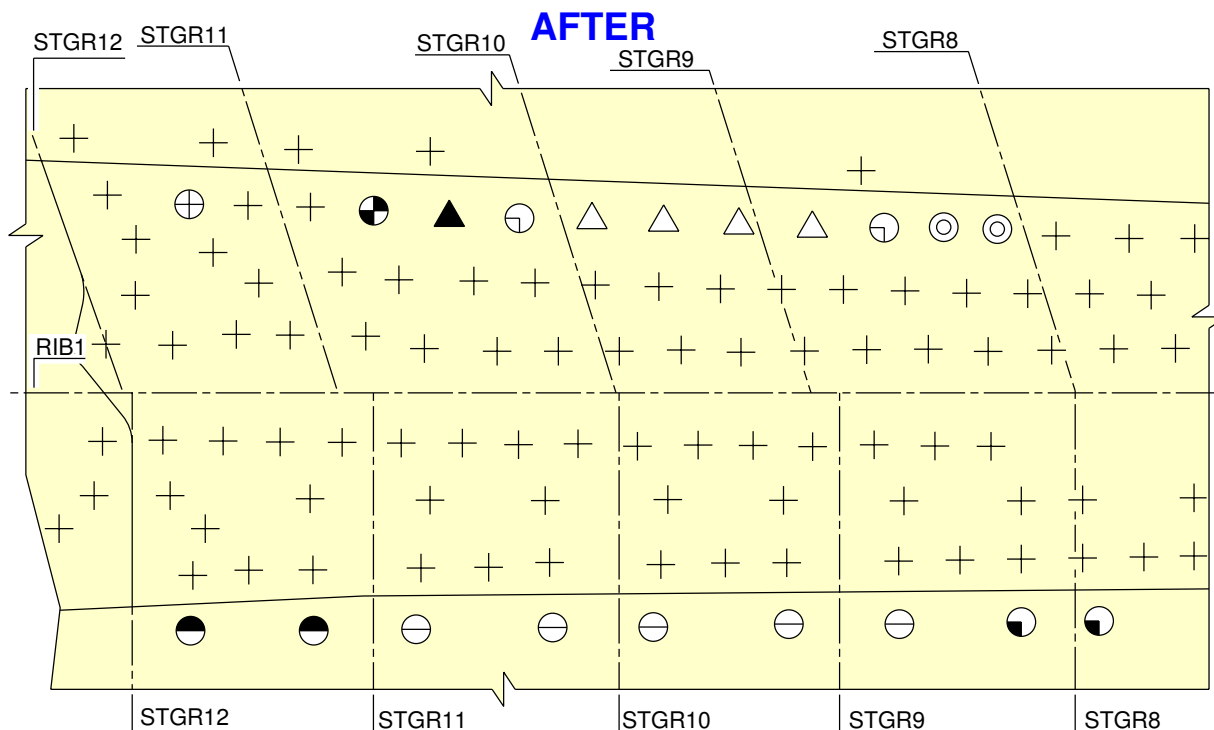
IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

N_SB_571140_5_CGAA_05_01

Figure A-FCGAA - Sheet 05
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 001**



HOLE	NEW ITEM	
	11 OR	37 44
	11 OR	43 44
	13 OR	75 76
	13 OR	74 76
	13 OR	73 76
	10 OR	78 35
	10 OR	34 35
	10 OR	82 35
	15 OR	80 11

NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

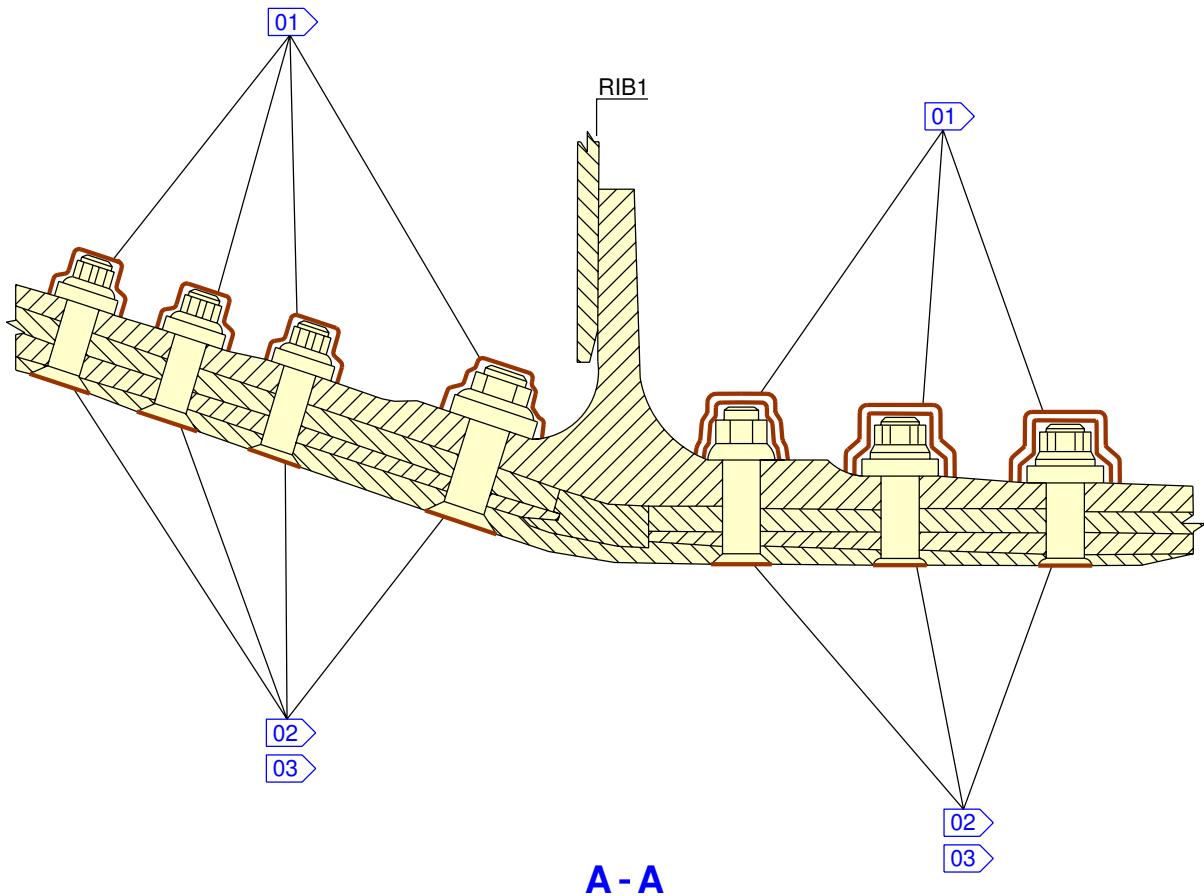
IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

N_SB_571140_5_CGAA_06_01

Figure A-FCGAA - Sheet 06
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 001**

SEALING PRINCIPLE



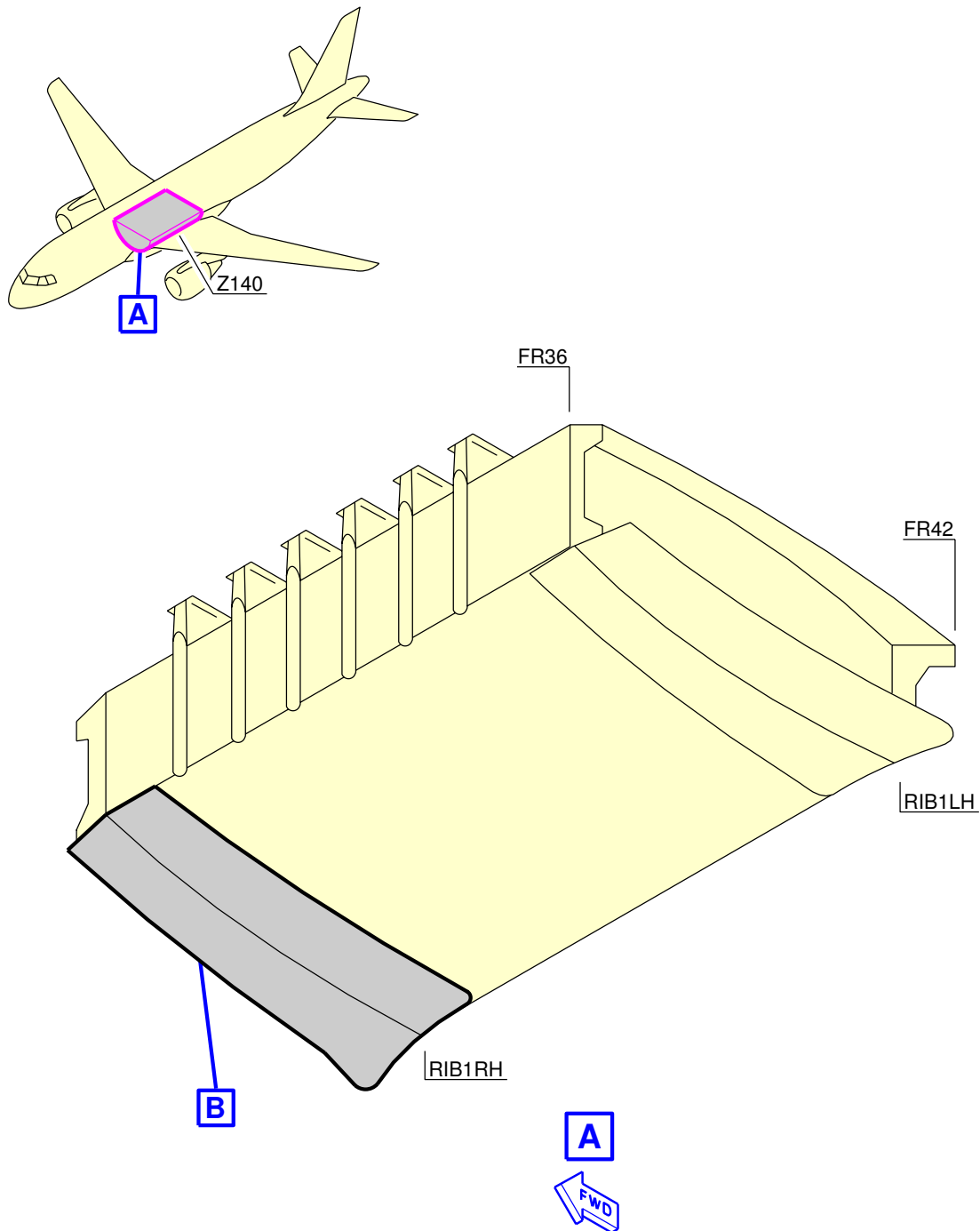
NOTE:

- 01** APPLY MATERIAL No 06AAB1 AND MATERIAL No 06PAG1 ON THE NUTS.
- 02** APPLY MATERIAL No 04EAC2, MATERIAL No 04CAA2 AND MATERIAL No 04JAA3 ON THE BOLTS.
- 03** WET ASSEMBLY, APPLY MATERIAL No 06ABB1.

N_SB_571140_5_CGAA_07_01

Figure A-FCGAA - Sheet 07
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 002**



N_SB_571140_5_CGAB_01_01

Figure A-FCGAB - Sheet 01
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

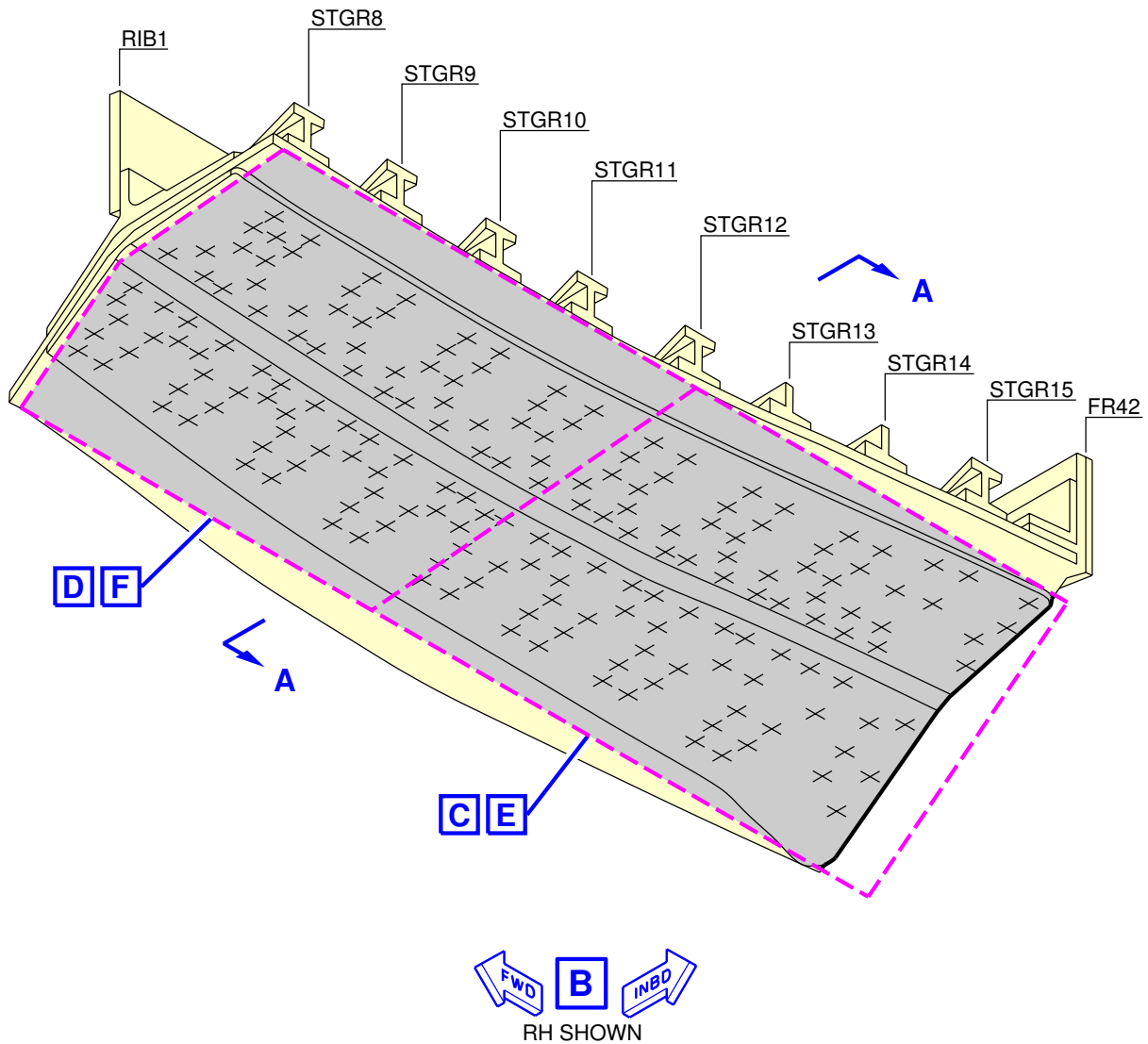
5 DATE: Nov 21/06

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****CONF 002**



NOTE:

+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CGAB_02_01

Figure A-FCGAB - Sheet 02
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

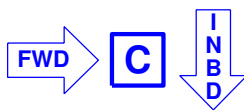
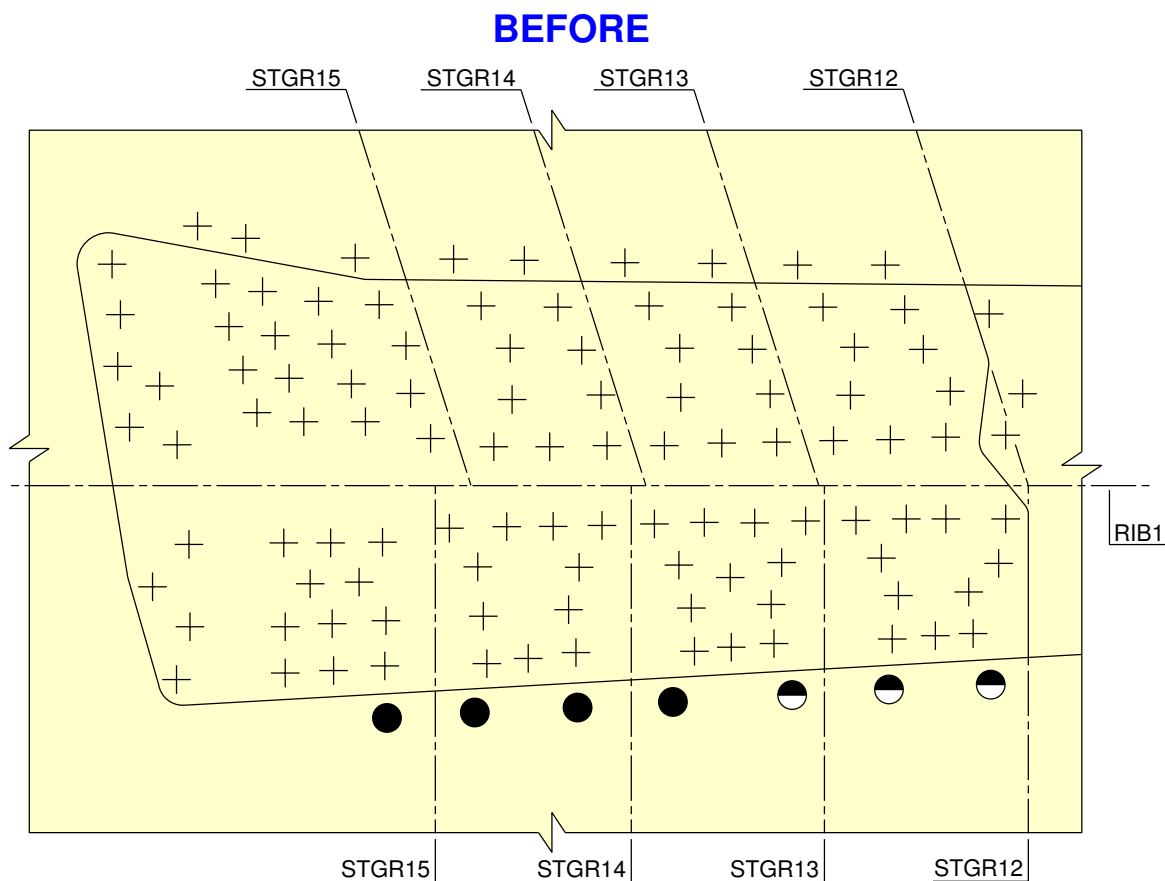
5 DATE: Nov 21/06

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Page: 244

****CONF 002**



RH SHOWN
VIEW LOOKING UP

HOLE	OLD ITEM
	(13) OR (72) (13)
	(13) OR (71) (13)

NOTE:

+ FASTENERS NOT AFFECTED.

N_SB_571140_5_CGAB_03_01










Figure A-FCGAB - Sheet 03
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

SERVICE BULLETIN

The diagram shows a 2D grid with various symbols and labels. The grid is divided into several regions by dashed lines. The top region is labeled "BEFORE" in blue. The bottom region is divided into five columns labeled "STGR12", "STGR11", "STGR10", "STGR9", and "STGR8" from left to right. The leftmost column is labeled "RIB1". The grid contains various symbols: plus signs (+), minus signs (-), circles with plus or minus signs, circles with a triangle inside, circles with a black or white half, and solid black triangles. The grid is also labeled with "STGR12", "STGR11", "STGR10", "STGR9", and "STGR8" at the top and bottom.



 RH SHOWN
 VIEW LOOKING UP

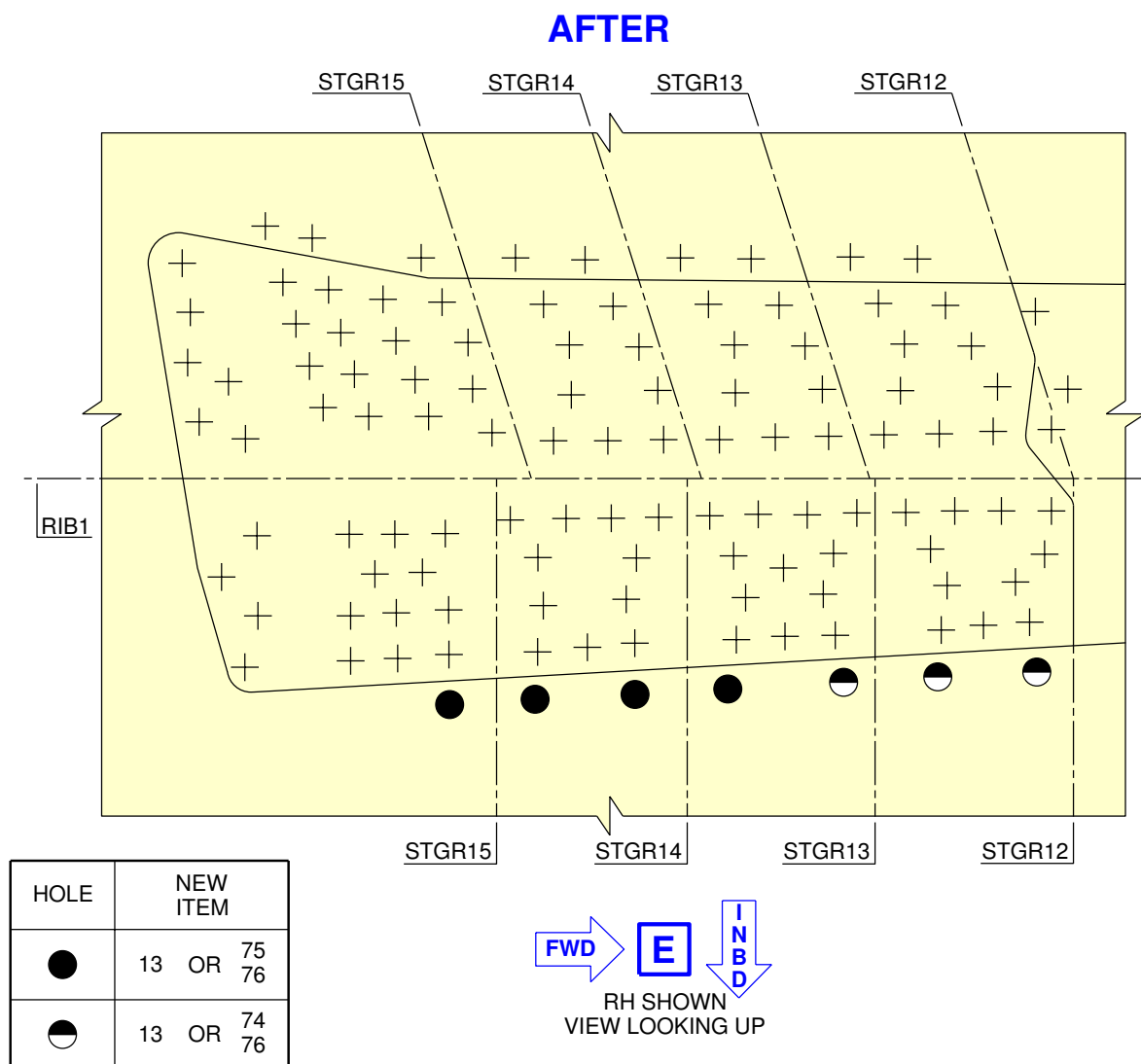
HOLE	OLD ITEM
	(11) OR ⁽¹⁴⁾ ₍₁₁₎
	(11) OR ⁽¹²⁾ ₍₁₁₎
	(13) OR ⁽⁷²⁾ ₍₁₃₎
	(13) OR ⁽⁷¹⁾ ₍₁₃₎
	(13) OR ⁽⁷⁰⁾ ₍₁₃₎
	(10) OR ⁽⁷⁷⁾ ₍₁₀₎
	(10) OR ⁽⁷⁾ ₍₁₀₎
	(10) OR ⁽⁸¹⁾ ₍₁₀₎
	(13) OR ⁽⁷⁹⁾ ₍₁₅₎

✦ FASTENERS NOT AFFECTED.

N_SB_571140_5_CGAB_04_01

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****CONF 002**



NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

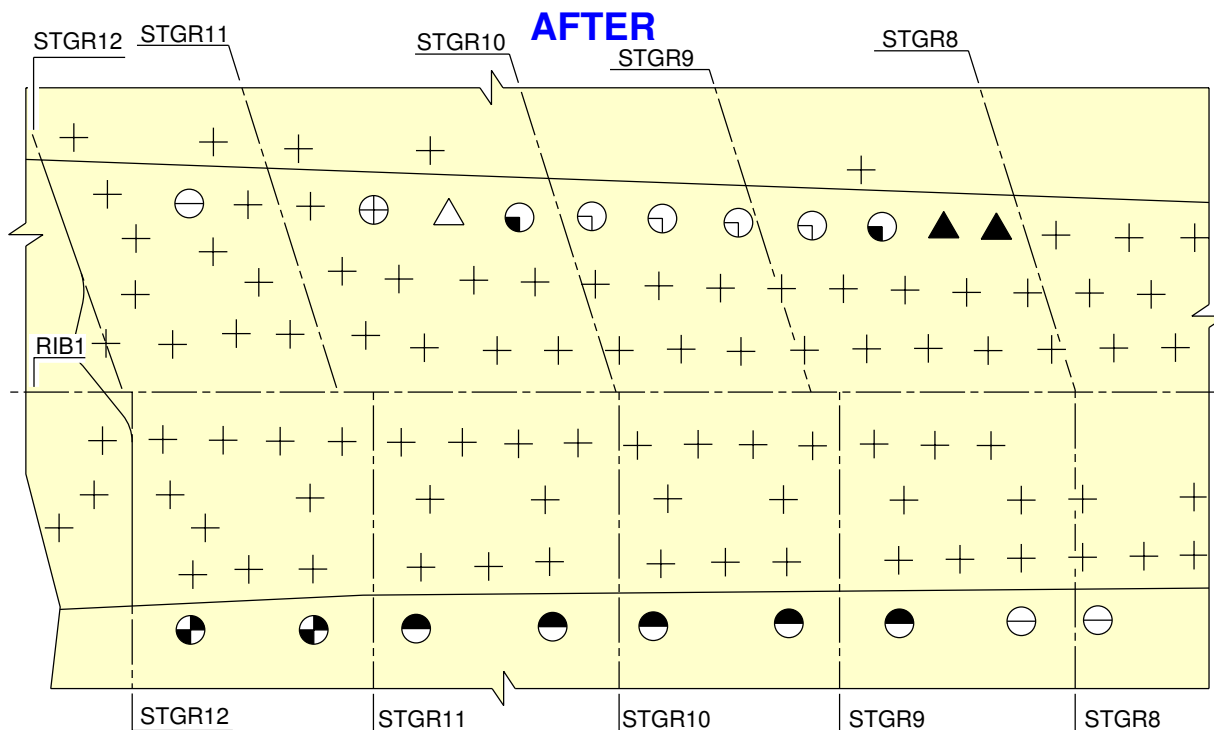
IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

N_SB_571140_5_CGAB_05_01

Figure A-FCGAB - Sheet 05
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 002**



HOLE	NEW ITEM	
⊖	11 OR	37 44
⊕	11 OR	43 44
⊗	13 OR	75 76
●	13 OR	74 76
⊖	13 OR	73 76
⊗	10 OR	78 35
⊡	10 OR	34 35
⊡	10 OR	82 35
▲	15 OR	80 11

NOTE:

+ FASTENERS NOT AFFECTED.

FOR THE INSTALLATION OF THE FASTENERS, DO A CHECK OF THE WASHER THICKNESS, REFER TO FIG. A-FCCAA OR FIG. A-FCDA.

IF YOU REPLACE ONLY THE NUT, TORQUE THE NUT TO BETWEEN 3.80 daN AND 4.60 daN (342 lbf.in AND 411 lbf.in).

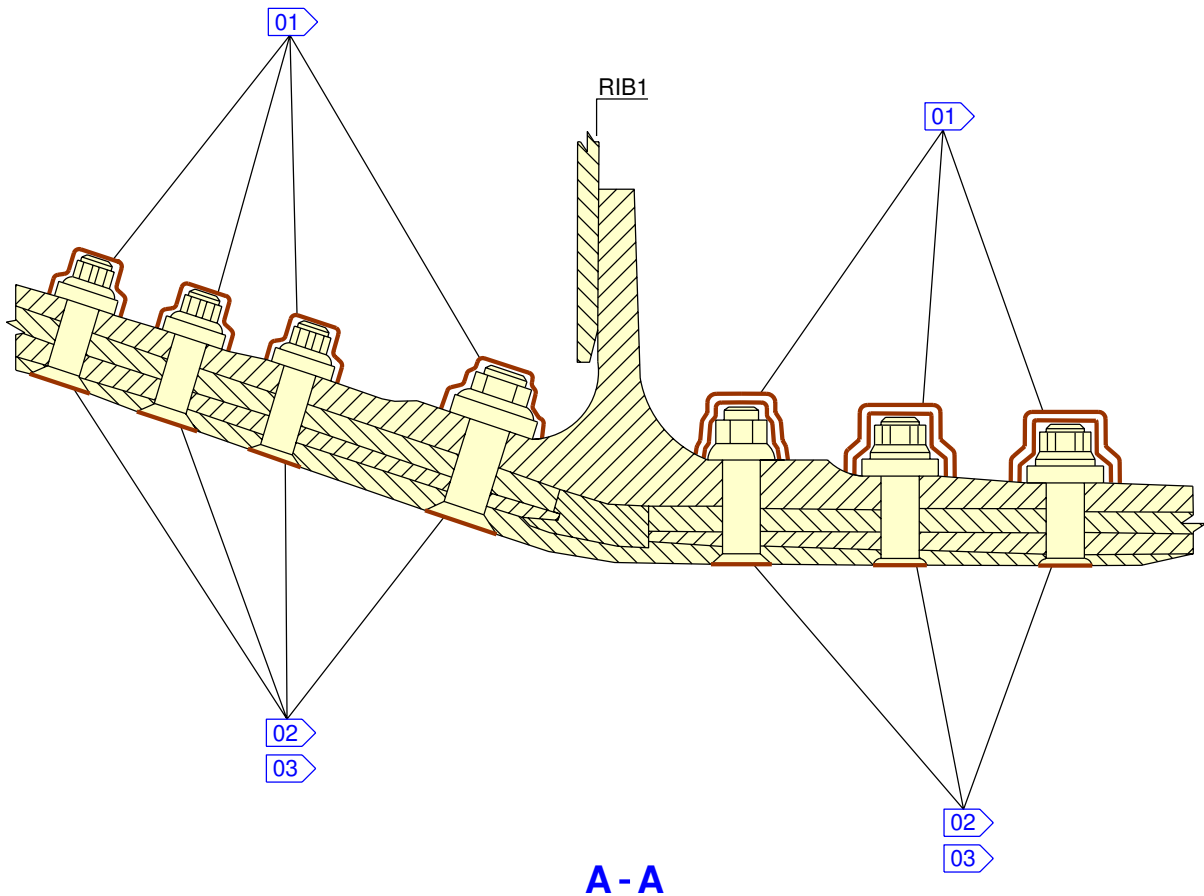
IF YOU REPLACE THE NUT AND THE SCREW, TORQUE THE NUT TO BETWEEN 5.60 daN AND 6.70 daN (493 lbf.in AND 592 lbf.in).

N_SB_571140_5_CGAB_06_01

Figure A-FCGAB - Sheet 06
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

****CONF 002**

SEALING PRINCIPLE



NOTE:

- 01** APPLY MATERIAL No 06ABB1 AND MATERIAL No 06PAG1 ON THE NUTS.
- 02** APPLY MATERIAL No 04EAC2, MATERIAL No 04CMA2 AND MATERIAL No 04JAA3 ON THE BOLTS.
- 03** WET ASSEMBLY, APPLY MATERIAL No 06ABB1.

N_SB_571140_5_CGAB_07_01

Figure A-FCGAB - Sheet 07
RH Side, Replacement of the Lower Panel Fasteners for the ADDITIONAL WORK

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A318/A319/A320/A321

SERVICE BULLETIN APPENDIX 01 - Weight Variant (WV) correspondence table

****CONF ALL**

(1) Weight Variant (WV) correspondence table

SERIES	AIRCRAFT MODEL	WEIGHT VARIANT (WV)	ASSOCIATED MODIFICATION	ASSOCIATED SERVICE BULLETIN
A320-200	A320-211 A320-212 A320-231	000		
		001	20966	SB A320-00-1002
		002	21601	
		003	22269	SB A320-00-1006
		004	21532	
		005	21711	
		006	22436	
		007	23264	SB A320-00-1013
		008	23900	SB A320-00-1030
		009	23900 , 22269	SB A320-00-1006 OR SB A320-00-1030
		010	23900 , 23264	SB A320-00-1013 OR SB A320-00-1030
		011	30307	SB A320-00-1077
		012	30479	SB A320-00-1069 OR SB A320-00-1227
		013	31132	SB A320-00-1060
		014	31385	SB A320-00-1063

SERIES	AIRCRAFT MODEL	WEIGHT VARIANT (WV)	ASSOCIATED MODIFICATION	ASSOCIATED SERVICE BULLETIN
A320-200	A320-214 A320-232 A320-233	000		
		002	21601	
		003	22269	SB A320-00-1006
		007	23264	SB A320-00-1013
		008	23900	SB A320-00-1030
		009	23900 , 22269	SB A320-00-1006 OR SB A320-00-1030
		010	23900 , 23264	SB A320-00-1013 OR SB A320-00-1030
		011	30307	SB A320-00-1077
		012	30479	SB A320-00-1069
		013	31132	SB A320-00-1060
		014	31385	SB A320-00-1063
		015	34047	SB A320-00-1126
		016	34094	SB A320-00-1143

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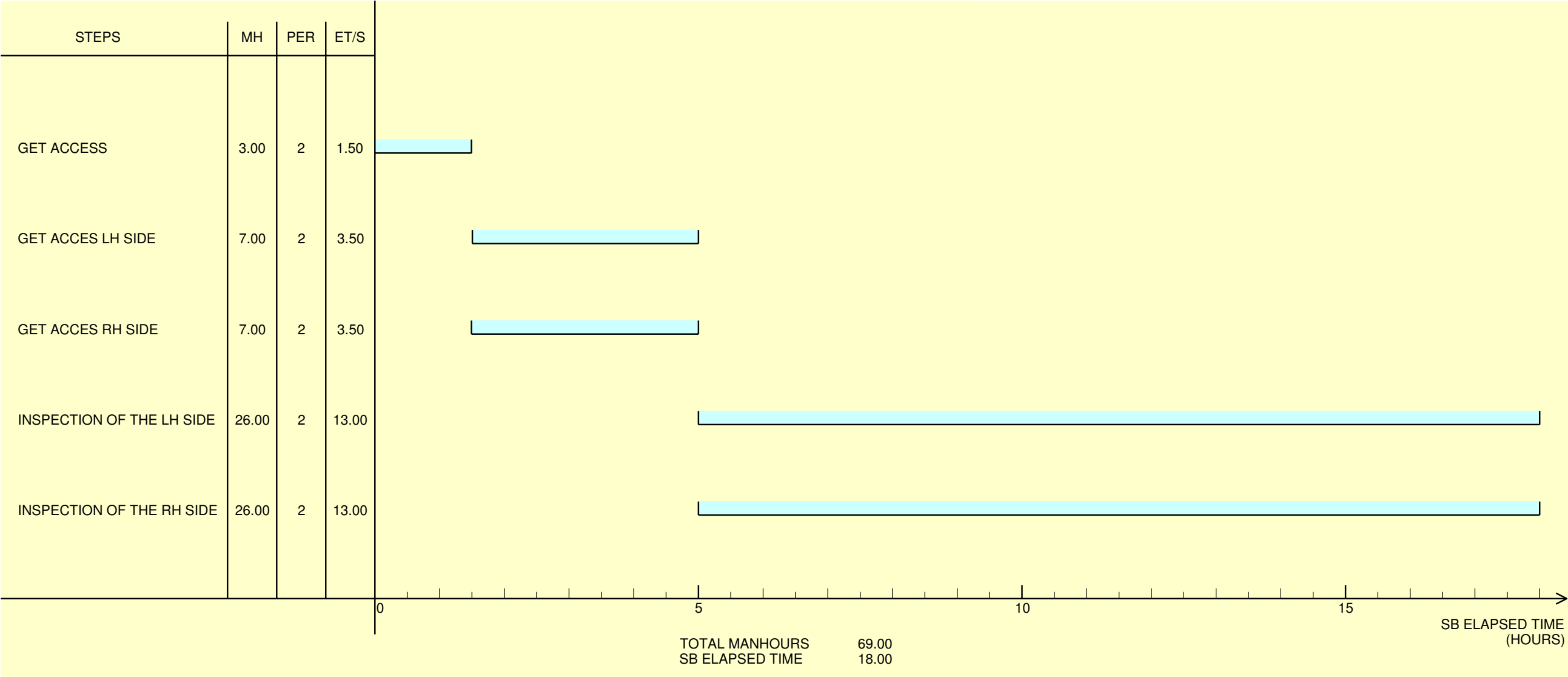
****CONF ALL**

The following chart is only a proposal. Operators may determine that another way to do the work is more suitable for them.

****CONF ALL**

ELAPSED TIME ASSUMPTION SERVICE BULLETIN No A320-57-1140

01
02



NOTE:
01 THIS CHART IS ONLY A PROPOSAL.
OPERATORS MAY DETERMINE THAT ANOTHER WAY
TO DO THE WORK IS MORE SUITABLE FOR THEM
02 IF A VSB IS INVOLVED, THE ASSOCIATED MANHOURS
ARE NOT REFLECTED IN THIS GANTT CHART

MH : MANHOURS
PER: NUMBER OF PERSONS
ET/S: ELAPSED TIME PER STEP

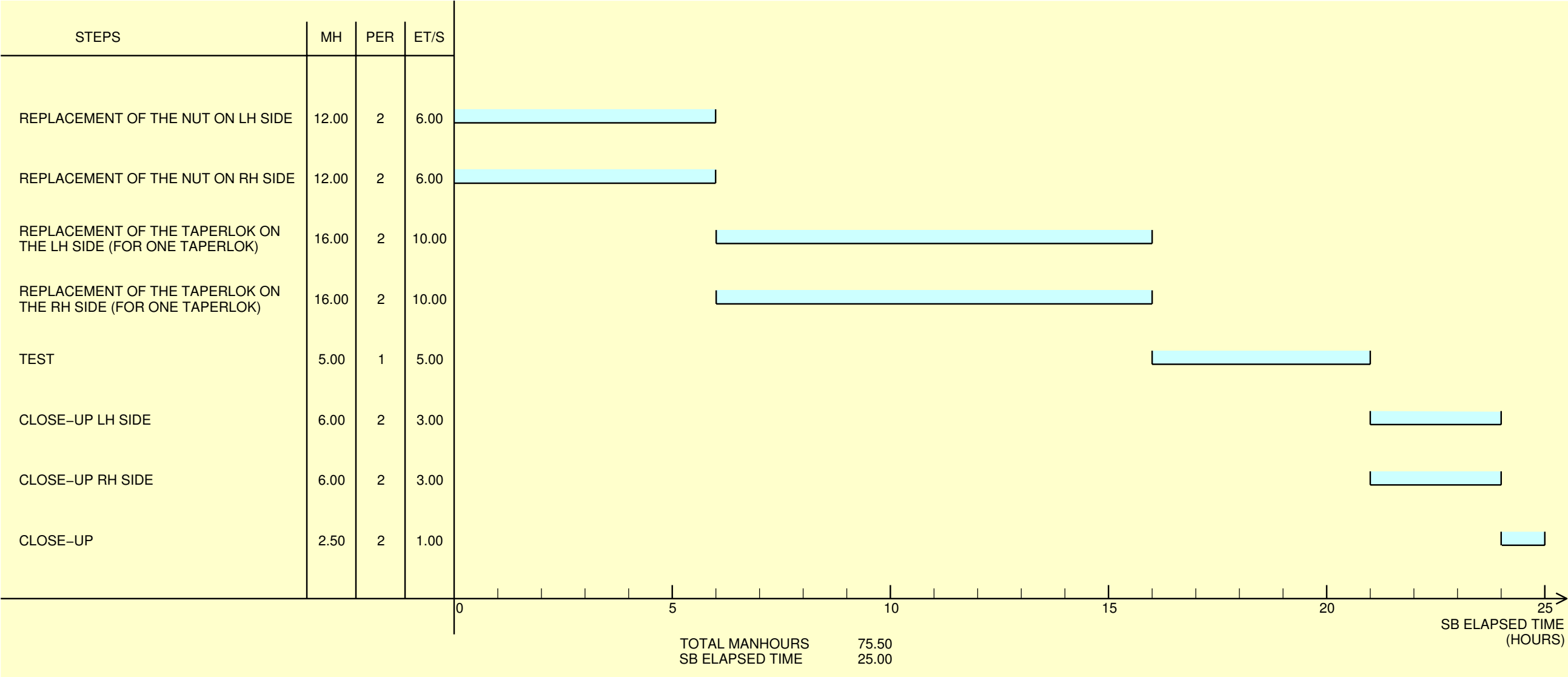
N_SB_571140_5_ABAA_01_00

Figure A-FABAA - Sheet 01
Elapsed Time Assumption Inspection Task

****CONF ALL**

ELAPSED TIME ASSUMPTION SERVICE BULLETIN No A320-57-1140

01
02



NOTE:
01 THIS CHART IS ONLY A PROPOSAL
OPERATORS MAY DETERMINE THAT ANOTHER WAY
TO DO THE WORK IS MORE SUITABLE FOR THEM
02 IF A VSB IS INVOLVED, THE ASSOCIATED MANHOURS
ARE NOT REFLECTED IN THIS GANTT CHART

MH : MANHOURS
PER: NUMBER OF PERSONS
ET/S: ELAPSED TIME PER STEP

N_SB_571140_5_ACAA_01_00

Figure A-FACAA - Sheet 01
Elapsed Time Assumption Repair Task

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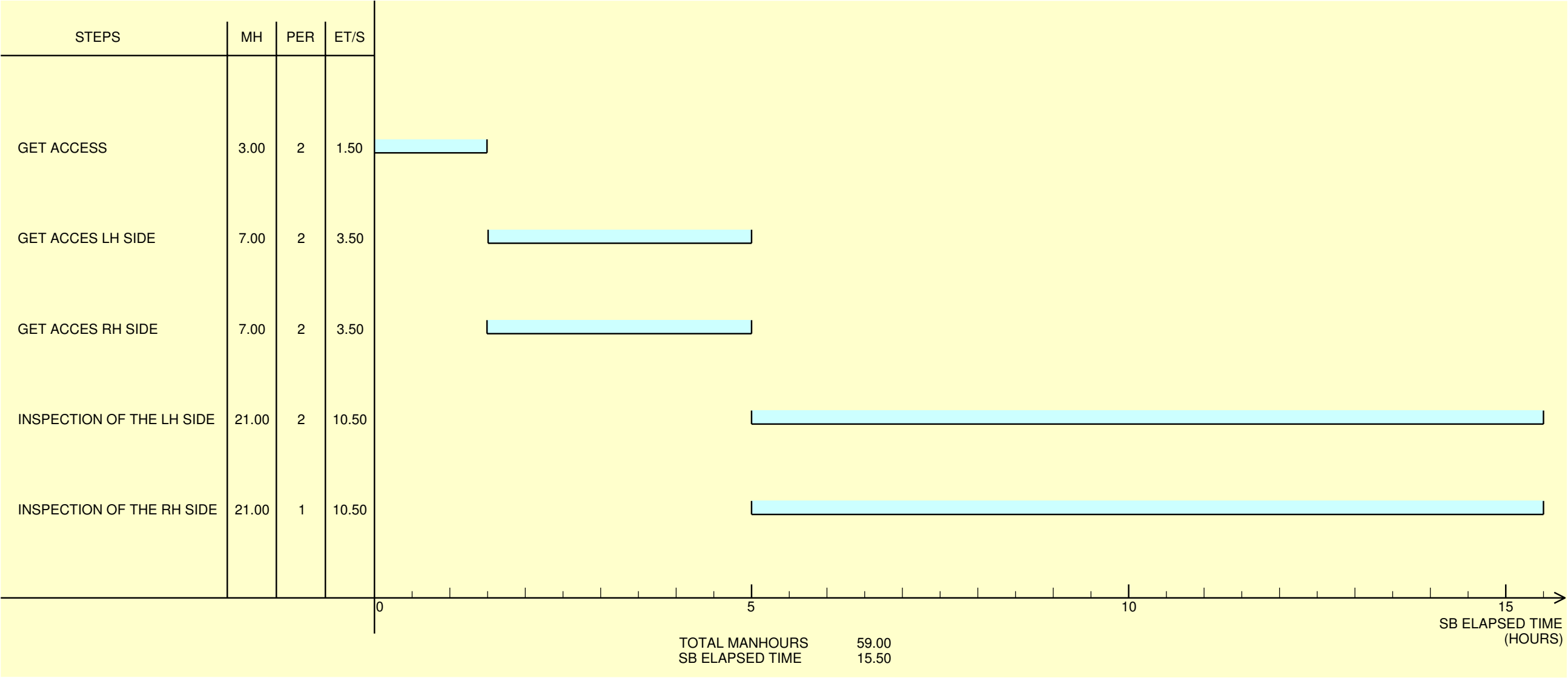
****CONF ALL**

The following chart is only a proposal. Operators may determine that another way to do the work is more suitable for them.

****CONF 001**

ELAPSED TIME ASSUMPTION SERVICE BULLETIN No A320-57-1140

01
02



NOTE:
01 THIS CHART IS ONLY A PROPOSAL.
OPERATORS MAY DETERMINE THAT ANOTHER WAY
TO DO THE WORK IS MORE SUITABLE FOR THEM
02 IF A VSB IS INVOLVED, THE ASSOCIATED MANHOURS
ARE NOT REFLECTED IN THIS GANTT CHART

MH : MANHOURS
PER: NUMBER OF PERSONS
ET/S: ELAPSED TIME PER STEP

N_SB_571140_5_ADAA_01_00

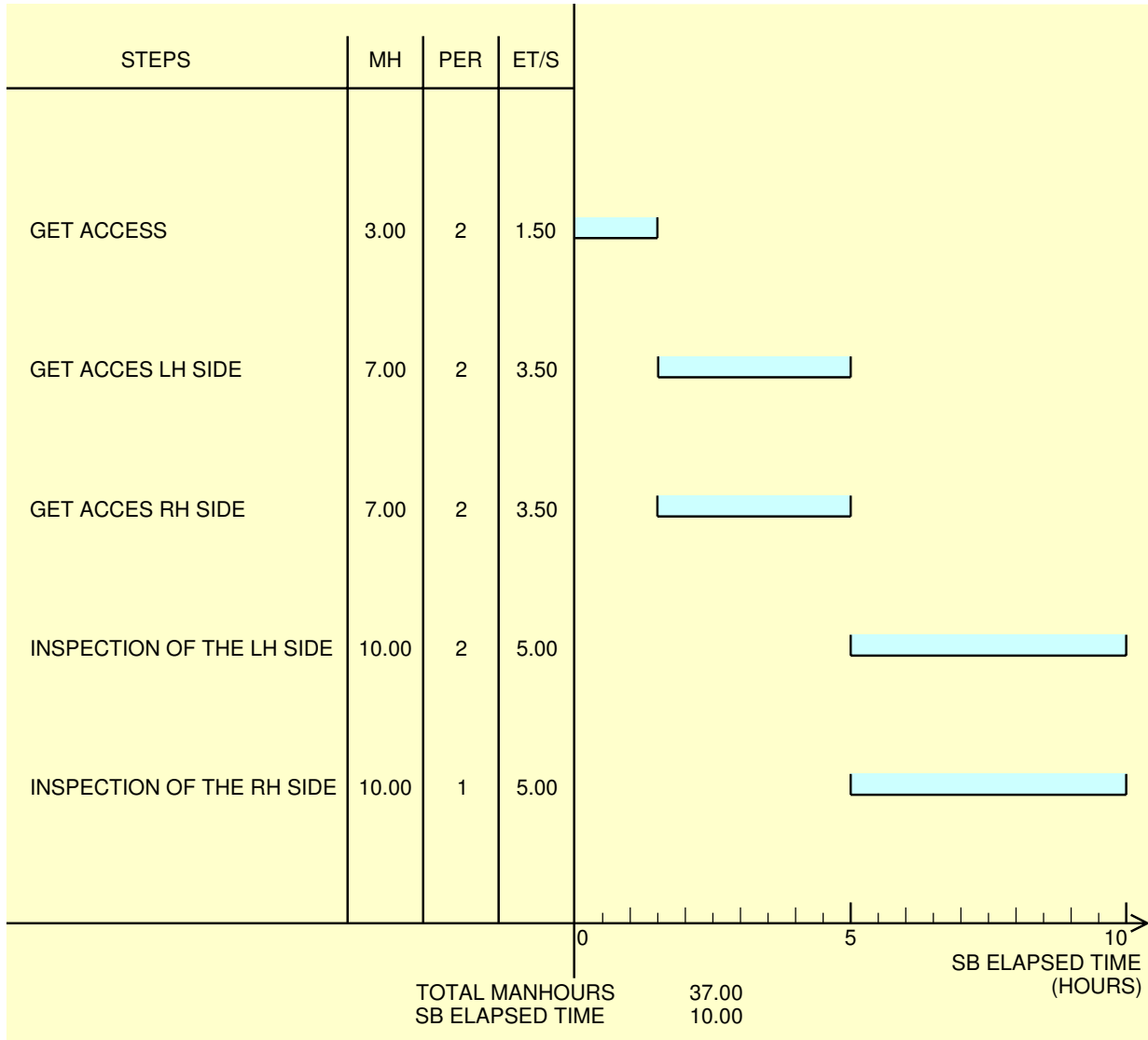
Figure A-FADAA - Sheet 01
Elapsed Time Assumption Inspection Task for ADDITIONAL WORK

SERVICE BULLETIN APPENDIX 03 - Elapsed Time Assumption (ADDITIONAL WORK)

****CONF 002**

ELAPSED TIME ASSUMPTION SERVICE BULLETIN No A320-57-1140

01
02



NOTE:

- 01** THIS CHART IS ONLY A PROPOSAL
OPERATORS MAY DETERMINE THAT ANOTHER WAY
TO DO THE WORK IS MORE SUITABLE FOR THEM
- 02** IF A VSB IS INVOLVED, THE ASSOCIATED MANHOURS
ARE NOT REFLECTED IN THIS GANTT CHART

MH : MANHOURS
PER : NUMBER OF PERSONS
ET/S : ELAPSED TIME PER STEP

N_SB_571140_5_ADAB_01_00

Figure A-FADAB - Sheet 01
Elapsed Time Assumption Inspection Task for ADDITIONAL WORK

5 DATE: Nov 21/06

SERVICE BULLETIN No.: A320-57-1140

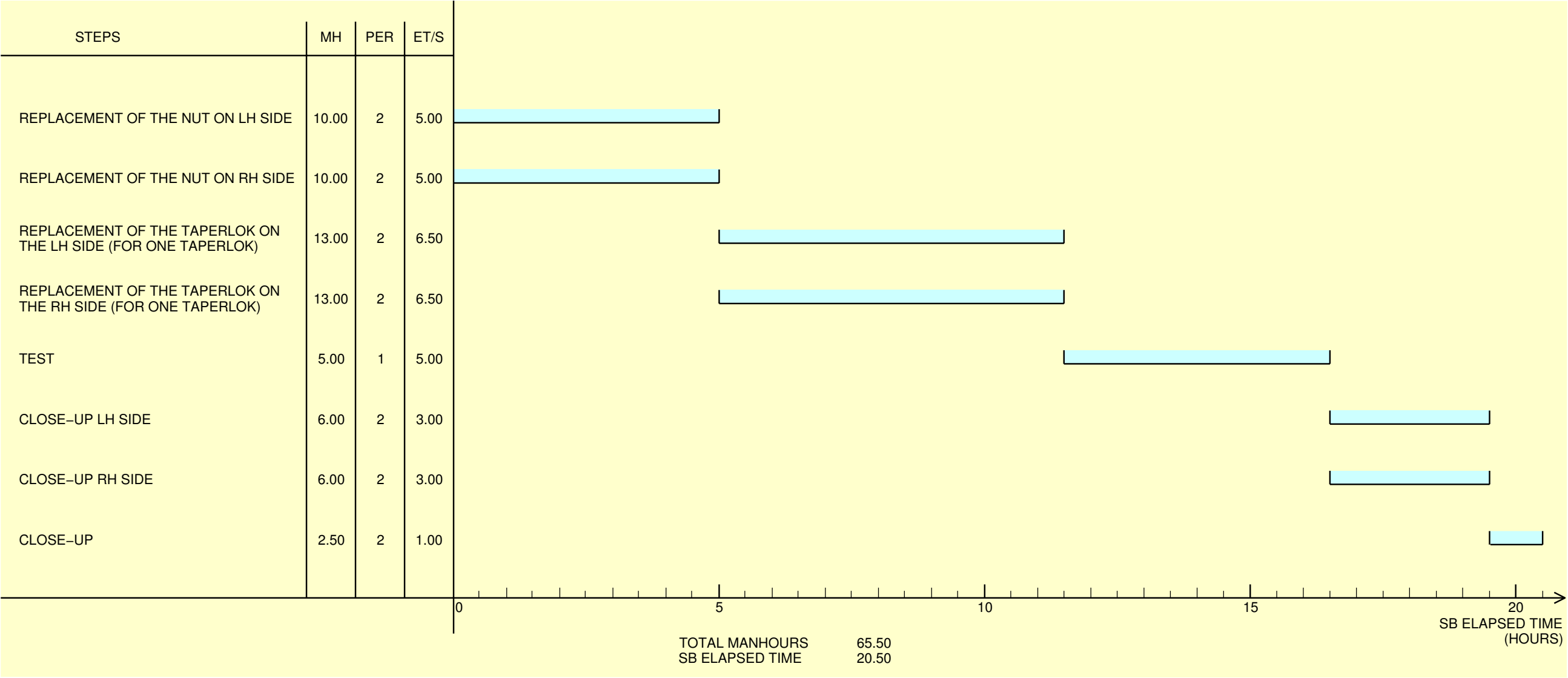
REVISION No.: 03 - Apr 19/19

Page: 3

****CONF 001**

ELAPSED TIME ASSUMPTION SERVICE BULLETIN No A320-57-1140

01
02



NOTE:
01 THIS CHART IS ONLY A PROPOSAL
OPERATORS MAY DETERMINE THAT ANOTHER WAY
TO DO THE WORK IS MORE SUITABLE FOR THEM
02 IF A VSB IS INVOLVED, THE ASSOCIATED MANHOURS
ARE NOT REFLECTED IN THIS GANTT CHART

MH : MANHOURS
PER: NUMBER OF PERSONS
ET/S: ELAPSED TIME PER STEP

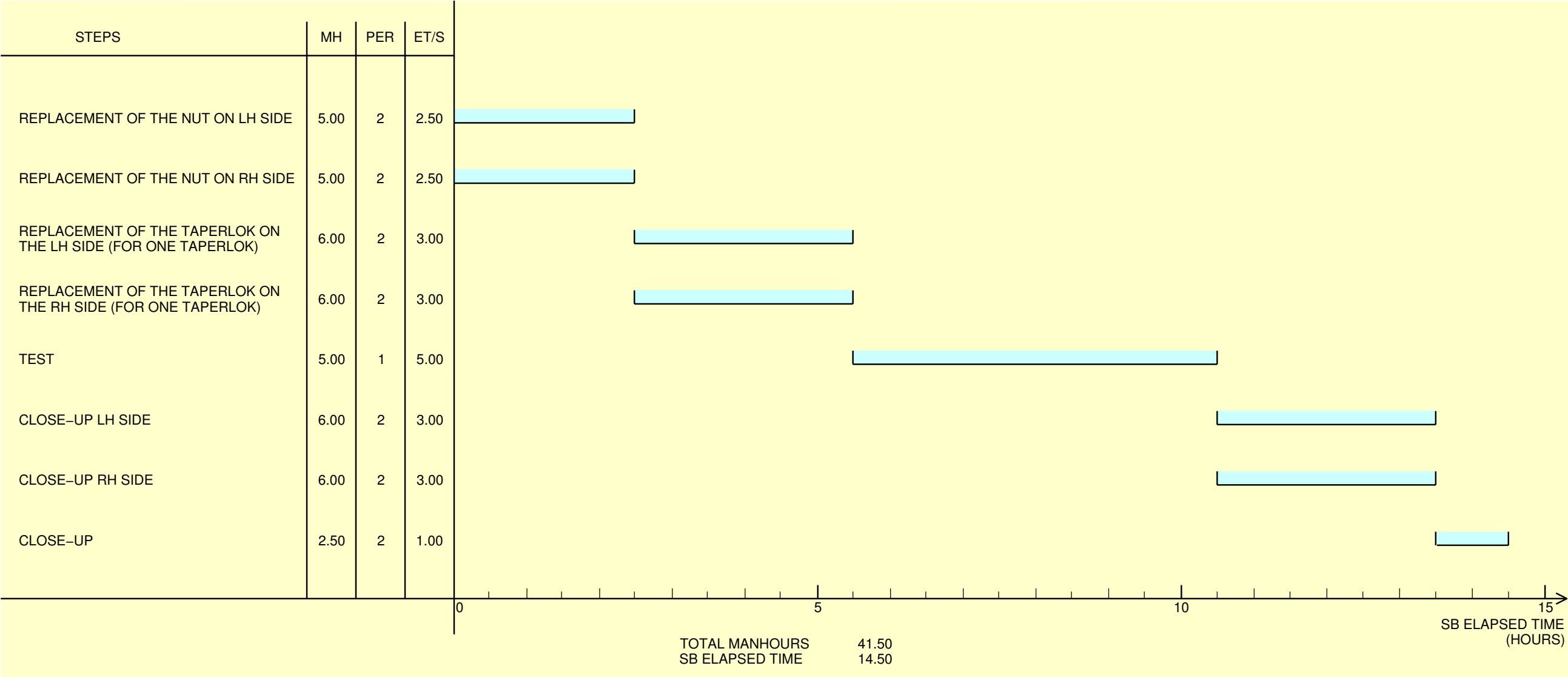
N_SB_571140_5_AEAA_01_00

Figure A-FAEAA - Sheet 01
Elapsed Time Assumption Repair Task for ADDITIONAL WORK

****CONF 002**

ELAPSED TIME ASSUMPTION SERVICE BULLETIN No A320-57-1140

01
02



NOTE:
01 THIS CHART IS ONLY A PROPOSAL
OPERATORS MAY DETERMINE THAT ANOTHER WAY
TO DO THE WORK IS MORE SUITABLE FOR THEM
02 IF A VSB IS INVOLVED, THE ASSOCIATED MANHOURS
ARE NOT REFLECTED IN THIS GANTT CHART

MH : MANHOURS
PER: NUMBER OF PERSONS
ET/S: ELAPSED TIME PER STEP

N_SB_571140_5_AEAB_01_00

Figure A-FAEAB - Sheet 01
Elapsed Time Assumption Repair Task for ADDITIONAL WORK

****CONF ALL**

The results of the inspection, including no findings, replacement or actions to be done, must be uploaded in accordance with ISI 00.00.00179 using the on-line reporting application.

****CONF ALL**

INSPECTION REPORT					
SERVICE BULLETIN No : A320-57-1140			UPLOAD THE COMPLETED INSPECTION REPORT SHEET IN ACCORDANCE WITH ISI 00.00.00179		
A/C MSN :					
FLIGHT CYCLES :					
FLIGHT HOURS :					
DATE OF INSPECTION :					
FINDINGS? : YES <input type="checkbox"/> NO <input type="checkbox"/>					
INSPECTION METHOD : <input type="checkbox"/> DETAILED INSPECTION					
CENTER WING BOX					
DOES THE GAGE FIT AND MOVE?					
SIDE	<input type="checkbox"/> YES		<input type="checkbox"/> NO		
	THREAD NOT DAMAGED	THREAD DAMAGED	THREAD MARKS FOUND	THREAD NOT DAMAGED	THREAD DAMAGED
LH					
RH					
OUTER WING BOX					
DOES THE GAGE FIT AND MOVE?					
SIDE	<input type="checkbox"/> YES		<input type="checkbox"/> NO		
	THREAD DAMAGED	THREAD NOT DAMAGED	THREAD MARKS FOUND	THREAD DAMAGED	THREAD NOT DAMAGED
LH					
RH					
PLEASE PRECISE OTHER FINDINGS:					

N_SB_571140_5_AAAA_01_01

Figure A-FAAAA - Sheet 01
Inspection Report

TITLE: WINGS - GENERAL - ONE TIME INSPECTION OF TAPERLOK BOLT AT WING TO FUSELAGE JUNCTION.

MODIFICATION No.: None

SB Reporting status

This Service Bulletin will only be incorporated into affected customized Maintenance and Flight Operations Products if it is duly reported to Airbus.

General information

The SB Reporting policy is published in ISI 00.00.00135. This document provides information regarding: the SB Reporting means and benefits, the update of customized Maintenance Products and customized Flight Operations Products.

If this Service Bulletin has an operational impact, refer to paragraph "Flight Operations Products" for further information on the update of affected Flight Operations Products.

SB Reporting procedure

Standard practice to report SB is to use the online reporting application on AirbusWorld.

Refer to ISI 00.00.00179:

- For complete information on reporting means to Airbus. Question 2 "What are the standard practices for SB reporting?"
- If this SB requires previous or simultaneous accomplishment of other SBs, Question 3 "What about the reporting of prerequisite Service Bulletin?"

NOTE: ISI 00.00.00179 covers all the frequently asked questions about SB Reporting. It is revised on a regular basis.

SERVICE BULLETIN
QUALITY PERCEPTION FORM

Use this form to tell us what is your perception of the quality of this Service Bulletin. The reported data that you provide us will be used to analyse areas of difficulties and to take corrective action to further improve the quality of our Service Bulletins.

We thank you for the time you have taken in completing this form.

(Please rate on a scale of 1 to 4, with 4 being the highest score)

- Quality rating of this SB	4	3	2	1
- Quality rating of the Accomplishment Instructions	4	3	2	1
- Quality rating of the Illustrations	4	3	2	1
- Is this SB easy to understand ?	Y / N			

If you have had difficulties in the accomplishment of this SB please quote below the area(s) and give a short description of the issue.

Planning	Material	Instructions
X Effectivity	X Kit content	X Preparation
X Reason	X List of Materials Operator Supplied	X Mod/Inspection
X Manpower	X Industry support	X Test
X References	X Re-identification	X Close-Up
X Publication	X Tooling	X Illustrations

Comments:

Operator:

Date:

Name/Title:

Please return this form to:

AIRBUS
CUSTOMER SERVICES DIRECTORATE
1 Rond Point Maurice Bellonte
31707 BLAGNAC CEDEX FRANCE

Attn : SEM4 Service Bulletins Management
FAX : (33) 5 61 93 42 51

Or via your Resident Customer Support Office.